

AFIT/GAE/ENY/98M-03

EFFECT OF HIGH FREE STREAM TURBULENCE ON FILM COOLING  
USING DOUBLE ROW OF 30° SLANT-HOLE INJECTORS

THESIS

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19980423 065

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THESIS

Presented to the Faculty of the School of Engineering

of the Air Force Institute of Technology

Air University

In Partial Fulfillment of the

Requirements for the Degree of

Master of Science in Aeronautical Engineering

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March 1998

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## **Acknowledgments**

This study was a continuation of a project conducted at the Wright Research and Development Center (WRDC), Aero Propulsion and Power Laboratory, Components Branch (POTC) Turbine Engine Division. The test setup and data acquisition used for the study was done by Major Dino Ishikura.

I wish to express my deep appreciation and thanks to my faculty advisor Dr. Paul King for his technical guidance, encouragement, mentoring and patience throughout the time span spent to conduct this thesis research.

I would also like to thank Dr. Richard Rivir and Mr. Greg Cala from the Aero and Power Laboratory for providing assistance and the background information needed to continue the project.

Finally, I am deeply indebted to my husband, Scott and my children, Nichole and Raymond, for their support and understanding during this research.

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## List of Symbols

$b$	- wall jet nozzle exit, (cm)
$B$	- blowing ratio
$B_{opt}$	- optimum blowing ratio
$CD$	- coefficient of discharge
$C_f$	- skin friction coefficient
$C_p$	- specific heat at constant pressure (J/kg $^{\circ}$ K)
$d$	- injection hole diameter (cm))
$D$	- injection hole diameter (cm)
$h$	- heat transfer coefficient (W/m $^2$ $^{\circ}$ K)
$h_o$	- heat transfer coefficient with film cooling (W/m $^2$ $^{\circ}$ K)
$I$	- momentum flux
$J_u$	- integral time scale (s)
$k$	- conductivity (W/cm $^{\circ}$ K)
$K$	- Von Karman constant
$l$	- thickness (cm)
$L$	- integral length scale (cm)
$M_j$	- wall jet momentum flux
$P_{bar}$	- barometric pressure (N/m $^2$ )
$P_{kiel}$	- kiel probe measured pressure (N/m $^2$ )
$P_{tank}$	- pressure in main air supply plenum chamber (N/m $^2$ )
$P_{tankcool}$	- pressure in main coolant plenum chamber (N/m $^2$ )
$P/D$	- pitch to diameter ratio

$Pr$	- Prandlt number
$q$	- heat flux ( $W/m^2$ )
$Re$	- Reynolds number
$R_{air}$	- gas constant for air ( $J/kg\ ^\circ K$ )
$S$	- spanwise hole spacing (cm)
$St$	- Stanton number
$St_0$	- Stanton number with low free stream turbulence
$t$	- time (s)
$T$	- static temperature ( $^\circ K$ )
$T_{bs}$	- temperature at bottom surface of the plate ( $^\circ K$ )
$T_{hw}$	- free stream recovery temperature ( $^\circ K$ )
$T_r$	- recovery temperature ( $^\circ K$ )
$T_{tank}$	- temperature in main air supply plenum chamber ( $^\circ K$ )
$T_{tankcool}$	- temperature in coolant supply plenum chamber ( $^\circ K$ )
$Tu$	- turbulence
$T_w$	- wall temperature ( $^\circ K$ )
$T\phi$	- air-coolant mixture temperature ( $^\circ K$ )
$u$	- instantaneous velocity fluctuation (m/s)
$u'$	- velocity fluctuation (m/s)
$u_o$	- local maximum velocity at a station (m/s)
$U$	- velocity (m/s)
$U_j$	- wall jet nozzle exit velocity (m/s)
$U_m$	- local maximum velocity at a station (m/s)

$U_x$	- total velocity component in the x direction (m/s)
$U_\tau$	- friction velocity (m/s)
$U_{\tau_{au}}$	- friction velocity (m/s)
$U_\infty$	- free stream velocity at injection (unless otherwise specified, m/s)
$U_{inf}$	- free stream velocity at injection (unless otherwise specified, m/s)
$U_1$	- free stream velocity (m/s)
$\nu$	- kinematic viscosity ( $m^2/s^2$ )
$x$	- longitudinal distance from nozzle exit to a station (cm)
$X$	- longitudinal distance from injection (cm)
$X_{inj}$	- longitudinal distance from nozzle exit to injection (cm)
$X_L$	- longitudinal distance from nozzle exit to a station (cm)
$y$	- vertical distance from the plate(m)
$y_m$	- vertical distance where the profile velocity is a maximum (m)
$y_{m/2}$	- vertical distance where the profile velocity is half its maximum value (m)
$z$	- lateral distance from the plate centerline (cm)

#### Greek Symbols

$\delta$	- boundary layer thickness (cm)
$\delta_*$	- displacement thickness(cm)
$\gamma$	- friction factor
$\eta$	- film cooling effectiveness
$\nu$	- kinematic viscosity ( $m^2/s$ )
$\rho$	- density ( $kg/cm^3$ )



$\tau_w$  - wall shear stress (N/m<sup>2</sup>)

#### Subscripts

aw - adiabatic wall

c - coolant

j - nozzle exit

w - wall

$\infty$  - free stream

## **Abstract**

Film cooling has an inherent complex flow structure due to the injection of coolant into the mainstream boundary layer. Phenomena such as film blow-off at high blowing rates are part of the experience when using discrete holes to inject the coolant. Inside a turbine, this flow structure is further complicated by the introduction of free stream turbulence. Knowledge of the effect free stream turbulence has on film cooling effectiveness serves to improve the prediction of coolant requirements during turbine operation .

In this study free stream turbulence levels from 7.3 % to 17.8 % are applied to film cooling over a flat plate with two rows of 30° slant-holes using a wall jet as the main stream air supply. Blowing ratios are varied from 0.25 to 2.0 and free stream velocities at injection range from 10 m/s to 85 m/s. A constant density ratio of 1.07 is kept throughout the experiment.

Results show that for different magnitudes of blowing ratio free stream turbulence has a different influence on effectiveness. At the forward stations, where blow-off was present at high blowing ratios, high free stream turbulence increased effectiveness. While at mid to aft stations for all blowing ratios high free stream turbulence decreased effectiveness.

A correlation for effectiveness with a non-dimensional group of film cooling parameters is offered. The effectiveness correlated best 10 to 15 diameters after injection where blow-off effects are no longer present, and before the very far downstream stations where uncertainties in the measurements were greater.

# 1. Introduction

## 1.1 Background

A common practice to improve the performance of advanced gas turbine engines is to increase engine operating temperatures in order to increase the thermodynamic efficiency. Current engine operating temperatures are in the neighborhood of 1750°K. To maintain structural integrity under such a high temperature environment cooling of turbine blades is still necessary, despite major material advances leading to the use of components made from nickel superalloys and ceramics. Consequently, film cooling continues to be an important technology development area.

Film cooling is applied through the injection of coolant onto the surface of the blade. This is done by bleeding air from the compressor and redirecting it to channels inside the turbine blade. The air is then injected onto the blade through holes on the surface of the blade. At present, less than 10 % of total engine compressor flow is used for cooling. The subtraction of this air from the main compressor flow produces unwanted work losses. The reduction of these losses is a big motivation for the achievement of better prediction of coolant requirements.

The parameter used to quantify the performance of film cooling, and therefore predict coolant requirements, is referred to as the adiabatic effectiveness and defined as:

$$\eta = \frac{T_{aw} - T_{\infty}}{T_c - T_{\infty}} \quad (1-1)$$

where  $T_{aw}$  is the adiabatic wall temperature,  $T_{\infty}$  is the free stream temperature and  $T_c$  is the coolant temperature. (For high speed flow the flow temperatures would be replaced by recovery

temperatures.) Additional insight into these parameters is provided in the following chapter on film cooling theory.

Another important parameter is the indicator of the mass flux ratio of coolant injection to free stream air, known as the blowing ratio and defined as:

$$B = \frac{\rho_c U_c}{\rho_\infty U_\infty} \quad (1-2)$$

where  $\rho_c$  and  $\rho_\infty$  are the density of the coolant and free stream, respectively; and  $U_c$  and  $U_\infty$  are the velocities of the coolant and free stream, respectively.

In situations like blade cooling where high strength requirements exclude the use of a continuous slot and discrete hole injection is necessary, the blowing rate becomes a critical parameter in terms of film coverage of the plate. For high blowing rate injection, mainly  $B > 1.0$ , the cooling jets detach from the surface and penetrate into the mainstream. This leaves the immediate area after injection with little protection and allows mainstream air to come in contact with the surface. This occurrence is commonly known as blow-off, and is expected to have a marked effect on how free stream turbulence affects effectiveness in this experiment.

Like the blowing ratio, the choice of film cooling configuration parameters also has an impact on the magnitude of the effectiveness achieved. For instance, the more parallel to the surface the injection of coolant is, the better the coverage and reach of the film over the surface. Therefore, the smallest possible injection angle should be used. However, current manufacturing technology limits this angle to  $30^\circ$ . The number of injection hole rows used also affects the end effectiveness. In experiments by Han and Menhendale (1986), and Cirriello (1991), two rows of holes have been shown to be superior to one row. Another important configuration parameter is the injection hole pitch ( $P/D$ ). A pitch of 3.0 produced higher effectiveness than a pitch of 5.4 in

studies by Jubran and Brow (1985) with two rows of holes 10D apart. Most film cooling studies have been performed at  $P/D = 3.0$ . The present work was done using  $P/D = 4.0$ , to provide data within the range of  $3.0 < P/D < 5.4$  which could be used to find the largest  $P/D$  less than 5.4 that still produces comparable values of  $\eta$ . Relative placement of the rows with one another also affects the performance of film cooling. Afejuku et al. (1980 and 1983) tested in-line and staggered configurations and showed the latter configuration to be superior.

The ratios of thermodynamic properties, such as density, between coolant and free stream also affect the performance of film cooling. In studies by Goldstein et al. (1966 and 1974) and Perderson et al. (1977) using Helium, Freon-12 and  $\text{CO}_2$  to produce different density ratios ( $DR = \rho_c/\rho_\infty$ ), maximum  $\eta$  occurred at a higher density ratio with higher  $B$ . During actual turbine operation the coolant density is considerably higher than that of the free stream; the density ratio can be as high as 2.0. In the present experiment the secondary flow equipment limits the density ratio to 1.07.

The complexity of the flow structure inherent in the film cooling process also plays a major role on the achieved effectiveness. Secondary flow injection into a free stream increases the boundary layer thickness,  $\delta$ , which in turn decreases heat transfer to the surface (Goldstein, 1971). Injection also causes the main flow speed to decrease upstream of the injecting jets creating an increase of pressure at this location while a decrease in pressure occurs at the downstream side of injection (Goldstein, 1971). This phenomenon results in a deformation of the jets. Vortices are generated within the jets which produce an intermixing of the coolant and free stream. Coolant flow is expelled from the jet while free stream air is introduced into the coolant region. Additionally, the interface or shear region between coolant and mainstream is characterized by

large scale turbulent structures which promote intermixing between the flows (Kohli and Bogard, 1996).

In practice, the complexity of the flow interaction between primary and secondary flow during film cooling is further complicated by the presence of free stream turbulence. Turbine blades are typically subjected to free stream turbulence in excess of 20 %. Even though this is the case, a relatively small number of film cooling studies have been performed with high free stream turbulence. One of the first comprehensive studies was conducted by Kadotani and Goldstein in 1979. Using turbulence generation grids, Tu levels from 0.3 % to 20.6 % with a length scale of 0.06d to 0.3d were applied to film cooling with one row of injection holes. They concluded that the free stream turbulence affects the effectiveness through changes in boundary layer thickness, mixing between mainstream and injected flow, shape of the injected flow due to vortex formation, and penetration height of the injected flow. For different blowing ratios the predominant factor among those listed varied. They also found that as the flow moves downstream, the effect of free stream turbulence becomes smaller. In a later study by Jumper (1988), a wall jet was used to generate free stream turbulence of 14 % to 17 %. Film cooling was applied through a single row of 30° slant-hole injectors. He found that at low B free stream turbulence decreased  $\eta$ . In a more recent study, Bons et al. (1994) used a single row of 35° slant-hole injectors,  $P/D = 3.0$  and  $DR = 0.95$ . Turbulence was generated by a grid for the low Tu (0.9 %) and by jets in cross-flow for the mid to high Tu (6.5 % to 17.4 %). Bons et al. (1994) found that the effect free stream turbulence has on effectiveness depends on the magnitude of the blowing ratio. For B between 0.55 and 0.95, where blow-off is not normally present, free stream turbulence decreased centerline  $\eta$ . At these blowing ratios the reduction in  $\eta$  was dominated by the increased dissipation of the coolant into the mainstream. For  $B > 0.95$ , where blow-off is present, high free stream turbulence increased centerline and midline effectiveness at the forward stations by reintroducing detached

coolant closer to the surface and by promoting the lateral spread of the coolant jets. Similar effects to all these described are expected in this study which includes a comparable set of parameters. Schmidt and Bogard (1996) conducted a study using a single row of 30° slant-hole injections,  $P/D = 6.5$  and Tu levels of 0.3 %, 10 %, and 17 %. In their study, laterally averaged  $\eta$  decreased with high Tu levels for low B. Kohli and Bogard (1997) studied the effect of free stream turbulence on the coolant and mainstream flow interaction using a single row of 35° slant-hole injectors,  $P/D = 1.05$  and Tu levels of 0.5 % and 20 %. They determined that the coolant dispersion was initially dominated by the shear layer generated turbulence, followed farther downstream by large turbulent structures from the free stream.

All of the studies mentioned above studied the effects of free stream turbulence on effectiveness using a single row of injectors. The present work will study the case for two rows of 30° slant-hole injectors, wall jet generated Tu levels of 7.3 %, 12.2 % to 17.8 %,  $DR = 1.07$  and  $P/D = 4.0$ . Additionally, each set of measurements will be taken for five different free stream velocities at injection.

Better understanding of high free stream turbulence effects on the mechanics of film cooling, and how these effects are influenced by other film cooling parameters, will enable improved modeling of the process and therefore more accurate prediction of  $\eta$ .

## 1.2 Problem

The goal of this study is to investigate the effect free stream turbulence has on film cooling effectiveness as manifested in measurements for the case of two rows of 30° slant-hole injectors.

### 1.3 Scope

Effectiveness was generated for various values of  $T_u$ ,  $B$  and  $U_\infty$  at injection

- Free stream turbulence was varied from 7.3 % to 17.8 %
- Blowing ratio was varied from 0.25 to 2.0
- $U_\infty$  was varied from 10 m/s to 85 m/s

Density ratio was constant at 1.07.

### 1.4 Assumptions

In order to create a near-adiabatic wall condition, no power was applied to the plate from which surface temperatures were measured. At zero voltage, conduction and radiation losses are still present. Effectiveness was corrected for these losses in the normal direction to the plate.

Losses in the lateral and longitudinal direction were assumed negligible.

### 1.5 Experimental Approach

Effectiveness was measured using a set-up designed to replicate film cooling conditions over a turbine blade using a practical range of parameters. A wall jet supplied the main stream air over the plate. The coolant air was injected through 2 rows of holes on the surface of the plate. The coolant temperature was an average of 10 to 20 degrees lower than the main stream air. With no power supplied to the plate, free stream and surface temperatures were collected along the plate.



This was done for each velocity and blowing ratio combination for all three levels of free stream turbulence.

## **1.6 Sequence of presentation**

Following the introduction the film cooling effectiveness and wall jet characterization is presented, followed by a description of the experimental apparatus, experimental procedures and data reduction, results and discussion, and finally the conclusion.

## 2. Film Cooling Effectiveness and Wall Jet flow Characterization

### 2.1 Film cooling Effectiveness

Inside a turbine, film cooling is primarily used to reduce the convective heat transfer rate from the hot gases to an exposed blade. The secondary fluid not only produces an insulating layer which protects the surface of the blade, but it also acts as a heat sink by lowering the mean temperature in the boundary layer. The heat flux from the hot gases to the surface can be written

$$q = h_0 (T_{aw} - T_w) \quad (2-1)$$

where  $h_0$  is the heat transfer coefficient with film cooling,  $T_w$  is the wall temperature, and  $T_{aw}$  is the adiabatic wall temperature. The latter temperature is the surface temperature for the limiting case of a perfectly insulated surface where the heat flux would be zero. Without film cooling injection this adiabatic temperature would be the same as the free stream temperature or for high speed flow the recovery temperature. With injection, the resulting  $T_{aw}$  depends on the temperatures of the free stream and the coolant stream. To bypass this temperature dependence, the adiabatic wall temperature can be expressed in a dimensionless form called the film cooling effectiveness as:

$$\eta = \frac{T_{aw} - T_\infty}{T_c - T_\infty} \quad (1-1)$$

where  $T_\infty$  is the free stream temperature and  $T_c$  is the temperature of the coolant at the point of injection (Goldstein, 1971). For high speed flows  $\eta$  is expressed as:

$$\eta = \frac{T_{aw} - T_{r\infty}}{T_{rc} - T_{r\infty}} \quad (2-3)$$

where  $T_{r\infty}$  is the free stream recovery temperature and  $T_{rc}$  is the recovery temperature of the coolant at injection. The limiting values of effectiveness are unity at the point of injection ( $T_{aw} = T_{rc}$ ), and zero far downstream of injection where the secondary flow is greatly diluted and the wall temperature approaches the free stream recovery temperature.

## 2.2 Wall Jet Flow

### 2.2.1 General characteristics

When a fluid is blown parallel to a surface, it forms a configuration known as a wall jet. Figure 2-1. Shows the typical mean velocity distribution across a two-dimensional wall jet and defines the nomenclature. The wall jet is a two-layer shear flow that combines the characteristics of a boundary layer and a free jet. The region from the wall to  $Y_m$  is the inner layer, in which the flow exhibits similarities in structure with that of a conventional turbulent boundary layer. The region extending from  $Y_m$  to the outer edge of the flow is the outer layer (Launder and Rodi, 1983).

Modeling the high turbulence intensity of real engine blades under typical operating conditions in a laboratory is difficult. Many researchers, such as MacMullin et al. (1988), Maciejewski and Moffat (1989), Jumper et al. (1988), etc., used the aggressive flow characteristics of a wall jet over a flat plate at atmospheric pressure and temperature to generate a high turbulence level. MacMullin et al. (1988) determined that turbulence intensity in the wall jet boundary layer varied as a function of longitudinal distance along the plate and was largely independent of nozzle velocity. This is an important fact because a wide range of Reynolds number may be covered by varying the nozzle exit velocity, without much change in the local turbulence level. MacMullin et

al. used a round ASME nozzle to generate the free jet with high turbulence intensity level for his study.

Eckerle (1988) used a 7 cm x 48 cm short radius rectangular nozzle to create a wall jet. He found that the rectangular geometry reduced the edge effects observed in previous tests with a circular wall jet, in which a variation was observed in the velocity field and turbulence intensity in the lateral border of the jet. Eckerle (1988) also found a region in the centerline of the flow at a distance three times the nozzle diameter, where the velocity field and the turbulence intensity suffered perturbations. This could have been caused by the recirculation over the plate, due to induced flow over the settling chamber. For the present research, a new longer rectangular nozzle was used to eliminate this problem.

### **2.2.2 Applicable scaling laws**

For the wall jet flow, the choice of scaling parameters or the application of a specific scaling law has not entirely been agreed on by researchers in the field. A study by Wygnanski et al. (1992) on the applicability of scaling laws to the turbulent wall jet provides valuable information on the subject. In the study, the wall jet flow was scaled by the local length and velocity scales, by outer flow variables, and by wall jet momentum flux.

For the initial scaling, Wygnanski et al. (1992) plotted velocity profiles for several values of  $Re_j$  in  $Y/Y_{m/2}$  and  $U/U_m$  coordinates. A good collapse was obtained at  $Y$  locations far from the wall ( $Y/Y_{m/2} \sim 1$ ), but near the wall the profiles were dependent on  $Re_j$ . This finding triggered their search for a different way of scaling the data which could sustain self-similarity for a range of  $Re_j$ .

Wyganski et al. (1992) obtained a good collapse for all velocity profiles and all  $Re_j$  through the use of outer variables,  $Y/Y_{m2}$  and  $(U-U_m)/U_\tau$ , where,

$$\frac{U - U_m}{U_\tau} = A \log\left(\frac{Y}{Y_m}\right) + A \Pi \left[ w\left(\frac{Y}{Y_m}\right) - w(1) \right], \quad (2-4)$$

and

$$w\left(\frac{Y}{Y_m}\right) = 3\left(\frac{Y}{Y_m}\right)^2 - 2\left(\frac{Y}{Y_m}\right)^3$$

is a wake function curve fit (White, 1991),  $\Pi$  is Coles wake parameter equal to 0.45 for a flat plate (White, 1991), and  $A$  is the universal constant of the inner scaling laws equal to 3.9 for the profiles in this experiment. (In his 1992 paper Wyganski et al. points out that several researchers, Myers et al. (1963), Bradshaw and Gee (1962) and R. P. Patel (1962), question the universality of the constant  $A$  and propose constants varying from 3.9 to 4.75). Wyganski et al. (1992) showed that outer variables correlate well beyond the maximum velocity point ( $U = U_m$ ) and well below the location where the Reynolds stress,  $(-\overline{\rho u'v'})$ , vanishes near the solid wall. Equation 2-4 is also known as the Outer Law.

Additionally, Wyganski et al. (1992) scaled the flow by the wall jet momentum flux,  $M_j = U_j^2 b$ , and compared his results with those of Narasimha et al. (1973). In their separate works this scaling was shown to be independent of  $Re_j$ , for large values of the same parameter. Wyganski determined the threshold to be  $Re_j > 5000$ . For the present work  $Re_j$  values ranged from 43,300 to 368,000 which fall well above the threshold value. Both researchers used the same form of the following power law:

$$U_m \frac{v}{M_j} = A_u \left( X_L \frac{M_j}{v^2} \right)^n, \quad (2-5)$$

where  $X_L$  is distance in the flow direction measured from the nozzle exit to a station ( $X_L = X_{inj}$ ) for the current application, i. e., the distance from the nozzle exit to the coolant injection point). Each study produced slightly different values for the exponent  $n$ , however, and also fairly different values of the constant  $A_u$ . Wygnanski obtained  $A_u = 1.473$  and  $n = -0.472$ , while Narasimha obtained  $A_u = 4.6$  and  $n = -0.506$ . Wygnanski attributes the differences in the constants to the scatter in Narasimha's original data. The set of constant and exponent obtained by Wygnanski was used in this experiment. Further detail on how the appropriate set was selected is given in Chapter 5.

The development of the wall jet scaling by momentum flux from Wygnanski's study also included an equation for the  $Y$  location where  $\eta$  becomes half of its original value. This equation takes the form

$$Y_{m/2} \frac{M_j}{v^2} = A_y \left( X_L \frac{M_j}{v^2} \right)^m, \quad (2-6)$$

where  $A_y = 1.445$  and  $m = 0.881$ .

Also from the same development, the friction velocity,  $U_\tau = (\tau_w / \rho)^{1/2}$ , can be obtained through a combination of equation (2-5) and equation (2-7)

$$\frac{\tau_w}{\rho} \left( \frac{v}{M_j} \right)^2 = A_\tau \left( X_L \frac{M_j}{v^2} \right)^k, \quad (2-7)$$

where  $A_\tau = 0.146$  and  $k = -1.07$ , which yields

$$\frac{U_m}{U_\tau} = \frac{A_u}{A_\tau^{1/2}} \left( X_L \frac{M_j}{U^2} \right)^{n-k/2} \quad (2-8)$$

Finally, substituting the respective constants and exponents gives

$$\frac{U_m}{U_\tau} = 3.85 \left( X_L \frac{M_j}{U^2} \right)^{0.063} \quad (2-9)$$

The scaling of the wall jet flow by the momentum flux provided the means to obtain the input parameters of equation (2-4) and later solve for U to generate velocity profiles. This is the main capacity the wall jet scaling development is used for in the present work.

### **3. Experimental Apparatus**

The equipment and set up described in this section were used for preliminary measurements to qualify the configuration of a rectangular nozzle and instrumented table, as well as for the collection of the data to be analyzed.

Previous research by Eckerle (1988), MacMullin (1986), and Jumper (1988) was conducted in the same facility in room 21 at WRDC/POTC, with small variations in configuration. To eliminate recirculation over the plate, due to induced flow over the settling chamber and surroundings, a long radius ASME rectangular nozzle was built and installed. An additional step was to add one 60 % open perforated plate in the settling chamber to break up the flow structure coming from the wall jet generator. The perforated plate was positioned upstream of the egg crate and honeycomb flow straightener already existent.

The complete table set up consists of a test section (flat plate), two extension tables, and an injection table. In the following subsections the flat plate and extension tables are described first, followed by the injection table, the traversing system, and finally the data acquisition system. Figures 3-1(a), (b) and (c) illustrate the complete apparatus layout.

#### **3.1 Flat Plate and Extension Tables**

The test section (flat plate) is used for the evaluation of the film cooling effectiveness from measurements of surface temperatures using embedded thermocouples (see Figure 3-2 for detailed layout). The plate is laid out into fourteen stations starting at  $x/d = 5$  to 70 in increments of 5, where  $x$  is measured from the injection point and  $d$  is the diameter of the hole. At each station



there are eight laterally embedded thermocouples. The flat plate substrate consists of a 46.04 cm x 1.549 m stainless steel plate, insulated by fiberglass, wood, and Styrofoam with thickness of 0.476 cm, 3.65 cm, 15.24 cm, respectively, placed underneath the plate. This flat plate substrate is supported by a wood structure and is positioned immediately after the injection table.

Two extension tables measuring 1.22 m x 33.3 cm and 1.22 m x 1.33 m, are used to increase the distance from nozzle exit to the injection holes to a specific downstream location where the  $Tu$  intensity level is known. These tables are inserted directly between the nozzle exit and the injection table. Data was recorded at three different free stream turbulence levels,

- $Tu = 7.3\%$  - no extension table, injection table at nozzle exit,  $X_{inj} = 34.69$  cm.
- $Tu = 12.2\%$  - one extension table inserted immediately upstream of the injection table,  $X_{inj} = 68.02$  cm.
- $Tu = 17.8\%$  - two extension tables inserted immediately upstream of the injection table,  $X_{inj} = 2.013$  m.

### 3.2 Air Supply (Wall jet)

The main stream air is supplied by a compressor rated at 300 psi and 7.5 lb./sec., and passes through a gate valve into a cylindrical settling chamber with a diameter of 0.81 m and a length of 1.52 m. Inside the settling chamber the wall jet stagnation temperature was measured by a 24 gauge iron-constantan thermocouple. The wall jet is produced over a flat plate, at the outlet of a nozzle. The nozzle is an ASME long radius rectangular nozzle with a 48.0 cm by 6.5 cm cross section and a length of 38.0 cm. This apparatus is designed to provide clean flow (absence of oil and dirt), with uniform turbulence level and velocity fields at nozzle exit, steady flow, and absence of swirl.

Side walls to ensure the two dimensionality of the flow are positioned at both sides of the plate. The 15.2 cm height side walls are much higher than the boundary layer formed up to 3 m from the nozzle exit, therefore the two-dimensionality of the main flow is guaranteed. Moreover, despite the increase of the boundary layer by the secondary flow injection, the lateral wall is still high enough to avoid interference of room air from the sides into the region where the measurements were taken.

### 3.3 Injection Table

The injection table is a 1.22 m x 37.62 cm horizontal flat surface with a total of eleven injection holes distributed between two rows, 5 holes upstream and 6 holes downstream. The row-to-row distance is  $3.64d$ , with the holes at the vertices of an isosceles triangle, as shown in Figure 3-5, and a pitch to diameter ratio of 4. The downstream edge of the second row serves as the reference line for measuring the distance from the nozzle exit to injection and from injection to each station.

The coolant flow is supplied by an A/M 32C-4 aircraft ground air conditioner unit. This unit is capable of supplying air at a constant temperature in the range of 20 to 100°F. Prior to injection, the coolant is delivered to a plenum chamber with measurements of 20 x 50 x 22 cm.

The secondary flow is injected through stainless steel tubes with an internal diameter of 1.83 cm. One end of the tubes is glued to the upper plate of the plenum chamber and the other end to the injection table, flush to the test surface. The film cooling injection tube length was established as three times the injection tube diameter. With this tube length the velocity profile of the cooling flow at the point of injection is not fully developed. This is a desirable condition since the objective is to simulate as close as possible an actual turbine application. If the tube is too long the velocity profile would be fully developed, and would influence the penetration of the cooling jet

into the main stream and the structure of the film cooling. The injection angle is  $30^\circ$  which is often used in actual turbine blades. This angle was also chosen because it facilitates comparison with results obtained by other researchers.

### **3.4 Traversing System**

The traversing system included manual adjustment of the vertical and lateral probe location. A hot-film probe and a boundary layer thermocouple were held at the same height, side by side, on a support bar, 3.55 m in length. These thermocouples were used to collect velocity and temperature data of the main stream flow at the centerline of the flat table.

### **3.5 Data Acquisition System**

The data acquisition hardware was controlled by a Hewlett Packard HP 9845C computer controller, which was connected directly through an HP-IB interface bus (IEEE-488 standard) to an 80 channel HP 3497A Data Acquisition Controller Unit and two HP 3478A digital multimeters. Velocity and turbulence intensity were calculated by the HP computer and the results stored on HP 9885M flexible disk drive, 8 in. floppy disks. The data acquisition software provided a real time display of the velocity and turbulence intensity profile.

Velocity measurements were obtained from a single hot-film sensor controlled by a Thermo-System Inc. (TSI) IFA-100 constant temperature anemometer. The anemometer RMS outputs were read by the HP 3478A multimeters. The HP 3497A controller unit read the thermocouple and DC anemometer voltages. The anemometer output was also connected in parallel to an HP 3562A Dynamic Signal Analyzer for autocorrelation integral time scale measurements. A standard model TSI 1210-20 hot-film probe was used throughout the test.

Additional equipment included a TransAmerica CEC 2500 Digital Barometer, which measured the laboratory ambient pressure; a TSI 1125 Calibrator for hot-film calibration; and a Tektronix 2430A Digital Oscilloscope to measure the frequency response and stability of the film anemometer.

## **4. Experimental Procedures and Data Reduction**

Two major types of measurements were performed during this research: preliminary measurements (diagnostic or validation) and actual data collection measurements. The following section describes procedures and data reduction for all measurement collections. Since the author did not take part in the collection of the data the following descriptions were gathered in part by reconstructing the events based on the evidence revealed by the data records, information acquired from the thesis prospectus, and by accounts from researchers present at the time the experiment was taking place. Any small deviation from actual events is purely unintentional. The preliminary measurements taken to assess the two dimensionality of the main stream flow over the flat plate are described first, followed by the velocity and temperature profiles and finally the film cooling effectiveness measurements.

### **4.1 Preliminary Measurements**

#### **4.1.1 Two Dimensionality Analysis**

An open flat plate measuring 2.4 m by 4.9 m was placed tangentially in front of a short rectangular nozzle without lateral constraints. The velocity distribution was determined at nine locations along the plate with coordinates  $x$ ,  $y$ ,  $z$ , where  $x$  is the distance from the nozzle,  $y$  is the height where the maximum velocity was achieved (local  $Y_m$ ) and  $z$  is the lateral distance from centerline. The  $x$  distances along the plate were,  $x = 22.9, 45.7, 66.7, 100.0, 133.4, 166.7, 215.9, 279.4, \text{ and } 333.4$  cm. Figure 4-1 shows the results plotted in contour and three-dimensional plots.

The lateral traverse of velocity showed that only the region very close to the centerline ( $|z| < 15$  cm) and close to the nozzle ( $x < 45$  cm), i.e. the core region, had a near two-dimensional profile. Outside this region, the distribution had no level plateau of velocity, indicating that the flow over the open flat plate was not close to two-dimensional. The lateral jet spread was determined by the locus of lateral positions where the velocity is 50 % of the local centerline velocity. The value of the jet spread angle measured without side walls was about 6 degrees, as can be seen in Figure 4-2. The results indicated that a simple flat plate is unsuitable, and side walls are required to avoid lateral jet spread and/or induction of air. These results led to the experiment to check the two-dimensionality of the flow including side walls, described in the next section.

As part of the two dimensionality check, the velocity measurements were used to calculate the turbulence intensity, integral length scale and velocity fluctuation. The turbulence intensity ( $Tu$ ) is defined as:

$$Tu = \frac{u'}{U} \quad (4-1)$$

where  $U$  is the mean velocity, and the rms velocity fluctuation ( $u'$ ) is defined as:

$$u' = \sqrt{\overline{u^2}} \quad (4-2)$$

in which  $u$  is the instantaneous velocity fluctuation. The total velocity component in the x direction ( $U_x$ ) is defined as :

$$U_x = U + u \quad (4-3)$$

The integral length scale ( $L$ ) is defined as:

$$L = UJ_u \quad (4-4)$$

where  $J_u$  is the integral time scale defined as:

$$J_u = \int_0^{\infty} \frac{U_x(t)U_x(t+\tau)}{u^2} d\tau \quad (4-5)$$

where  $\tau$  is the time increment.

Figures 4-3, 4-4, and 4-5 show the resulting turbulence intensity level, integral length scale, and velocity fluctuation in contour and three-dimensional plots. The figures show that all these parameters change considerably in the longitudinal and lateral direction.

#### 4.1.2 Two Dimensionality with Side walls

The boundary layer velocity was acquired with the previously described apparatus. The velocity fluctuation, turbulence intensity, and integral length scale were evaluated as described in the previous section. Here a long radius ASME rectangular nozzle was used, with two side walls 45 cm apart, exactly the width of the rectangular nozzle. The side wall height was 15.2 cm. This wall height assured that the flow was contained in the channel, since the maximum thickness ( $Y_m$ ) measured previously was only 5.1 cm at a distance 3 m downstream of the nozzle exit.

The results from the experiments with side walls were plotted in contour plots, and are shown in Figure 4-6. In all plots the contour lines are approximately perpendicular to the mean flow, showing that two-dimensional flow was achieved.

## 4.2 Velocity and Temperature Profiles (no film cooling injection)

Two velocity profile data sets were collected at  $U_j = 55.8$  m/s to characterize the main stream flow vertically. One set was taken at 14 stations with 30 data points each ranging from  $Y = 0.508$  cm to 17.78 cm and a  $Y$  increment of 0.254 cm. At each  $Y$  location turbulence intensity and velocity fluctuations were measured. A temperature profile was also taken at all 14 stations along the plate. A second velocity profile set was taken at 6 stations with 20 data points each ranging from 0.53 mm to 10.53 mm (did not include  $Y_m$  point) and a  $Y$  increment of 0.1 mm. The coefficient of friction ( $C_f$ ) and the friction velocity ( $U_\tau$ ) were calculated at each station for this set of profile data.

## 4.3 Film Cooling Effectiveness Tests

Surface temperature measurements were taken at 14 stations with coordinates  $x, y, z$ , where  $x$  is the longitudinal distance from injection,  $y$  is the vertical distance ( $y = 0$ ) and  $z$  is the lateral distance from the centerline. Eight lateral measurements were taken at each station from which an average  $T_w$  was calculated. Using  $T_w$  average and the corresponding free stream temperature at  $x, y, 0$ , a laterally averaged  $\eta$  was calculated. Effectiveness was obtained for five different  $U_\infty$  at injection values, 10, 18, 37, 60, and 85 m/s, at three  $Tu$  levels 7.3 %, 12.2 %, and 17.8 %, with a  $B$  range of 0.25 to 2.0. There was no heat applied to the plate in an attempt to create a near adiabatic surface condition. A correction was then applied to account for any losses still present due to conduction and radiation. This correction approach is based on similar procedures outlined by Mick and Mayle (1988) and Ligrani et. al., (1992), wherein the adiabatic film cooling effectiveness (equation 2-3) is rewritten including a correction term as



$$\eta = \frac{T_w - T_{\infty}}{T_{rc} - T_{\infty}} + \frac{q_{rad} + q_{cond}}{h_0(T_{rc} - T_{\infty})} \quad (4-6)$$

where  $T_w$  is the actual measured wall temperature and  $h_0$  is the heat transfer coefficient with film cooling. The latter parameter was not available for this experiment and was estimated using a Stanton number correlation (Kays and Crawford, 1980)

$$St_0 Pr^{0.4} = 0.0287 Re_x^{-0.2} \quad (4-7)$$

and an expression to account for the effects of free stream turbulence on the Stanton number by Simonich (1978)

$$\frac{St}{St_0} = 1 + 5Tu \quad (4-8)$$

The derivation of equation (4-6) is presented in Appendix A.

Kinematic and thermodynamic properties of the main stream and secondary flow were obtained through a mixture of direct measurements and calculations. Free stream pressure and temperature measurements were taken at centerline with a Kiel probe and boundary layer thermocouple located at  $Y_m$  where  $U_{\infty}$  is at a maximum. From these measurements the free stream velocity was calculated. Coolant temperatures were calculated from coolant tank total temperature and coolant velocity values. The coolant velocity, was calculated using the coefficient of discharge and coolant tank pressure and density values. The secondary flow injection was kept at a temperature 10 to 20 degrees lower than the free stream. Wall temperatures were registered using thermocouples imbedded in the plate. A description for the uncertainties in the measurements and the propagation of these to the calculation of the effectiveness is presented in Appendix B.

The procedure for collecting the data was the following:

1. Set  $U_{\infty}$  at injection by adjusting the pressure in the main tank
2. Set film cooling flow by adjusting the pressure in the cooling tank, which in turn sets the discharge coefficient (see Appendix C for CD correlation used )
3. When desired conditions are met, take measurements for  $T_m$ ,  $T_{rx}$ ,  $P_{kiel}$ ,  $T_{tank}$ ,  $T_{tankcool}$ ,  $P_{tank}$ ,  $P_{tankcool}$ .
4. Reduce data (for thermocouple data reduction program see Appendix C)

The above procedure was repeated for each blowing rate for a given  $U_{\infty}$  at injection value and  $Tu$  level. The blowing rate was changed by resetting  $U_c$  (adjusting to desired coefficient of discharge) while keeping  $U_{\infty}$  at injection at a set value. The density ratio was maintained at slightly above one ( $\sim 1.07$ ). After the desired range of blowing rates was covered,  $U_{\infty}$  at injection would be changed next for the same  $Tu$  level until data was collected for all blowing rates. After data for all combinations of  $U_{\infty}$  and  $B$  had been collected an extension table was inserted to relocate the injection holes to the respective longitudinal location,  $X_{inj}$ , for the next turbulence level to be tested. The turbulence level achieved at a specific  $X_{inj}$  was independent of the value of nozzle jet exit velocity set. In other words it did not matter whether  $U_{\infty}$  at injection was set to 10 m/s or 60 m/s, the turbulence level at  $X_{inj} = 2.01$  m continued to be 17.8 %. This finding was determined by McMullin et al. (1988) who also used a wall jet (round ASME nozzle) as a source of free stream turbulence during his heat transfer research.

#### 4.3.1 Reconstruction of free stream temperature data

Subsequent inspection of the free stream temperature data by this author revealed that the thermocouple used to record the free stream temperature,  $T_{reo}$ , was positioned too close to the coolant jet stream and at high blowing rates, when the jet stream penetrated further into the main

stream, the temperature being registered was that of a mixture of coolant and main stream flow, not of the main stream alone. As a result, free stream temperature values needed to be replaced in all film cooling effectiveness calculations. Since the data could not be collected again (the test apparatus had been disassembled) an alternate way to obtain free stream temperature values was devised. Using the total temperature in the main air source plenum chamber,  $T_{\text{tank}}$ , and free stream velocities, new  $T_{\text{rco}}$  values were calculated (for calculation details see Appendix D). The recovery temperature for low  $B$  values was also reconstructed because there was no way of knowing at which blowing ratio the free stream temperature measurements began to be tainted by the coolant.

The above mentioned misplacement of the free stream thermocouple could have been avoided by taking a temperature profile along the plate with injection at high and low blowing rates to locate the best position for the thermocouple for the respective blowing rate.

It was also found that the film cooling effectiveness was calculated using static temperature values instead of recovery temperature values which were verified to be necessary at the higher  $U_{\infty}$  and  $U_c$  conditions used in this experiment. Therefore, all relevant temperatures ( $T_{\infty}$  and  $T_c$ ) were converted to recovery values and  $\eta$  values recalculated.

## **5. Results and Discussion**

For this research, laterally averaged film cooling effectiveness was generated from the collection of surface and free stream temperatures acquired while injecting cooled air over an adiabatic plate through two rows of 30° slant-hole injectors. The data was obtained for five values of free stream velocities at injection from a rectangular slot (10, 18, 37, 60 and 85 m/s), three free stream turbulence levels (7.3 %, 12.2 % and 17.8 %), and blowing ratios from 0.25 to 2.0.

This chapter presents the results of the experiment and offers a discussion of the behaviors observed. The chapter is divided into two parts. The first part deals with the effect of free stream turbulence on film cooling effectiveness. The second part presents the development of a film cooling effectiveness correlation.

### **5.1 Effect of Free Stream Turbulence on Film Cooling Effectiveness**

The effects of free stream turbulence on film cooling effectiveness involve a complex process, influenced by many factors. Film cooling effectiveness is affected by the presence of turbulence in the free stream through the enhanced mixing of the coolant with the main stream air. As the cooling air is injected, turbulence in the free stream spreads the air laterally, encouraging a uniform distribution of the injecting jets, and also vertically. For flows attached to the surface this vertical spreading usually implies a decrease in effectiveness as the coolant is diffused away from the surface; but as shown later in this section, effectiveness increases with increased turbulence if the flow has lifted from the surface due to a high blowing ratio. The method and the degree to

which free stream turbulence affects the performance of film cooling is influenced by such parameters as longitudinal distance from the injection holes ( $x/d$ ), blowing ratio ( $B$ ) and free stream velocity ( $U_\infty$ ). Various combinations of these parameters create different flow regimes, and the flow regime created determines the manner in which free stream turbulence affects film cooling effectiveness.

Two major regimes were detected, one for  $B < 0.75$ , where the cooling stream remains attached to the surface of the plate at all  $x/d$  and one for  $B \geq 0.75$ , where the cooling stream lifts off from the surface of the plate and reattaches at  $x/d \leq 15$ . To facilitate the discussion of the turbulence effects on film cooling effectiveness, the sections that follow are divided according to these different regimes, specifically:

- Effectiveness variation with  $x/d$  for  $B < 0.75$  and  $B \geq 0.75$
- Local effectiveness variation with blowing rate for  $x/d \leq 15$  and  $x/d > 15$

All plots include data for  $Tu = 7.3\%$  and  $Tu = 12.2\%$ , and many include  $Tu = 17.8\%$  also. For further reference, Appendix E contains the plots of cases for specific combinations of  $B$  and  $U_\infty$  at injection that fall within the range of a particular section but are not directly referred to in the discussion.

### **5.1.1 Effectiveness Variation with $x/d$ for $B < 0.75$**

In the low range of blowing ratios  $0.25 \leq B < 0.75$ , a rapid monotonic decrease in effectiveness with  $x/d$  occurred, with lower  $Tu$  values generally yielding higher values of effectiveness, as can be seen in Figure 5-1. Generally, for low blowing ratios, high free stream turbulence has a detrimental effect as turbulent mixing diverts streamwise propagation of the lower

momentum injection (lower  $U_c$ ) away from the plate, dissipating any amount of coolant reaching the end of the plate. In their study on the interaction between the coolant jet and mainstream flow ( $B = 0.4$ ), Kohli and Bogard (1997) found that high free stream turbulence not only dispersed coolant away from the wall but also thrust free stream air through the core of the coolant jet. This exchange of free stream and coolant fluid elements started at injection and became more pronounced downstream ( $x/d > 3$ ) where elements of free stream flow penetrated through to the wall.

Only for low  $B$  ( $B \leq 0.5$ ) at extreme downstream regions, did the low  $Tu$  case not produce higher values of  $\eta$ . In Figures 5-1, 5-2 and 5-3 for  $U_\infty = 10, 37$  and  $85$  m/s respectively, the  $Tu = 12.2\%$  case achieves a value of  $\eta$  higher than the  $Tu = 7.3\%$  case starting at  $x/d$  of 30 to 50 and continuing to the end of the plate. Bons et al. (1992) found (for a single row of holes) similar effectiveness enhancement at large  $x/d$  (starting at  $x/d \approx 30$ ) for  $B = 1.47$ , where the  $Tu = 6.5\%$  case produced higher values of centerline  $\eta$  than the  $Tu = 0.9\%$  case. He attributes this enhancement to the earlier merging of the cooling jet streams at higher  $Tu$  levels. For  $B > 0.5$ , higher effectiveness with low  $Tu$  occurred to the end of the measurement surface

### **5.1.2 Effectiveness Variation with $x/d$ for $B \geq 0.75$**

As  $B$  increases above 0.75, a transition occurs between attached flow and blow-off (Figure 3), a condition where the cooling jet tends to detach at the forward stations and penetrate into the main stream. For the transition blowing rates,  $0.75 < B < 1.0$ , the higher  $Tu$  cases keep the flow attached while for the lowest  $Tu$  case some degree of blow-off is possible, indicated by the presence of maximums in Figures 5-4 and 5-5. The progression from attached flow to blow-off can also be seen in Figures 5-6, 5-7 and 5-8 for the higher free stream velocity at injection,

$U_\infty = 18$  m/s, and in Figures 5-9, 5-10 and 5-11 for  $U_\infty = 37$  m/s. As seen in these latter six figures, higher free stream velocities tend to delay the onset of blow-off to values of  $B$  nearer to unity, and always with the high  $Tu$  case the most resistive to blow-off for any velocity. Also for the transition blowing rates ( $B < 1.0$ ) higher free stream velocities result in only small changes in effectiveness values (see Figures 5-4 and 5-10).

For higher blowing ratios (at or greater than unity), Figure 5-12 ( $B = 1.25$  and  $U_\infty = 10$  m/s) is representative of the overall trend in effectiveness with  $x/d$ . For the early stations effectiveness increased, reached a maximum, with a greater maximum for the high  $Tu$  cases, and decreased as the flow moved down the plate. The region of effectiveness increase between  $x/d = 5$  and 10 is an indication that cooling flow is being reattached (brought closer to the surface) by the mixing action of the free stream turbulence. Figure 5-12 also shows that higher levels of turbulence cause an earlier reattachment of blow-off.

Once the cooling flow is reattached, indicated by a maximum, effectiveness is reduced downstream for the highest  $Tu$  flow as seen earlier for low  $B$  ( $B < 0.75$ ) (see Figure 5-12). Again, low values of effectiveness for higher  $Tu$  develop as a result of the increased vertical dissipation of the film cooling air caused by the greater mixing action. Effectiveness values continue to have approximately the same differential between high and low  $Tu$  at stations toward the end of the plate. Each of the high free stream turbulence effects, early reattachment of blow-off, and accelerated effectiveness decay, was seen at similar downstream locations ( $x/d = 10\sim 15$ ) for midline  $\eta$  by Bons et al. (1994) in their work with only one row of holes but with similar magnitudes of free stream turbulence. The effectiveness in the present study was laterally averaged, therefore similarities with midline  $\eta$  are appropriate.

For very high blowing ratios up to 2.0, the various Tu-related differences in effectiveness are greatly reduced (curves come to a near-collapse in Figure 5-13) but overall levels of effectiveness are higher for higher blowing ratios. Thus for very high blowing ratios any free stream turbulence is limited in ability to improve effectiveness during blow-off.

Finally, an increase in  $U_\infty$  did not have a significant influence on the relative or absolute effects of free stream turbulence on effectiveness discussed above. See Figures 5-5 and 5-11.

Table 5-1. Summary of  $\eta$  variation with turbulence

Low B		High B pre blow-off reattach (forward stations)		High B post blow-off attach (mid to aft stations)	
Tu $\uparrow$	$\eta$ $\downarrow$	Tu $\uparrow$	$\eta$ $\uparrow$	Tu $\uparrow$	$\eta$ $\downarrow$

### 5.1.3 Local Effectiveness Variation with Blowing Rate

#### 5.1.3.1 Forward Stations ( $x/d \leq 15$ )

A view of local film cooling effectiveness variation with B offers the opportunity to observe the effects of free stream turbulence on  $\eta$  from a different perspective. Figures 5-14 and 5-15 for  $x/d = 5$  and 10, show the variation with B for  $U_\infty = 10$  m/s. At these forward stations with  $B < 0.75$ , the coolant film is attached. (These figures are representative of what also happens at  $U_\infty = 18$  m/s.) The resulting effectiveness decreases with increased levels of turbulence. From temperature profiles at low (0.9 %) and high (17.6 %) free stream turbulence levels and injection at a low blowing rate, Bons et al. (1994) was able to verify (for a single row of holes) substantial temperature dissipation due to greater mixing for the high Tu cases.



For the transition blowing ratio and beyond,  $B > 0.75$ , high free stream turbulence increases the value of effectiveness. As stated earlier in this chapter, high blowing rate injection produces blow-off at the forward stations as the higher momentum flow penetrates deeper into the mainstream. In this situation the dispersed film is more effectively redirected towards the plate due to the greater mixing of the higher free stream turbulence level. Also, for the higher blowing ratio,  $\eta$  generally tends to increase as  $B$  is increased.

At the intermediate station,  $x/d = 15$ , for  $B < 0.75$  high free stream turbulence causes a decrease in effectiveness while a lack of sensitivity to  $Tu$  is seen for  $B > 0.75$ . Figure 5-16 shows the curves for  $Tu = 17.8\%$ ,  $Tu = 12.2\%$  and  $Tu = 7.3\%$  merged at virtually the same effectiveness values.

At higher free stream velocities the limitations of the injection equipment restricted the maximum blowing ratio to 1.0 for  $U_\infty = 37$  m/s, 0.68 for  $U_\infty = 60$  m/s and 0.5 for  $U_\infty = 85$  m/s. Higher free stream velocities, however, did not alter the influence of free stream turbulence on  $\eta$  previously discussed at lower velocities. In Figures 5-18, 5-18, 5-19, 5-20, 5-21, 5-22 for  $B < 0.75$  and with  $U_\infty = 37$ , 60 and 85 m/s, respectively, the lower  $Tu$  level generally produces the higher effectiveness value. In Figures 5-17 and 5-18 for  $B > 0.75$  the higher  $Tu$  level produces the higher effectiveness value.

Free stream turbulence did not significantly affect the optimum blowing ratio ( $B_{opt}$ ) value. At  $x/d = 5$  for  $U_\infty = 18$  m/s (see Figure 5-23),  $B_{opt}$  changed from 0.25 to 0.50 when  $Tu$  increased from 7.3 % to 12.2%. When  $Tu$  increased further to 17.8 %,  $B_{opt}$  remained at 0.50. At  $5 < x/d < 15$   $B_{opt}$  remained at 2.0 for all  $Tu$  levels. Similar insensitivity was observed for  $U_\infty = 10$ , 37, 60, 85 m/s.

### 5.1.3.2 Aft stations ( $x/d > 15$ )

At  $x/d > 15$ , Figures 5-24, 5-25, 5-26, and 5-27 suggest that the film is in contact with the surface for any blowing rate and thus experiences a larger amount of vertical dissipation at higher turbulence. (i.e., for a given blowing ratio,  $B$ , as the longitudinal distance along the plate increases the mass of coolant reaching the end of the plate decreases). Also for higher blowing ratios  $\eta$  increases but is less dramatically affected by turbulence than at low blowing ratios. For instance, in Figure 5-24 for  $x/d = 20$ , for  $B = 0.25$ ,  $\eta$  is very sensitive to  $Tu$ . As the flow moves further aft this sensitivity decreases but  $\eta$  is always higher for the low  $Tu$  case.

There are several instances at  $x/d \geq 45$  for the higher free stream velocity of  $U_\infty = 85$  m/s where higher turbulence ( $Tu = 12.2$  %) produces higher values of  $\eta$  than lower turbulence ( $Tu = 7.3$  %). This occurrence again signals the fact that at very aft stations for low  $B$ , the medium  $Tu$  level of 12.2 % seems to be optimum.

As noted, in general  $\eta$  increases with  $B$  for all  $Tu$  levels for the higher velocities, 60 m/s and 85 m/s, however towards the middle and end of the plate there are some sharp dips in  $\eta$  registered for low blowing ratio,  $B = 0.35$ . Figures 5-28 - 5-31 show the effect (data was not collected for  $Tu = 12.2$  %). For stations further aft (Figures 5-32, 5-33, 5-34) this dip in  $\eta$  becomes more pronounced with streamwise distance along the plate. At this time there is insufficient data to identify the causes for these isolated decreases in effectiveness for high free stream velocities with low blowing rates. Another uncharacteristic decrease in effectiveness occurs at  $x/d \geq 65$  for  $Tu = 7.3$  % as  $B$  approaches 0.68 for  $U_\infty = 60$  (Figure 5-35) and 0.50 for  $U_\infty = 85$  m/s (Figure 5-36). Although these decreases fall within the uncertainty band of  $\pm 0.015$  for small values of  $\eta$  and low  $B$ , the fact that these decreases continue to show up in subsequent

stations strengthens the argument that these are true representations of the physics present in the flow and not an error in data collection.

The optimum blowing ratio did not change with free stream turbulence for  $x/d > 15$ .

Figure 5-23 for  $U_\infty = 18$  m/s shows that  $B_{opt}$  remained at 2.0 for all  $Tu$  levels. Similar insensitivity was observed for  $U_\infty = 10, 37, 60, 85$  m/s.

## 5.2 Film Cooling Effectiveness Correlation

Film cooling effectiveness correlated with a grouping of film cooling parameters as

$$\eta = A_1 \text{Re}_{\delta^*}^b \left( \frac{X}{Bd} \right)^c Tu^f \quad (5-1)$$

where  $X$  is longitudinal distance measured from the injection point and  $d$  is the diameter of the injection hole. The inclusion of  $\text{Re}_{\delta^*}$  required  $U(y)$  distributions to determine  $\delta^*$  at injection. Due to a lack of velocity profiles for all conditions, two methods were used to generate the required  $U(y)$  distributions. The velocity profile correlation known as the Outer Law (equation 2-4), valid when wall jet similarity is present, was used for injection at  $x/b \geq 30$ . Here  $x$  is the longitudinal distance measured from the jet nozzle to a station and  $b$  is the height of the nozzle slot. For injection at  $x/b < 30$ , velocity profiles were available from test set up measurements and were curve fitted using a self-preserving wall jet distribution (i.e., preserving the proportionality of the distribution shape for a given flow). Even though the local velocity distributions are not fully self-preserving at these early stations, this wall jet distribution provides better profile shapes than other curve fitting distributions previously used.

During the preliminary search for an appropriate correlation grouping an effort was made to correlate effectiveness with different combinations of B, X, I and S without success. The details of this undertaking are given in Appendix F.

The following sections begin with a discussion on the film cooling parameter grouping  $Re_{\delta^*}^b[(X/Bd)]^c$ , followed by a detailed description of the two velocity profile generation methods, the displacement thickness calculation, and finally, the development of the correlation.

### 5.2.1 Effectiveness Correlation with $Re_{\delta^*}^b[(X/Bd)]^c$

The grouping  $Re_{\delta^*}^b[(X/Bd)]^c$  was used to correlate effectiveness for several reasons. The method used to generate different free stream Tu levels (wall jet) required that the starting length prior to injection be increased for increased Tu levels. The parameter  $Re_{\delta^*}$  would therefore account for the effect of the differences in displacement thickness prior to injection for various Tu levels. Furthermore, this grouping and others similar to it, with the term  $[X/(Bd)]^c$  by itself or multiplied by a form of Re, had been successfully used in other effectiveness correlations (Goldstein, Shavit, and Chen, 1965).

The Reynolds number based on displacement thickness was calculated for several  $U_{\infty}$  velocities for each Tu level. The calculation of the displacement thickness for each of these conditions required velocity profiles. As mentioned earlier, velocity profiles were not available for all of the five free stream velocities used in the experiment; therefore, an alternate way to generate velocity profiles was developed based on the only profile available, collected during preliminary test set-up for  $U_j = 55.8\text{m/s}$ . Two sets of velocity profiles for  $U_j = 55.8\text{m/s}$  enabled the method, one

at 14 station locations with a Y increment of 0.254 cm and another at 6 station locations with a Y increment of 0.1 mm.

The procedure entailed the use of the self-similarity principle for a wall jet. The use of self-similarity considerations which are appropriate for values of  $x/b > 20$  to 30 (Narasimha et al., 1973, and Wygnanski et al., 1992), excluded the generation of velocity profiles for  $Tu = 7.3\%$  and  $Tu = 12.2\%$ . Cooling injection for these two turbulence levels was at  $x/b = 5.4$  and  $x/b = 10.5$ , both of which are upstream of the region in question. Therefore, the method used to generate velocity profiles based on self-similarity applies only for  $Tu = 17.8\%$ , with injection at  $x/b = 31.7$ .

#### **5.2.1.1 Velocity profiles for film cooling injection with $Tu = 17.8\%$**

The Outer Law (equation 2-4) was evaluated for its suitability to produce velocity profiles for the present wall jet flow. To ascertain the degree of self-similarity of the flow, fourteen station velocity profiles were plotted in Outer Law variables,  $Y/Y_{m2}$  and  $(U-U_m)/U_\tau$ , for  $Y^+ > 30$  (i.e. the region where the logarithmic velocity distribution applies). The collapse of the profiles was fair except for the earliest stations,  $x/b = 5, 10$  and  $15$ , with the first of these showing the most obvious separation (see Figure 5-37).

For a closer look at the region near the wall, Figure 5-38 shows a plot of the three station profiles with the best collapse,  $x/b = 35, 41$  and  $47$ , along with the Outer Law equation. As expected, self-similarity was not exhibited by the profiles at the forward stations. However, at  $x/b = 25$  the profile seemed to collapse with the others at several Y locations. Therefore, it is safe to assume that for this wall jet self-similarity begins between  $x/b = 25$  and  $x/b = 35$ .

Based on the collapse using the Outer Law variables, and the fact that self-similarity has been shown to be independent of  $Re_j$  when plotting profiles in this manner (Wygnanski et al.,

1992), the Outer Law was considered a suitable technique to generate velocity profiles for  $x/b = 31.7$  ( $Tu = 17.8\%$ ) and different values of free stream velocity.

#### 5.2.1.1.1 Determination of $U_m$ , $Y_m$ and $U_\tau$

To use equation 2-4, the values of  $U_m$ ,  $Y_m$  and  $U_\tau$  were produced using power laws developed through the momentum flux scaling of the flow by Wygnanski et al.(1992).

Figure 5-39 shows equation 2-5 plotted using two sets of the constant  $A_u$  and the exponent  $n$ . One set corresponds to the power law given by Wygnanski et al.(1992) and the other set belongs to the power law given by Narasimha et al. (1973), who also scaled the flow by  $M_j$ . To select which set was appropriate to use for the wall jet flow in this study the equation was plotted using  $U_j = 55.8$  m/s for  $x/b = 5, 15, 25, 35, 41$ , and  $47$ , along with the respective measured profile parameters in power law variables for the same stations. The measured values lie closest to Wygnanski's power law expression for  $x/b = 35, 41$ , and  $47$ . The point for  $x/b = 15$  intercepts the power law curve, but at this early station similarity has not truly been achieved and the flow would not normally be following any structured behavior. Consequently, Wygnanski's power law was considered suitable to generate values of  $U_m$  for profiles at  $x/b = 31.7$  ( $Tu = 17.8\%$ ).

The  $Y$  location for  $U_m$  was determined using the relation  $Y_m = 0.015Y_{m/2}$ , obtained from Wygnanski's development and due to Launder and Rodi (1981). The values for  $Y_{m/2}$  and  $U_\tau$  in turn were found using equation 2-6 and 2-8, respectively, also from the same development.

These equations completed the method for generating velocity profiles at  $X_{inj}$  for the different free stream velocities. The overall procedure consisted of inputting  $U_j$  and  $X_{inj}$  into

equations (2-5), (2-6), and (2-8) to obtain  $U_m$ ,  $Y_{m/2}$  ( $Y_m = 0.15 Y_{m/2}$ ) and  $U_\tau$  respectively. These quantities would then be inputted into equation (2-4) to generate  $U(Y)$  at the specified  $X_{inj}$ .

To verify this new procedure, six station profiles at  $U_j = 55.8 \text{ m/s}$  were plotted along with equation 2-4 generated at the same  $U_j$  and  $X_{inj}$  value. Figures 5-40 and 5-41 show the match of the generated profile with the measured profile to be fairly good at  $X_{inj} = 2.30 \text{ m}$  ( $x/b = 35$ ) and near perfect at  $X_{inj} = 2.67 \text{ m}$  ( $x/b = 41$ ). This was so in spite of the fact that the calculated  $U_m$ ,  $Y_m$  and  $U_\tau$  differed from the measured values by 3.9 %, 5 % and 15 %, respectively, at  $X_{inj} = 2.30 \text{ m}$  and by 2.8 %, 4.7 % and 15.8%, respectively, at  $X_{inj} = 2.67 \text{ m}$ . Table 5-2 shows the corresponding numerical values.

Table 5-2. Wall jet flow parameters measured and calculated from power laws

	Measured values for $U_j = 55.8 \text{ m/s}$		Calculated values using power laws	
	$X_{inj} = 2.30 \text{ m}$	$X_{inj} = 2.67 \text{ m}$	$X_{inj} = 2.30 \text{ m}$	$X_{inj} = 2.67 \text{ m}$
$U_m (\text{m/s})$	31.7	29.2	30.4	28.38
$Y_m (\text{cm})$	1.8	2.1	1.7	2.0
$U_\tau (\text{m/s})$	1.12	1.03	1.33	1.23

#### 5.2.1.2 Velocity profiles for injection with $Tu = 7.3 \%$ and $Tu = 12.2 \%$

The velocity profiles for the  $Tu = 7.3 \%$  and  $Tu = 12.2 \%$  cases could not be generated using equation 2-4 which required similarity of the flow. A means to include effectiveness values in the correlation for this  $Tu$  levels was to calculate  $\delta^*$  for  $Re_{\delta^*}$  using distributions from existing velocity profiles at locations near injection. For  $Tu = 7.3 \%$  and  $Tu = 12.2 \%$ , coolant was injected at  $x = 34.69 \text{ cm}$  ( $x/b = 5.4$ ) and  $x = 68.02 \text{ cm}$  ( $x/b = 10.5$ ), respectively. One set of velocity profiles for  $U_j = 55.8 \text{ m/s}$  was measured at 14 stations, two of which were  $x = 32.3 \text{ cm}$

( $x/b = 5.03$ ) and  $x = 64.6$  cm ( $x/b = 10.06$ ). These profiles were curve fitted using a self-preserving wall jet distribution of the form:

$$U = \frac{\tau_0^{1/2}}{k} \left[ \log \left( \frac{z}{z_0} \right) - \frac{z}{(\eta_1 l_0)} \right], \quad (5-2)$$

where,  $\eta_1 l_0 = Y_m$ ,  $k$  is Von Karman's constant equal to 0.41 (White, 1991),  $\tau_0$  is the kinematic wall stress (Townsend, 1976) and  $z$  is vertical distance from the wall. The parameters  $\tau_0$ , and  $z_0$ , were calculated using points from the velocity profile measured for  $U_j = 55.8$  m/s (see Appendix G for sample calculations).

Equation (5-2) applies to the inner layer of the wall jet, that is, the region from the surface of the plate to  $Y_m$ , where the maximum velocity ( $U_m$ ) is reached. For both locations  $x/b = 5.03$  and  $x/b = 10.06$ , the fit of the self-preserving distribution to the velocity profile was fair at the  $Y$  locations closest to  $Y_m$  and deteriorated at locations closer to the wall. Although, for  $x/b = 5.03$  this deviation of the velocity profile from the self-preserving distribution at the near-wall region was greater (see Figures 5-42 and 5-43). It should be noted that the fact that these velocity profiles were measured at  $Y$  increments of 0.254 cm contributes to the inaccuracy of the fit.

Effectiveness values for  $Tu = 7.3$  % and  $Tu = 12.2$  % used in the correlation were not directly measured. The free stream velocity settings at injection for which effectiveness measurements were taken did not include the  $U_\infty$  at injection values of 51.95 m/s and 54.56 m/s corresponding to  $x/b = 5.03$  and  $x/b = 10.06$  when  $U_j = 55.8$  m/s. Therefore, the effectiveness at these two  $U_\infty$  values used in the correlation was obtained from plots of  $\eta$  with  $U_\infty$  (See Figure 5-44 for an example plot).



### 5.2.1.3 Displacement Thickness Calculation

From the velocity profiles obtained, displacement thickness at injection was calculated for  $U_\infty = 10$  m/s, 18 m/s, and 60 m/s (nominal  $U_\infty$  values at  $X_{inj} = 2.01$  m) with  $Tu = 17.8$  %; for  $U_\infty = 54.46$  m/s with  $Tu = 7.3$  %; and  $U_\infty = 51.95$  m/s with  $Tu = 12.2$  %.  $U(Y)$  profiles generated with the Outer Law were valid only as low as  $Y^+ > 30$  ( $Y^+ = (YU_\tau)/\nu$ ); therefore, to calculate displacement thickness down to  $Y^+ = 0$  an approximation by Reichardt (Bejan, 1984)

$$\frac{U}{U_\tau} = 2.5 \ln(1 + 0.4y^+) + 7.8 \left( 1 - e^{-\frac{y^+}{11}} - \frac{y^+}{11} e^{-0.33y^+} \right) \quad (5-3)$$

was used to generate profiles for this very thin region. The addition of this small region was done for completeness; the actual contribution to the displacement thickness was relatively small (an average of 13.2 % of  $\delta^*$ ). Calculated values of displacement thickness and  $Re_{\delta^*}$  are presented in the following tables.

TABLE 5-3. Input parameters and results with Outer Law equation ( $Y$  at  $Y^+ = 30 < Y < Y_m$ ) and Reichardt's approximation ( $0 < Y < Y$  at  $Y^+ = 30$ )

$X_{inj}$ (m)	$U_i$ (m/s)	$Tu$ (%)	$U_\infty$ (m/s)	$Y_m$ (cm)	$U_\tau$ (m/s)	$\delta^*$ (cm)	$Re_{\delta^*}$
2.01	18.93	17.8	10.34	1.97	0.521	0.2635	1816
2.01	29.96	17.8	16.63	1.77	0.792	0.2236	2479
2.01	94.4	17.8	56.43	1.34	2.32	0.1459	5489

TABLE 5-4. Input parameters and results with self-preserving wall jet distribution equation ( $0 < Y < Y_m$ )

$X_{inj}$ (cm)	$U_i$ (m/s)	$Tu$ (%)	$U_o$ (m/s)	$\tau_o$ (m <sup>2</sup> /s <sup>2</sup> )	$\eta_{i,o}$ (cm)	$z_o$ (cm)	$\delta^*$ (cm)	$Re_{\delta^*}$
68.02	55.8	12.2	51.95	7.79	2.032	$7.373 \times 10^{-11}$	0.1402	4849
34.69	55.8	7.3	54.46	1.79	1.778	$2.150 \times 10^{-10}$	0.1296	4720

### 5.2.2 Development of Correlation using $Re_{\delta}^b [X/(B\delta)]^c Tu^f$

A discrete least square approximation with two independent variables was performed to find the constant  $A_1$  and the exponents  $b$  and  $c$  (see equation 5-1), using effectiveness values from the first eight cases listed in Table 5-5. These cases all have the same  $Tu$  level. The goal was to find coefficients for one  $Tu$  level, then introduce the other two levels and observe whether any separation in the collapse occurred. If separation of the collapse was apparent then the value of the exponent  $f$  for the turbulence term would be sought.

TABLE 5-5. Listing of conditions included in effectiveness correlation with  $Re_{\delta}^b [X/(B\delta)]^c Tu^f$

Case	$U_{\infty}$ (m/s)	B	Tu (%)
1	56.43	0.25	17.8
2	56.43	0.50	17.8
3	56.43	0.75	17.8
4	10.34	0.50	17.8
5	10.34	0.75	17.8
6	16.63	0.25	17.8
7	16.63	0.50	17.8
8	16.63	0.75	17.8
9	54.46	0.25	7.3
10	54.46	0.50	7.3
11	54.46	0.75	7.3
12	51.95	0.25	12.2
13	51.95	0.50	12.2
14	51.95	0.75	12.2
15	10.34	1.75	17.8
16	16.63	1.50	17.8

The approximating function in two variables is arrived at by taking the natural log of

$$\eta = A_1 \text{Re}_{\delta^*}^b \left( \frac{X}{Bd} \right)^c \quad (5-3)$$

which gives

$$\ln \eta = \ln A_1 + b \ln \text{Re}_{\delta^*} + c \ln \left( \frac{X}{Bd} \right) \quad (5-4)$$

and assigning the corresponding term to the form

$$y = a + bx + cz \quad (5-5)$$

so that the dependent variable is assigned as  $y = \ln \eta$ , the two independent variables are assigned as  $x = \ln \text{Re}_{\delta^*}$  and  $z = \ln(X/Bd)$ , and the coefficients are assigned as  $a = \ln A_1$ ,  $b = b$  and  $c = c$ .

The approximation produced values of  $b = 0.089$ ,  $c = -0.613$  and constant  $A_1 = 0.596$ . The exponent  $c$  was of the same order of magnitude as those found by Seban (1960), Scesa (1954), Wieghardt (1946), Nishiwaki et al. (1961), and Goldstein et al. (1965), whose exponent  $c$  ranged from -0.5 to -0.88 in film cooling effectiveness correlations without free stream turbulence including the term  $[X/(Bd)]^c$ . In the correlations by Nishiwaki et al. and Goldstein et al. the term  $[X/(Bd)]^c$  was multiplied by  $\text{Re}_x^b$  and  $\text{Re}_{\delta^*}^b$ , respectively. The exponent  $b$  in these correlations was also less than one as it was for this study. A plot of the parameters showed a grouping of data points. Although the grouping was fairly good, closer examination revealed that it was fragmented into three distinct turbulence levels. The turbulence term was then included and the exponent  $f$ , along with a new constant  $A_1$  were found through iteration. The resulting correlation is:

$$\eta = 0.345 \text{Re}_{\delta^*}^{0.089} \left( \frac{X}{Bd} \right)^{-0.613} Tu^{-0.3} \quad (5-6)$$

Figure 5-45 shows a plot of  $\eta$  for cases 1-14 in Table 5-5 along with the correlation equation (5-6). The blowing ratios used were all less than unity because most of the previous correlations referenced were successful in this blowing ratio range. The effectiveness correlates fairly good with the major scatter towards the end of the plate. The effectiveness values which lie closest to equation (5-6) are generally the cases for  $B = 0.50$  and  $B = 0.75$  with  $Tu = 12.2\%$  and  $Tu = 17.8\%$

The possibility of extending the correlation to blowing ratios greater than unity was explored by including the last two cases in Table 5,  $B = 1.50$  and  $B = 1.75$ . Figure 5-46 shows that these high blowing ratio cases fell practically on the correlation line, with the exception of the blow-off region of the data ( $x/d < 15$ ). As it stands, equation (5-6) is considered to be a suitable correlation for  $0.5 < \lambda < 1.0$ , where  $\lambda = Re_{\delta^*}^{0.089} [X/(Bd)]^{-0.613} Tu^{-0.3}$ . This interval excludes the blow-off region of the flow, where the flow may need to be modeled perhaps by a momentum flux term to account for the higher momentum injection at high  $B$ ; and the far downstream region of the flow, where uncertainties were largest when measuring small values of effectiveness for low  $B$ .

Another possibility of why the far downstream region of the correlation shows a greater separation is that in this region the turbulence present in the flow was at a somewhat higher level than at injection. Perhaps as high as 20 % judging from the 2-D validation test contours. Substituting a more representative level of  $Tu$  in the correlation for stations far from injection may produce a better collapse of the data at these far downstream stations. Unfortunately,  $Tu$  contours for the corresponding test runs from which an accurate turbulence increase could be extracted were not available.

For comparison purposes effectiveness values from three separate works, Bons (1992), Jumper (1988) and Jabbari et al.(1978), were included in Figure 5-47. Bons' effectiveness values for only one row of holes with  $Tu = 11.5\%$  fell under the correlation region; this result was

expected since one row of holes has been shown to produce lower effectiveness (Han and Mehendale, 1986, Cirriello, 1991). Jumper's effectiveness values also for one row of holes with  $Tu = 17\%$  fell within the grouping (except for a few very high values in the region close to injection). This result was unexpected since the previous one row values fell well below the correlation region. Although Jumper's hole configuration did have both a smaller  $P/D$  ratio and injection angle than Bons' configuration; it is not entirely clear whether those differences would account for such a large increase in effectiveness. Jabbari's effectiveness for two rows of holes, with free stream turbulence of less than  $1\%$  and  $B = 0.50$  and  $B = 0.20$ , fell on the correlation region as expected.

## 6. Conclusions

The objective of the present research was to investigate how the blowing ratio influences the effects of free stream turbulence on the film cooling effectiveness. Free stream and surface temperature measurements were taken along a flat plate for three different Tu levels, five free stream velocities and a blowing ratio ranging from 0.25 to 2.0.

The method by which free stream turbulence affects effectiveness was strongly dominated by two main factors: blowing ratio and location of the flow along the plate. For high blowing ratios at locations soon after injection high free stream turbulence levels counteract the effects of blow-off. The greater mixing action with high free stream turbulence forces detached coolant elements back down towards the surface of the plate increasing the effectiveness.

At mid to aft stations for all blowing ratios high free stream turbulence generally decreases the effectiveness. In these locations where the coolant stream is attached to the surface the mixing action of the higher turbulence dissipates the coolant away from the surface.

Far downstream of injection where the lowest Tu level (7.3 %) is expected to achieve the greater value of effectiveness, the medium Tu level (12.2 %) produced the highest values instead. This occurrence suggested that far downstream there may be an optimum level of free stream turbulence which produces the best effectiveness.

Free stream velocity was found not to have a significant influence on the impact of free stream turbulence on effectiveness. The main outcome of varying the free stream velocity was a change on the location a particular trend would take effect. For instance, if  $\eta$  began to increase with blowing rate as the flow moved further downstream, this trend change would begin to take place further upstream as  $U_\infty$  was increased. Another more subtle effect was the retardation of the

onset of blow-off for high  $B$  as  $U_\infty$  increased. Principally, changes in  $U_\infty$  did not change how free stream turbulence affected the film cooling effectiveness.

A correlation of effectiveness with  $Re_\delta^b (X/Bd)^c Tu^f$  is offered which incorporates all  $Tu$  levels used. Effectiveness correlated best at 10 to 15 diameters after injection where blow-off effects are no longer present, and before the very far downstream stations where uncertainties in the measurements were greater.

## Recommendation

An optimum level of free stream turbulence seemed to be evident at far downstream regions of the flow. Studies focused on the flow structures of the coolant-mainstream interaction at large  $x/d$  may help identify the turbulence qualities which produce higher effectiveness at these stations.

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## 8. Figures Referenced in Main Document

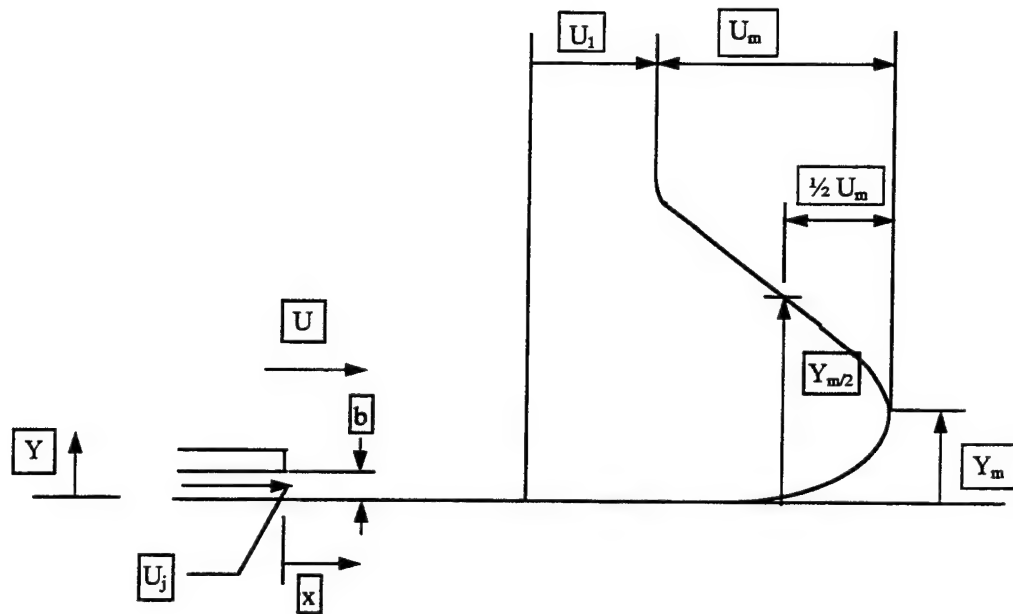


Figure 2-1. Turbulent Wall Jet Configuration

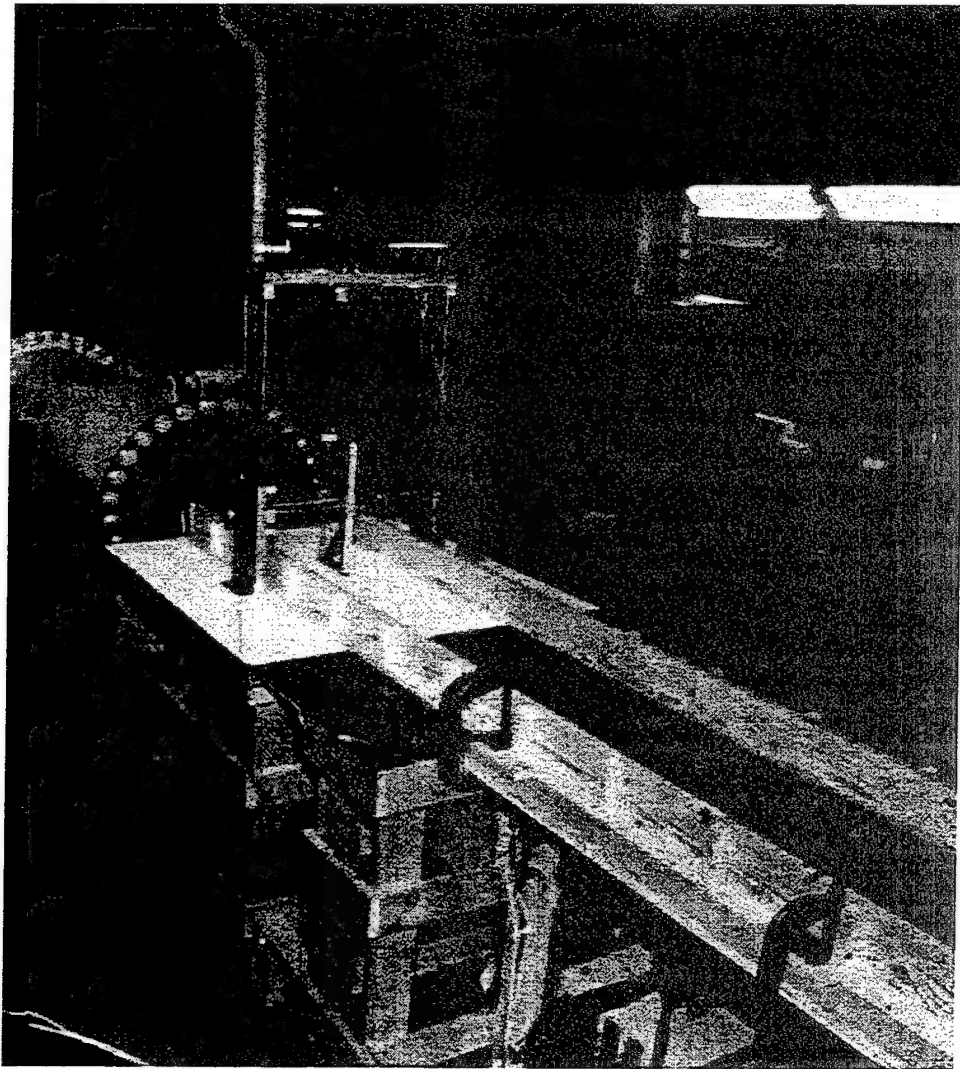


Figure 3-1. Complete Apparatus Layout (view a)

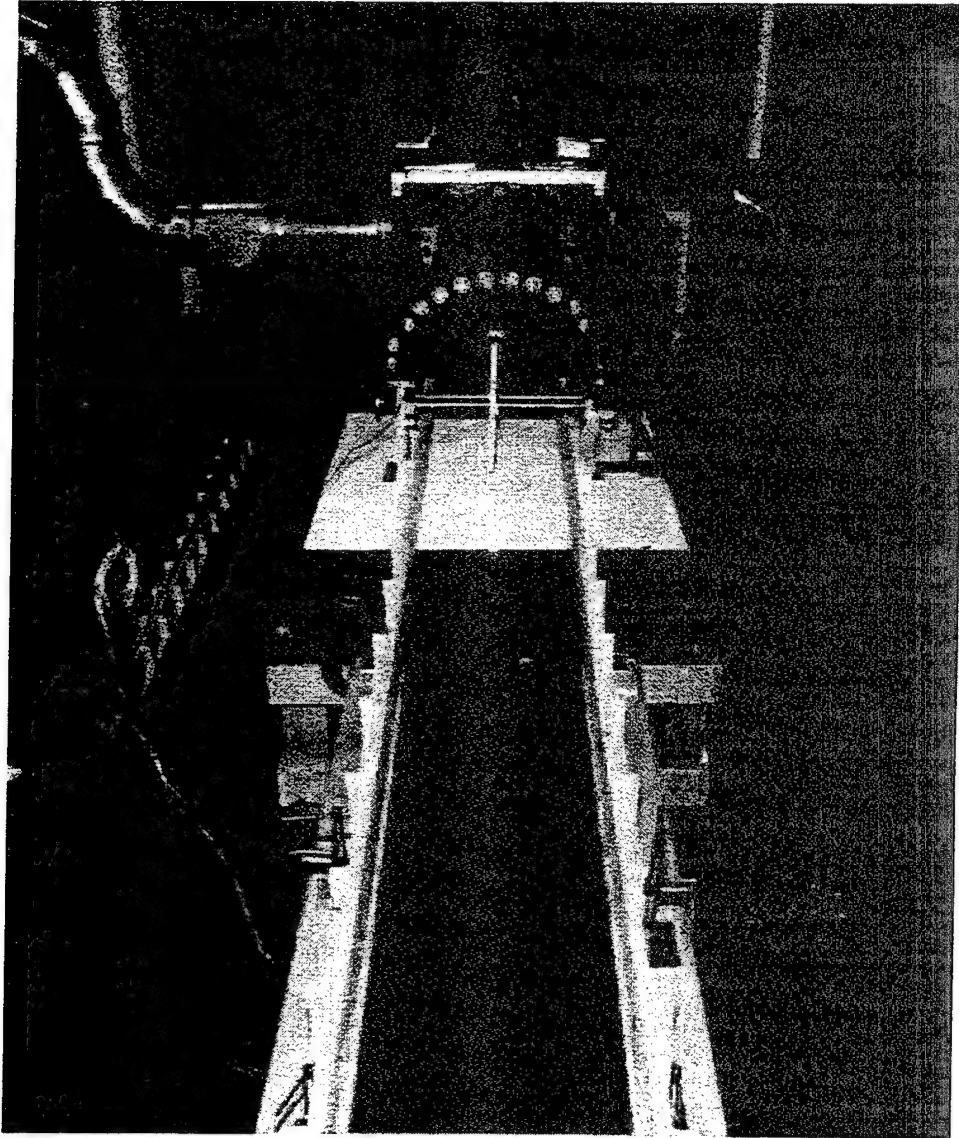


Figure 3-2. Complete Apparatus Layout (view b)

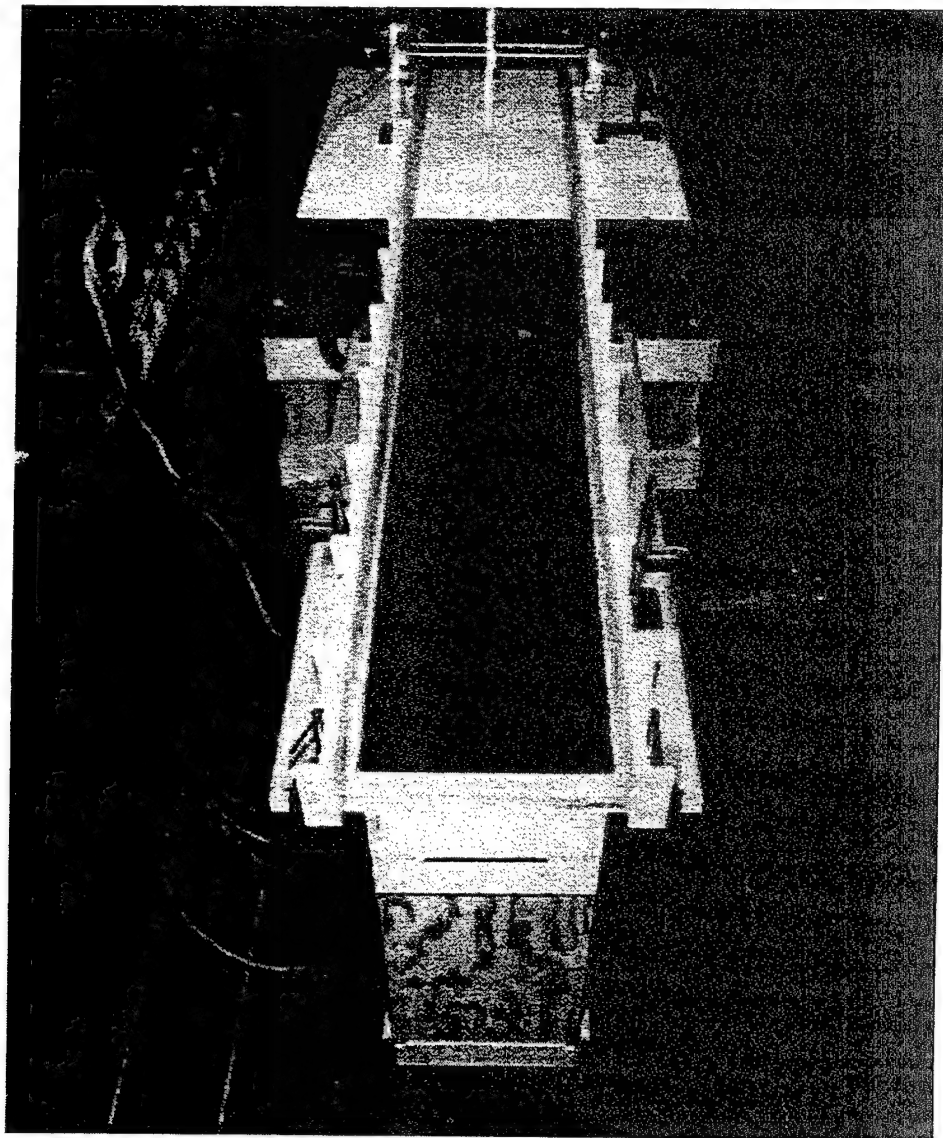


Figure 3-3. Complete Apparatus Layout (view c)

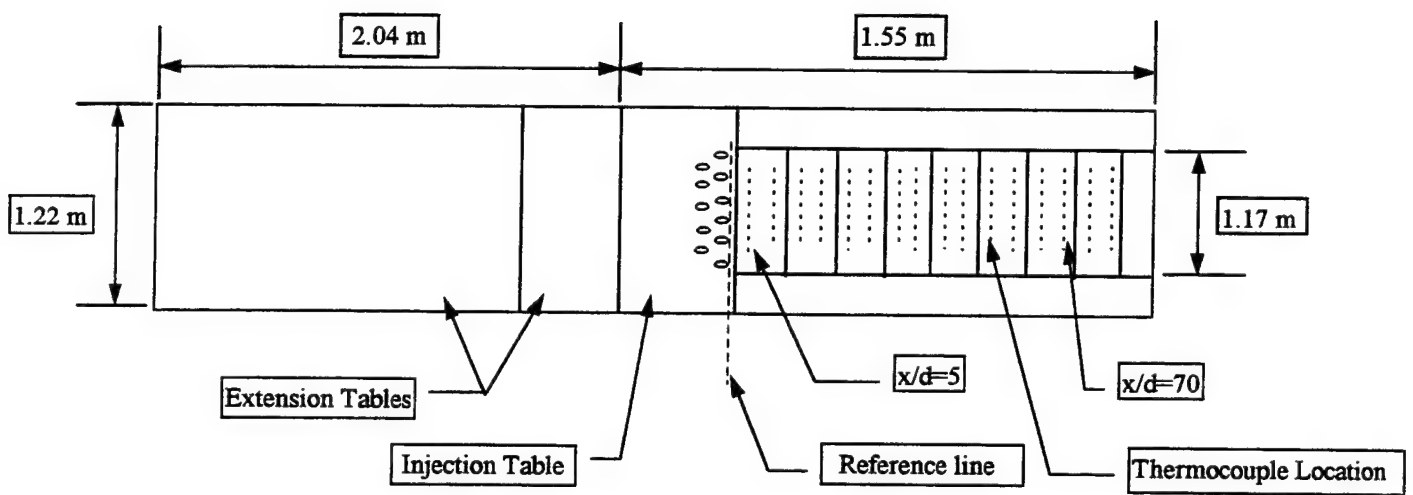


Figure 3-4. Thermocouple and Table Layout

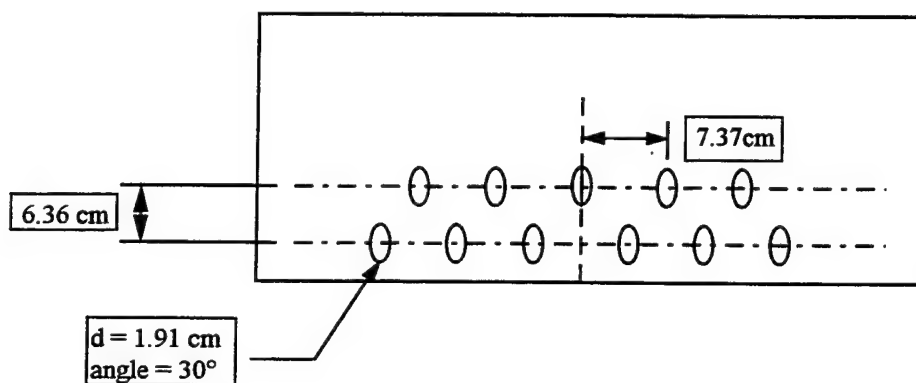
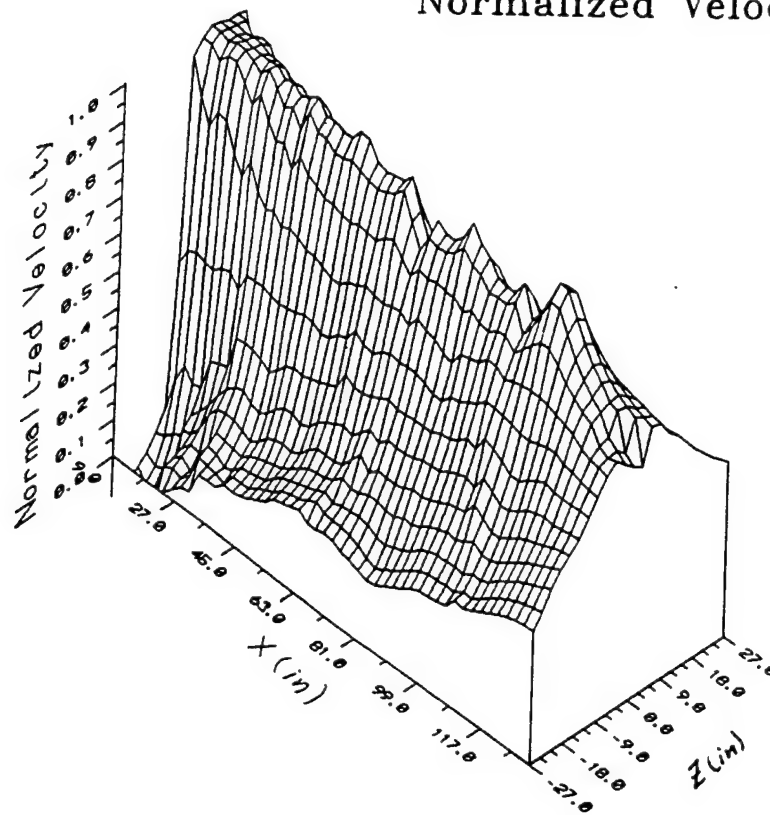


Figure 3-5 Injection Section of Flat Plate

## Normalized Velocity Profile



## Normalized Velocity Contour

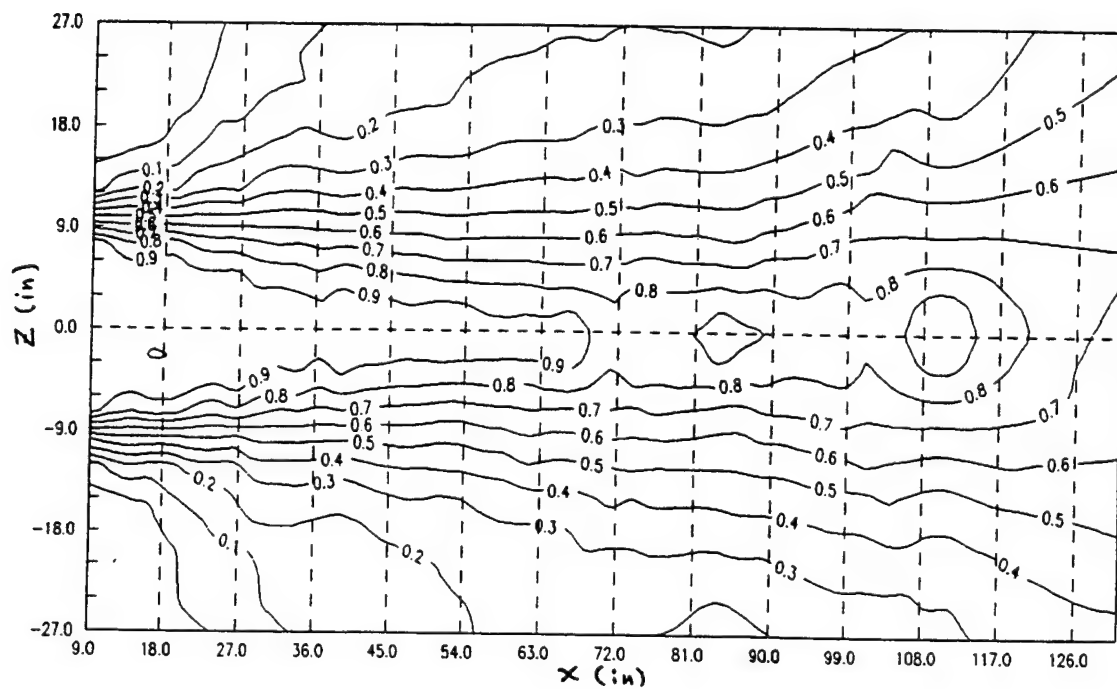


Figure 4-1. Velocity Results for Wall Jet without Side Walls



## Normalized Velocity Contour

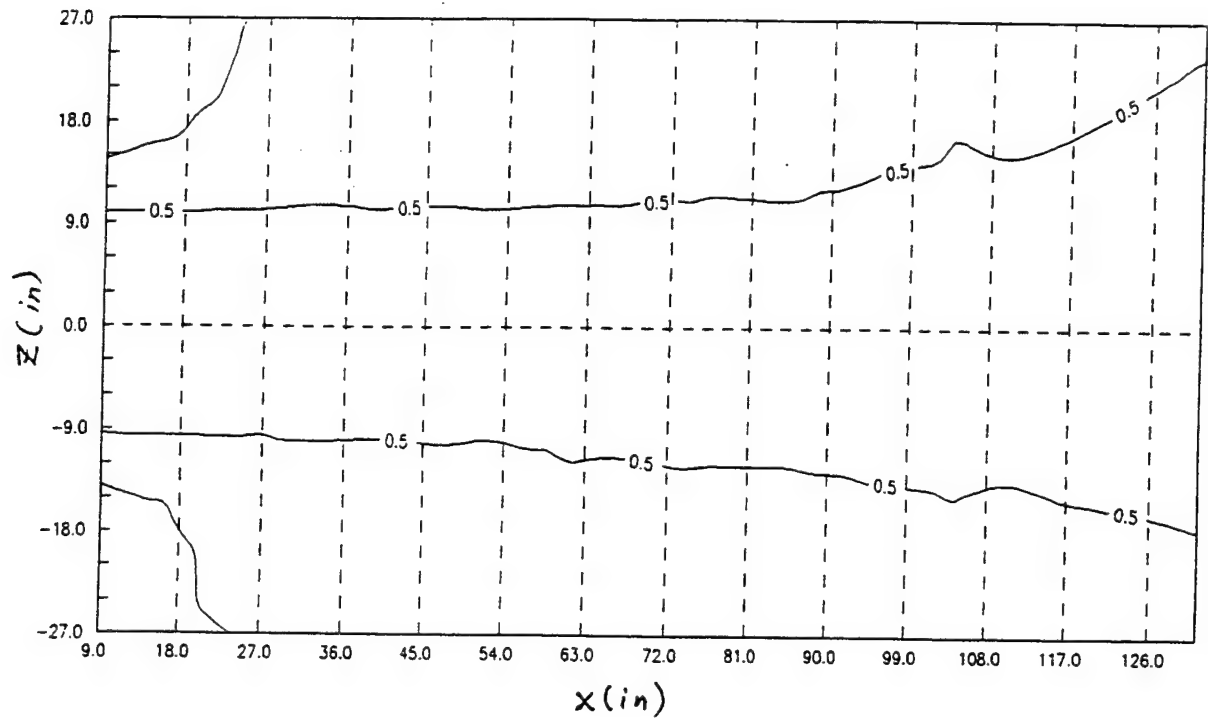


Figure 4-2. Contour Plot of Lateral Jet Spread

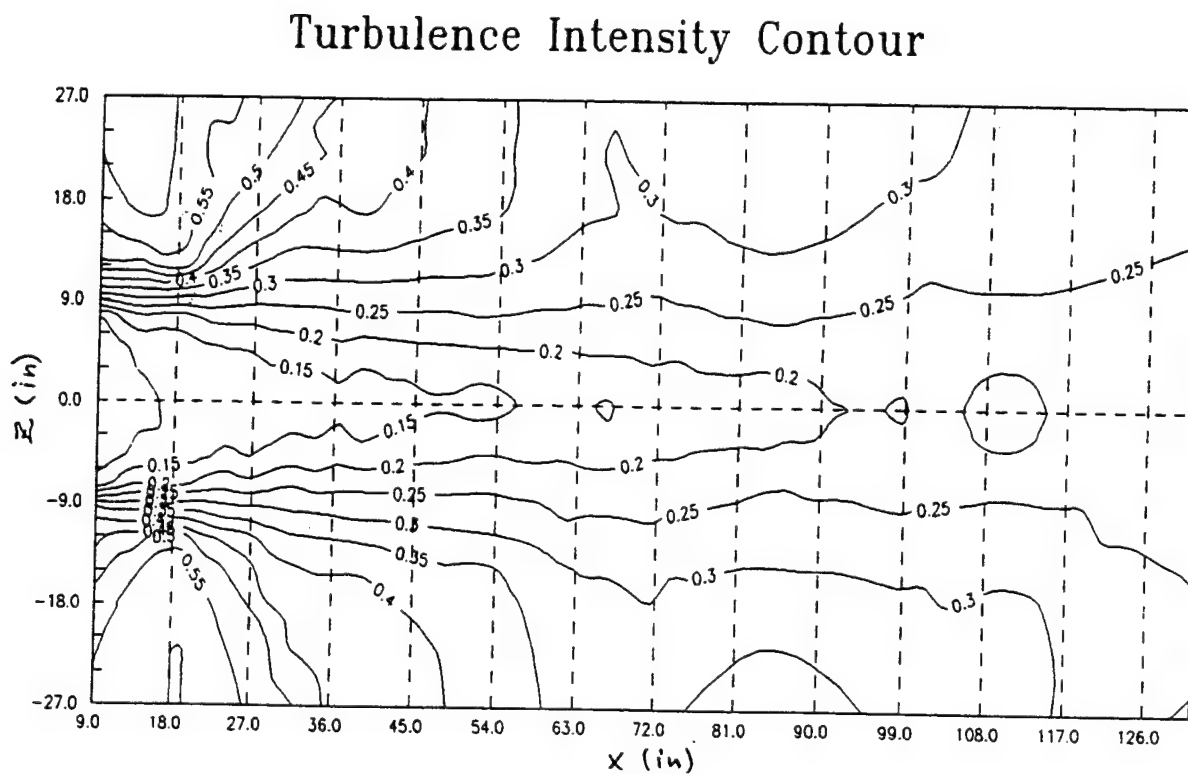
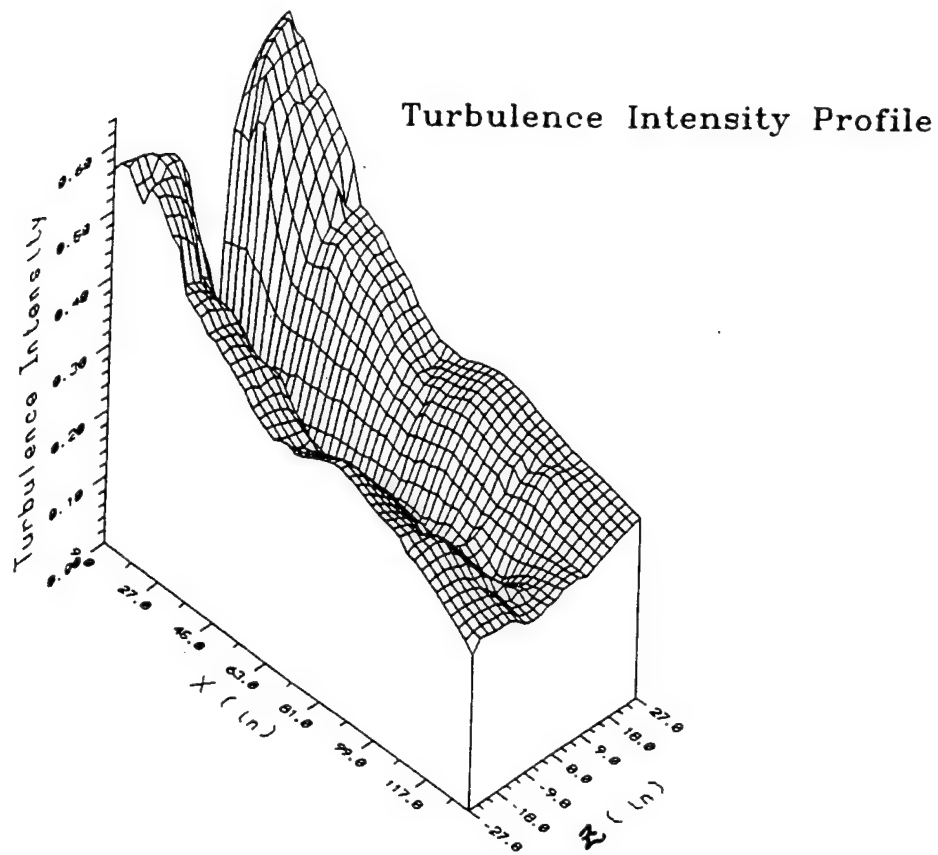


Figure 4-3. Turbulence Intensity Level Contour without Side Walls

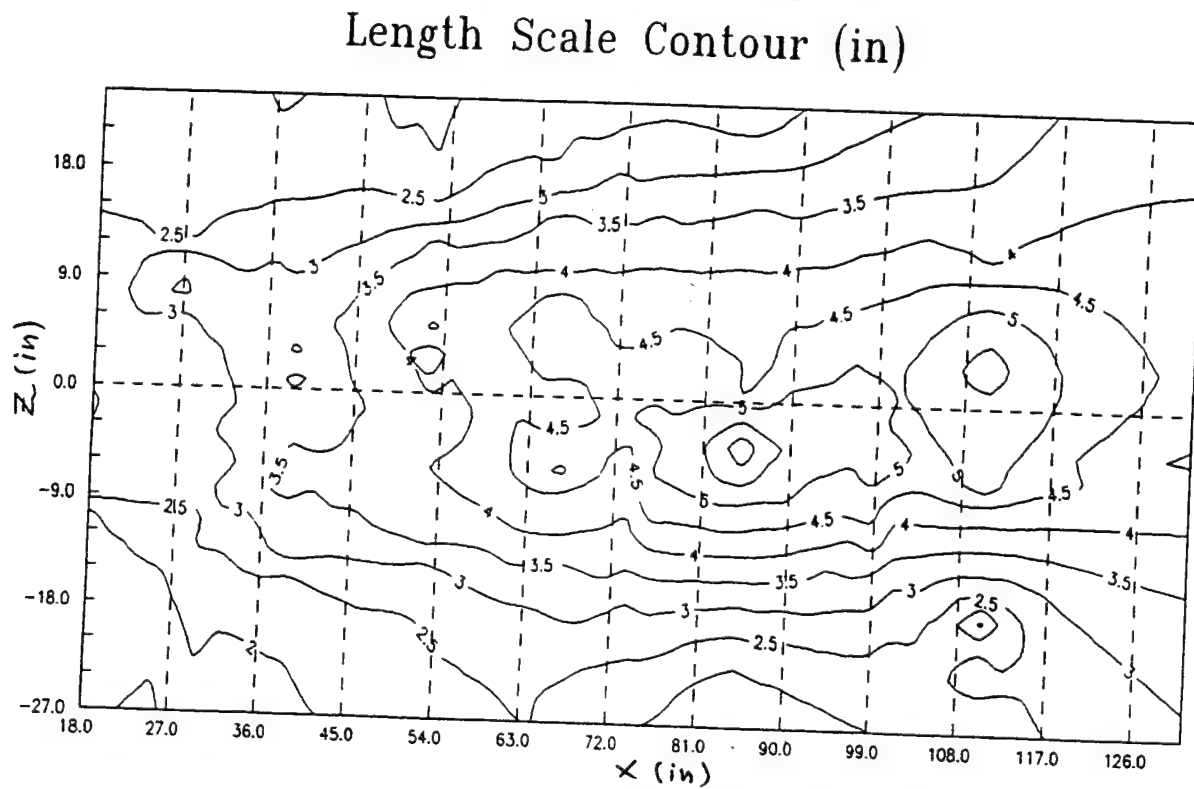
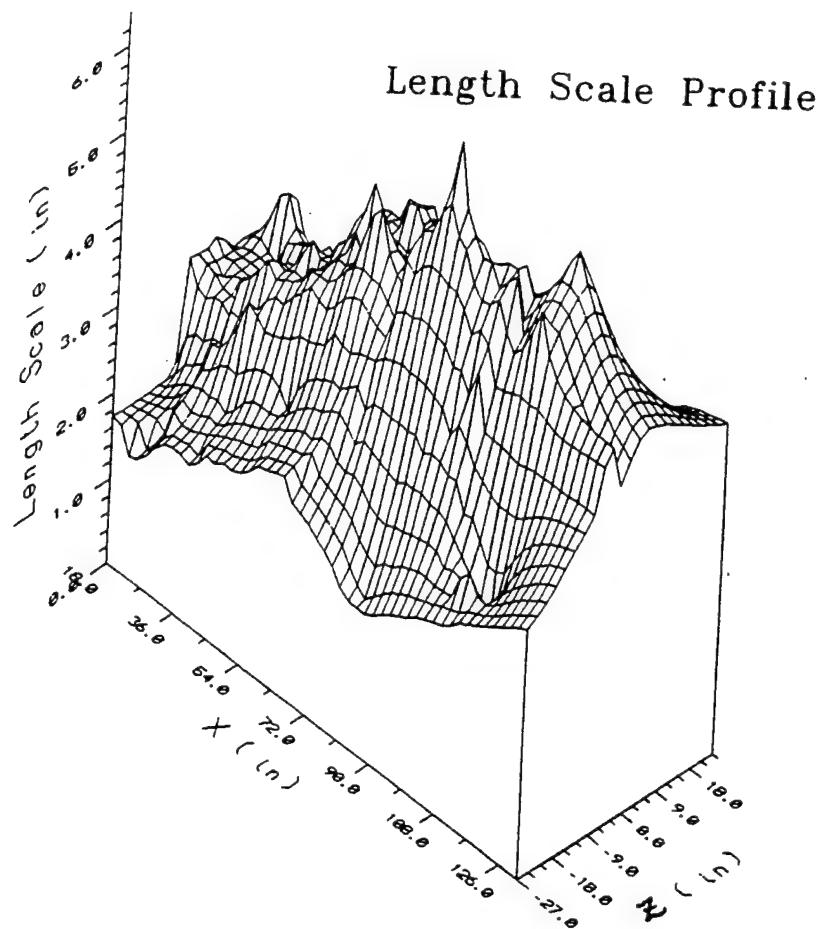


Figure. 4-4. Integral Length Scale Contour without Side Walls

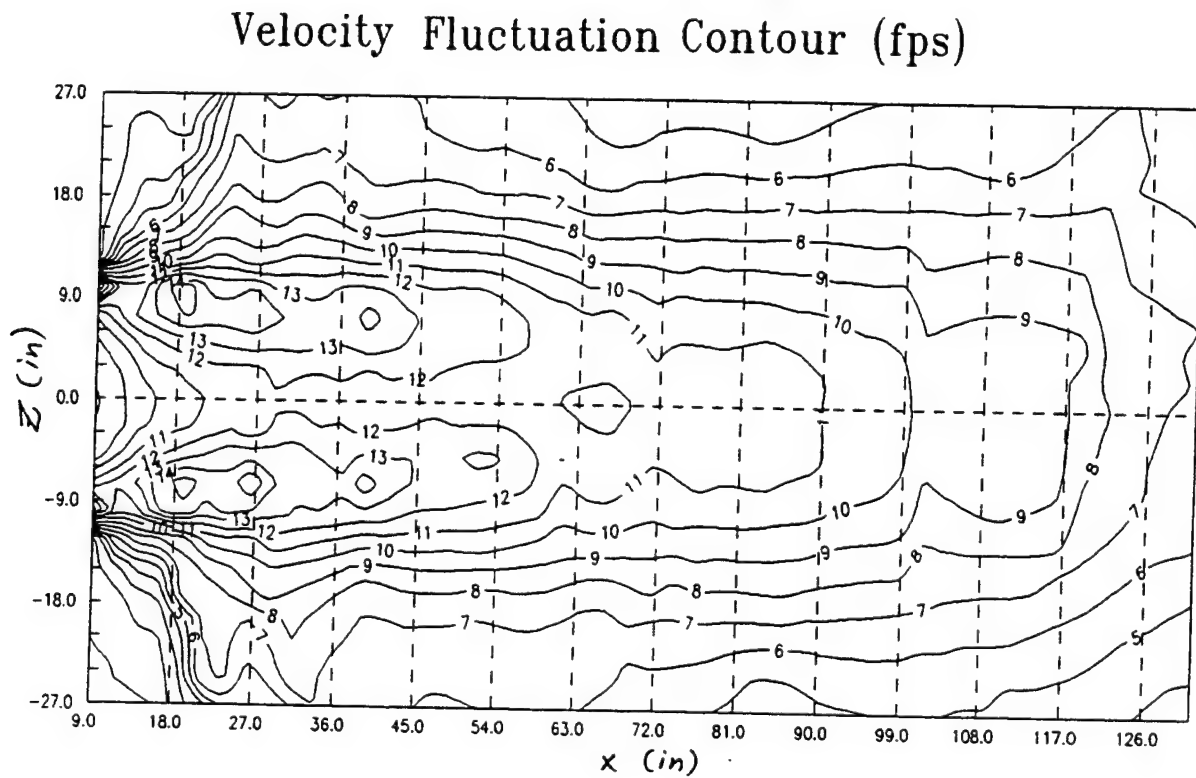
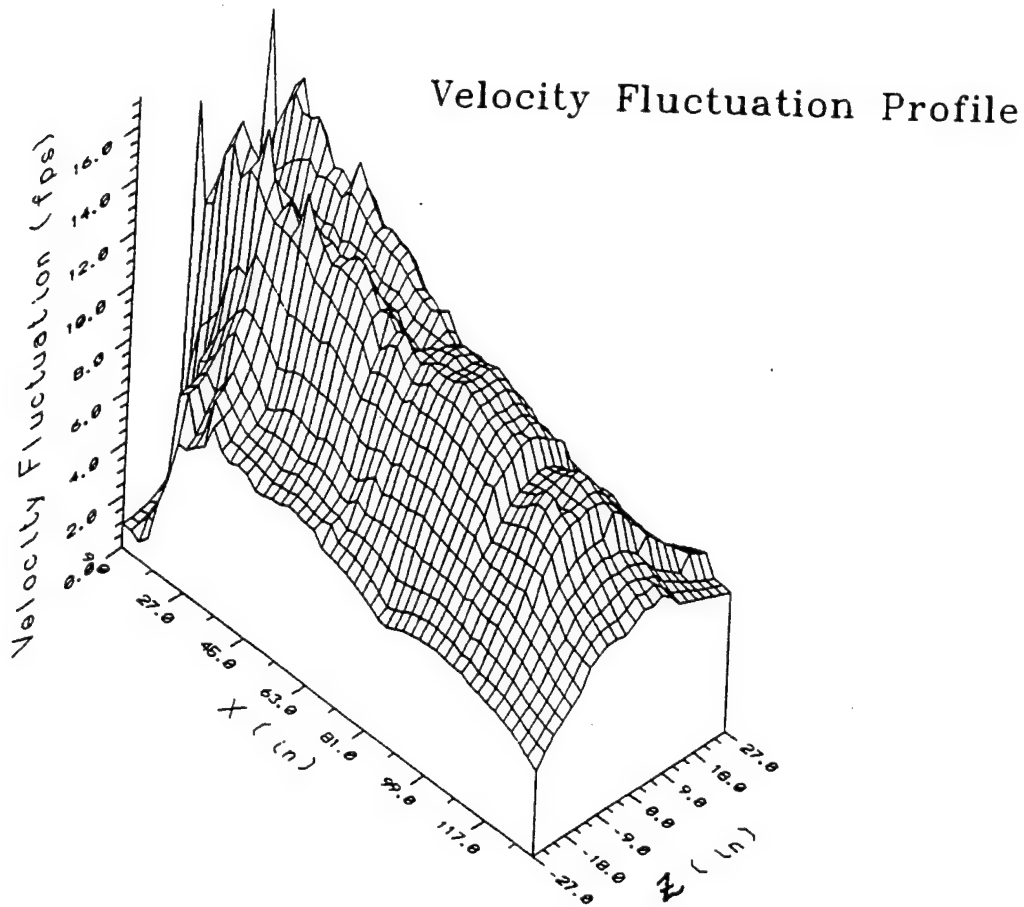
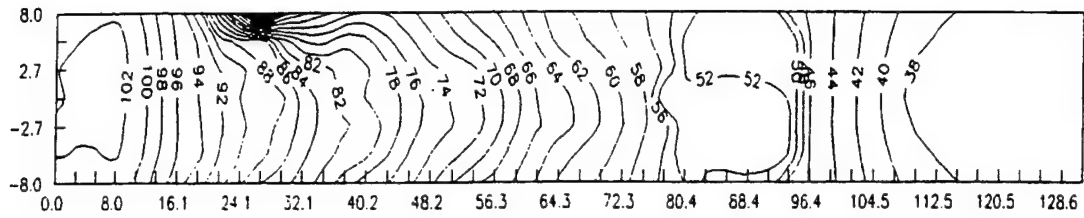
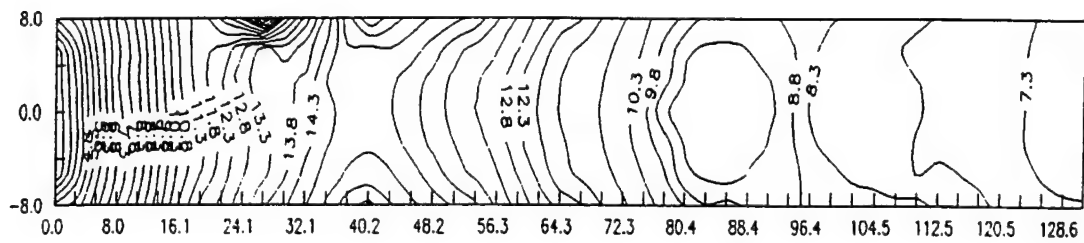


Figure 4-5. Velocity Fluctuation Contour without Side Walls

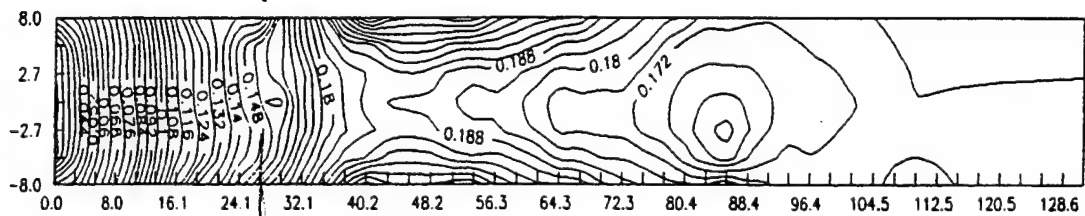
### Velocity Contour (fps)



### Velocity fluctuation (fps)



### Turbulence Contour



### Length scale (in)

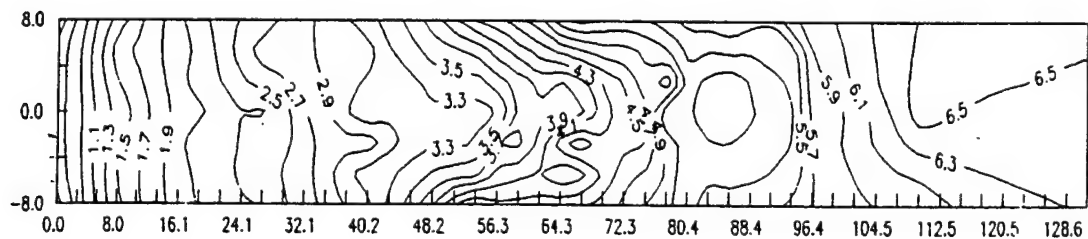


Figure 4-6. Results of Experiments with Side Walls

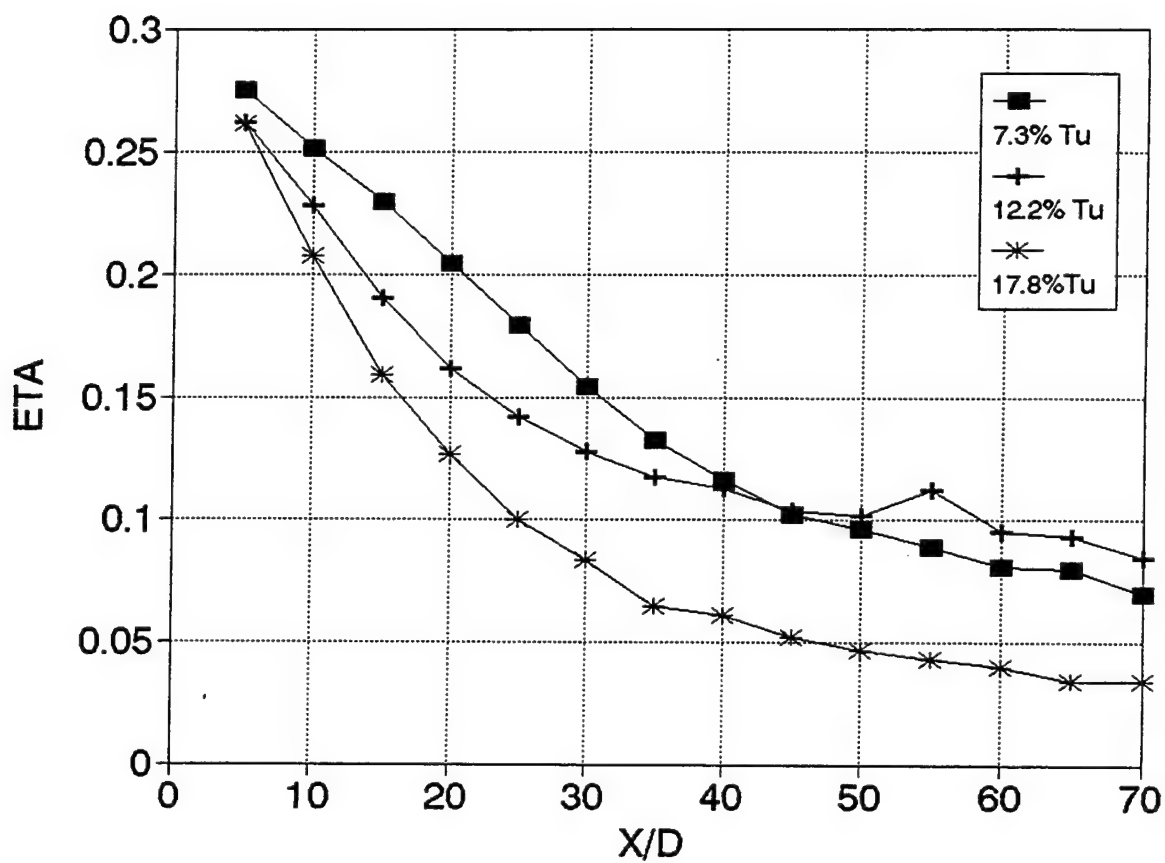


Figure 5-1. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.50$ , Free stream velocity at injection = 10 m/s

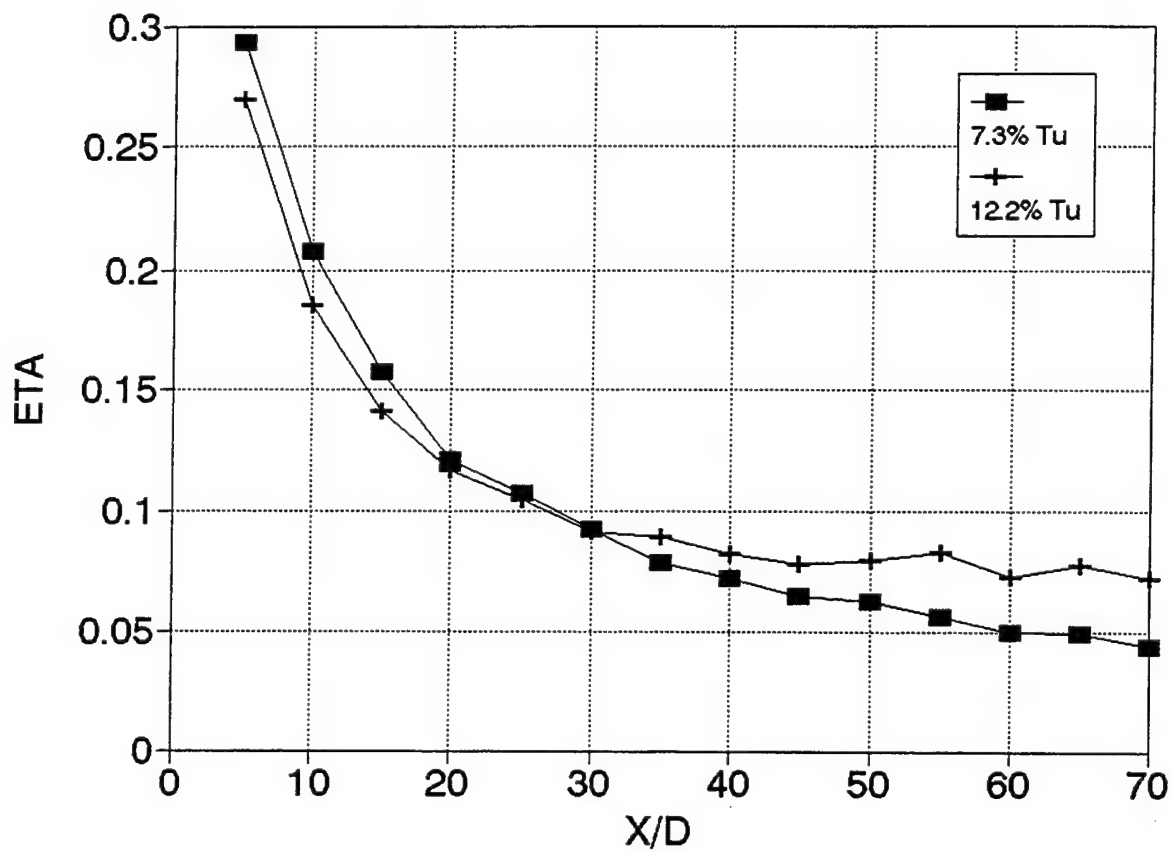


Figure 5-2. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.25$ , Free stream velocity at injection = 37 m/s

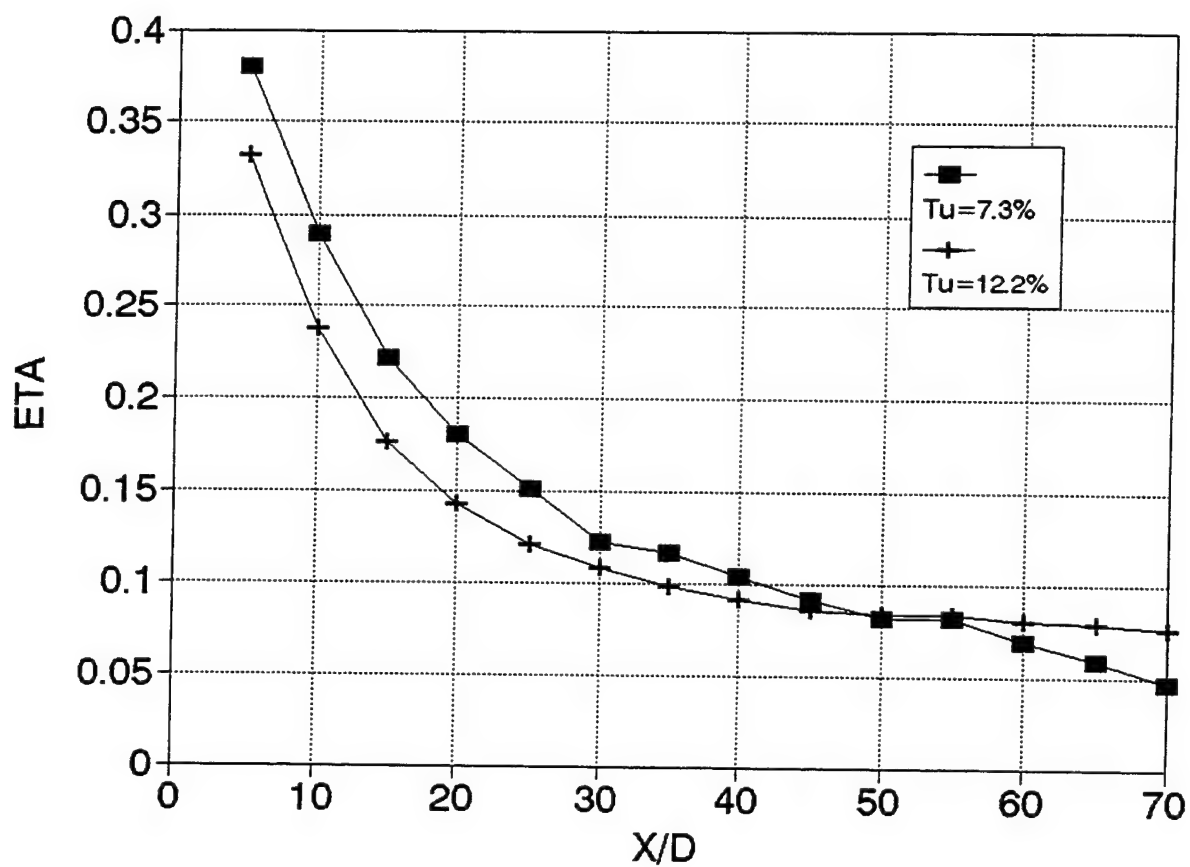


Figure 5-3. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.50$ , Free stream velocity at injection = 85 m/s



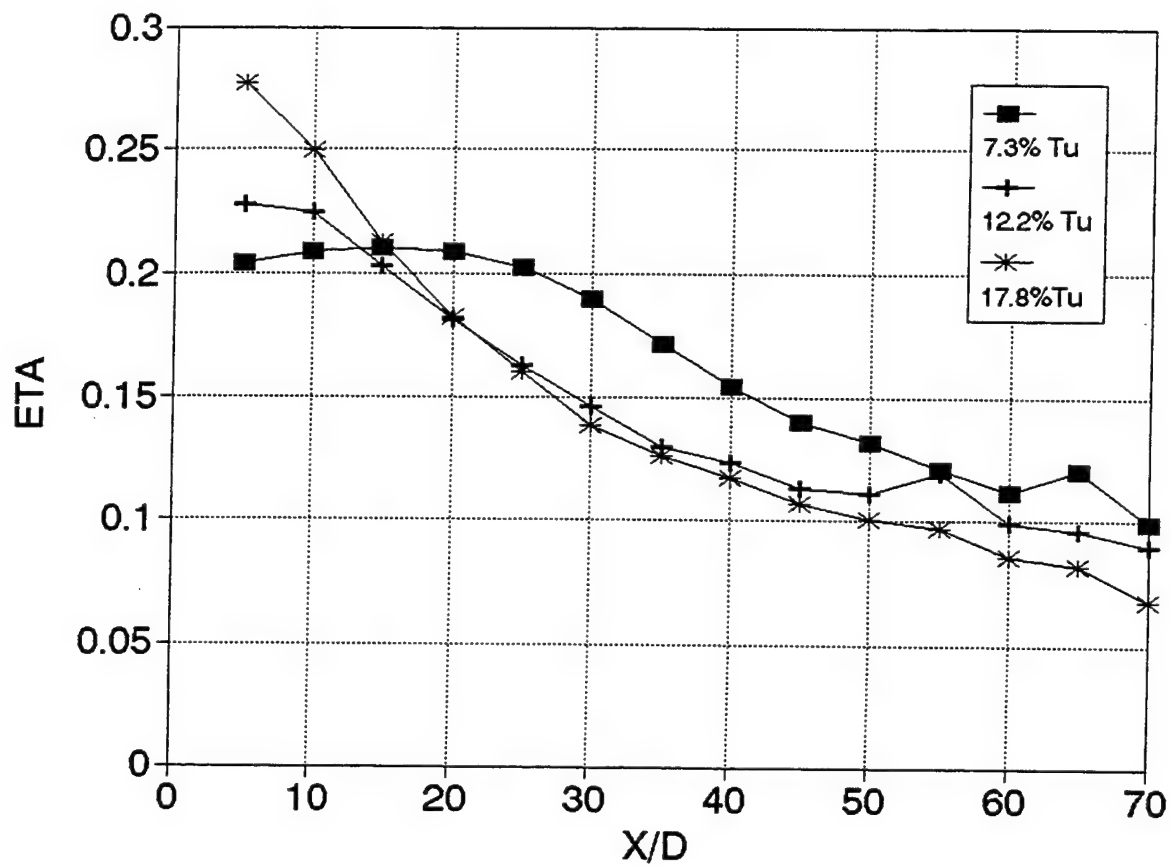


Figure 5-4. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.75$ , Free stream velocity at injection = 10 m/s

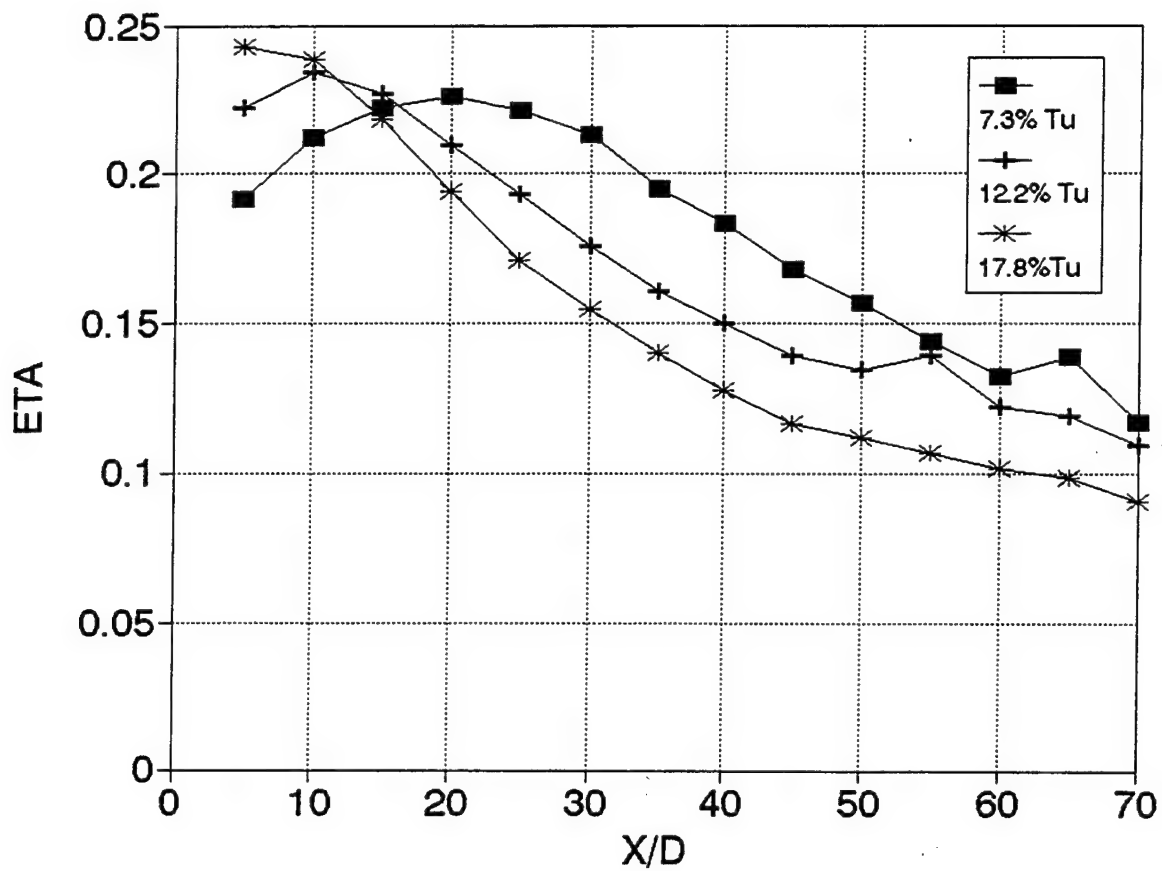


Figure 5-5. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.0$ , Free stream velocity at injection = 10 m/s

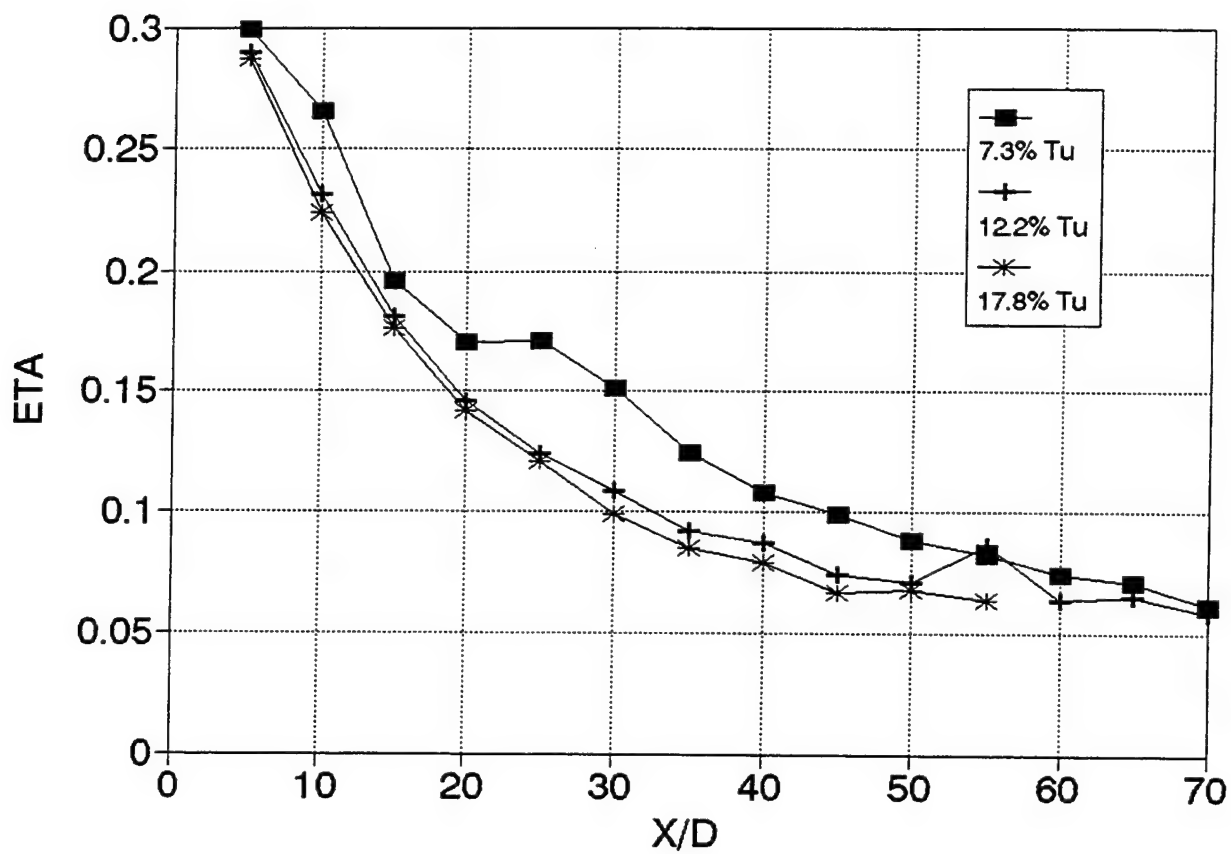


Figure 5-6. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.50$ , Free stream velocity at injection = 18 m/s

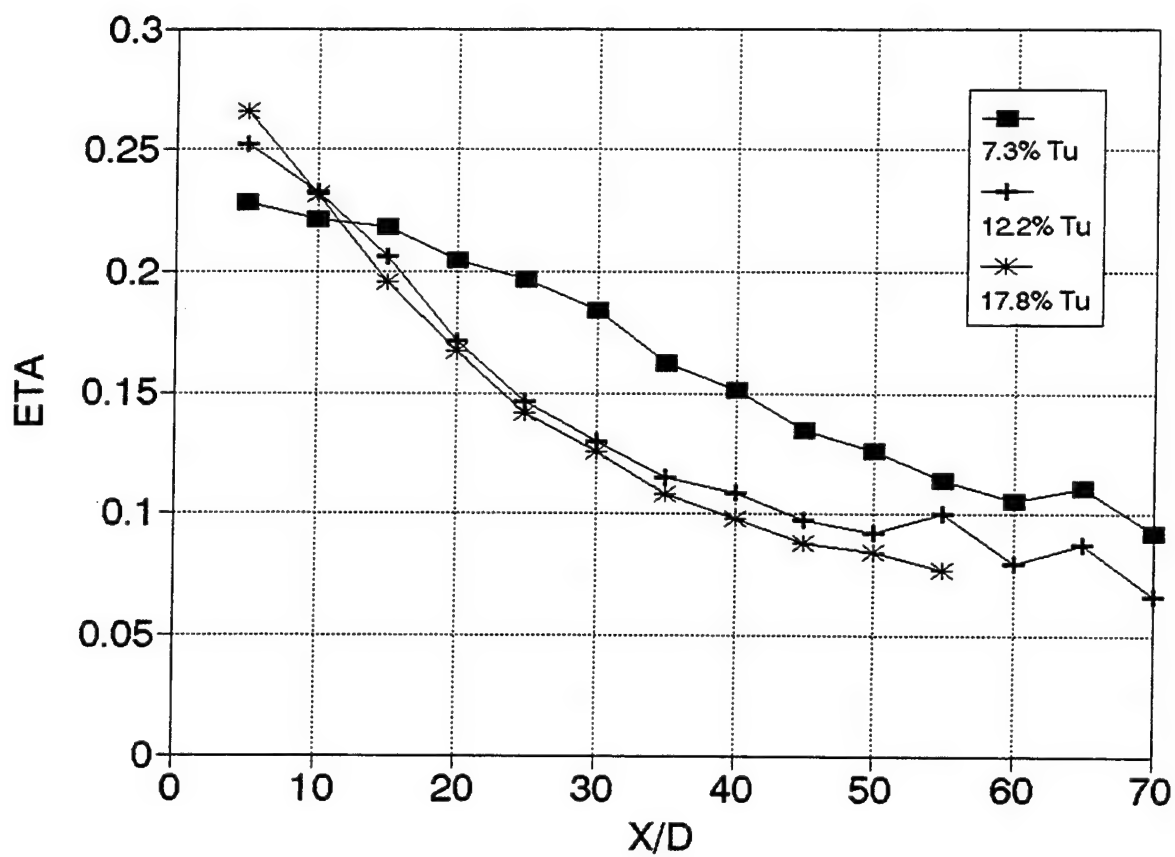


Figure 5-7. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.75$ , Free stream velocity at injection = 18 m/s

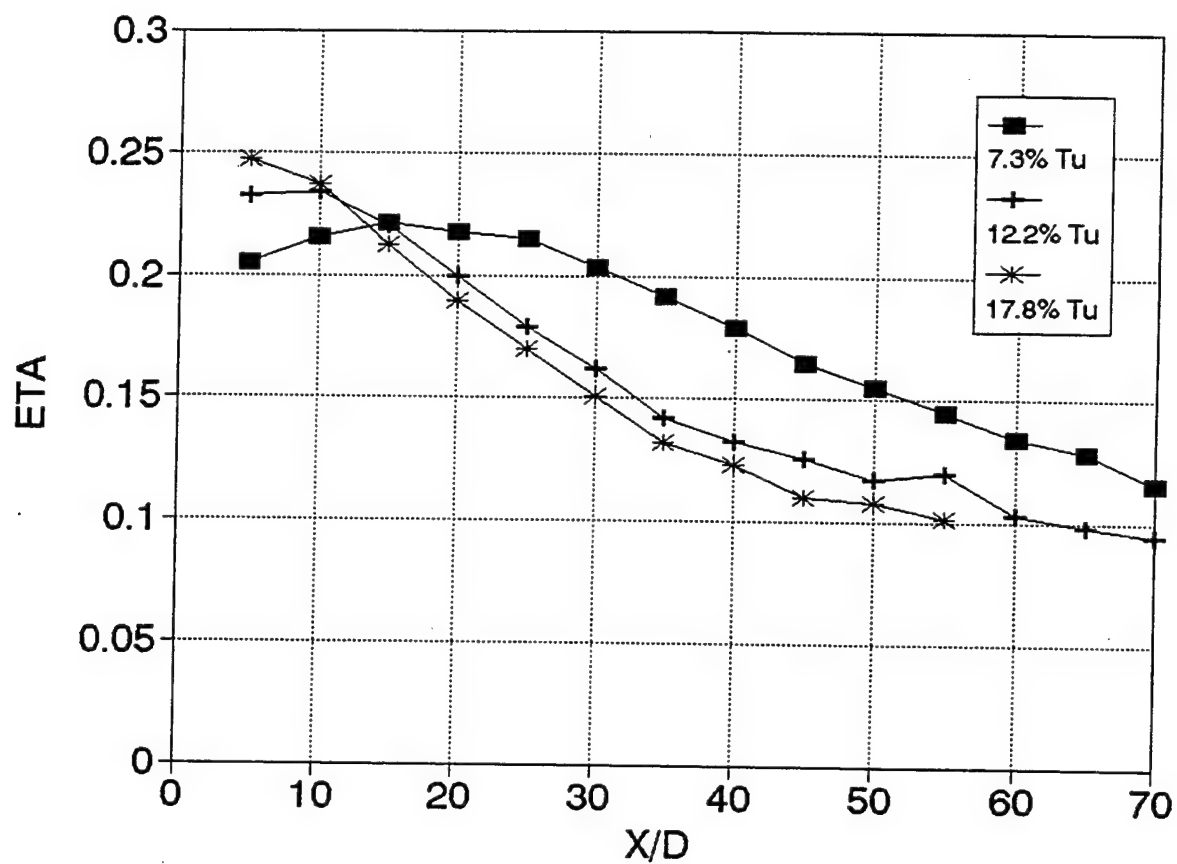


Figure 5-8. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.0$ , Free stream velocity at injection = 18 m/s

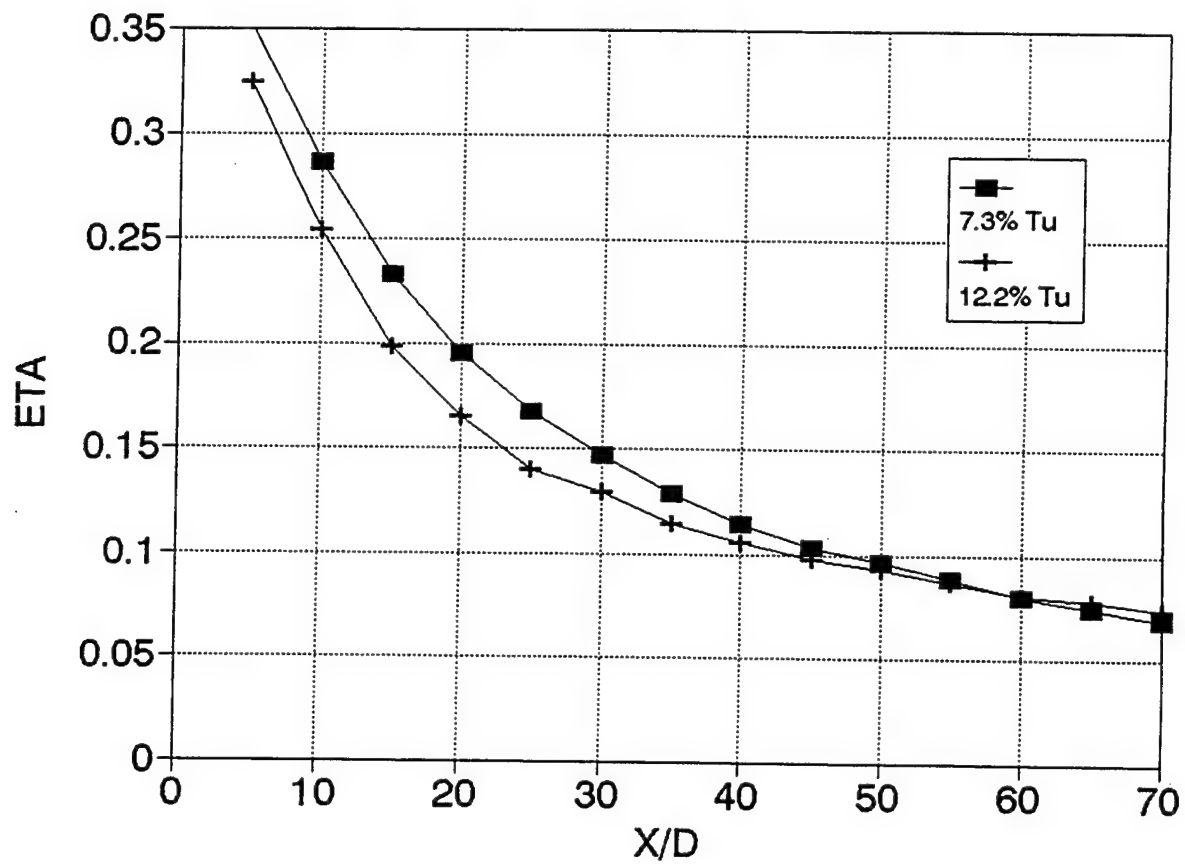


Figure 5-9. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.50$ , Free stream velocity at injection = 37 m/s

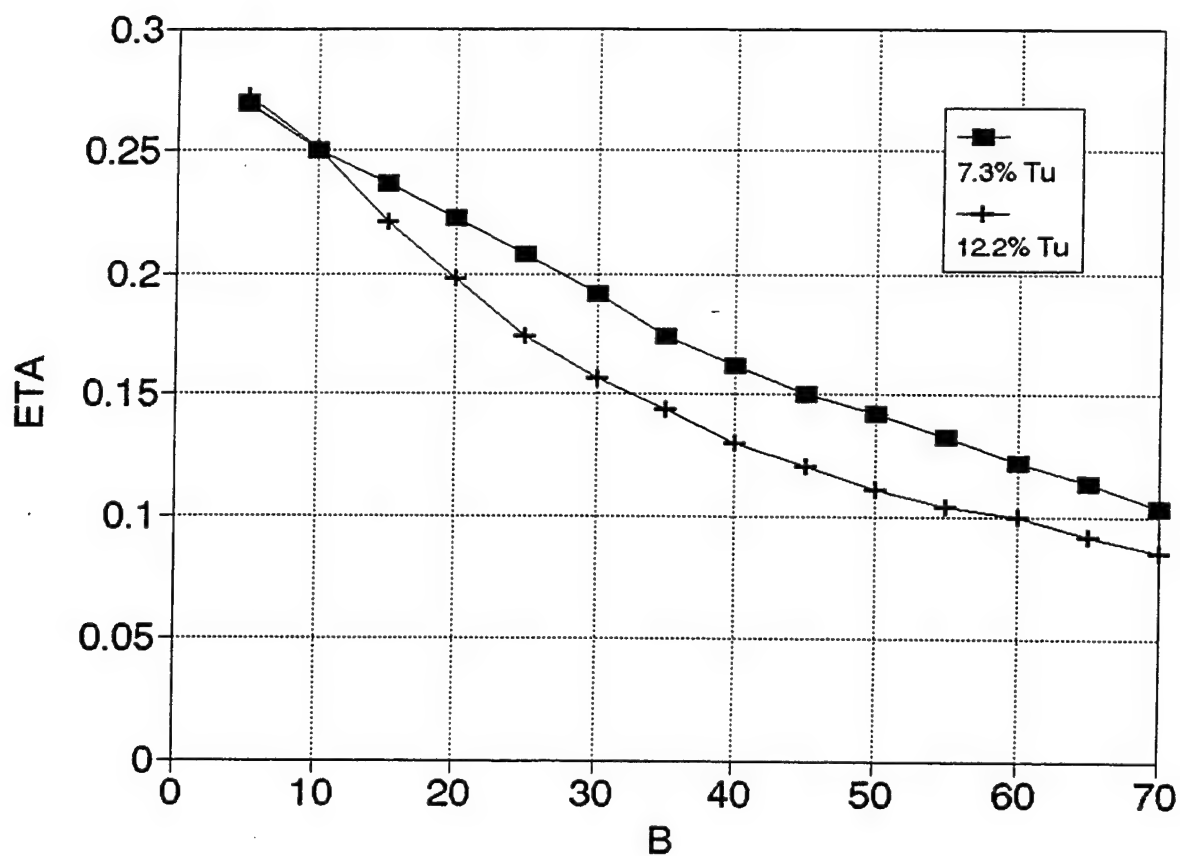


Figure 5-10. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.75$ , Free stream velocity at injection = 37 m/s

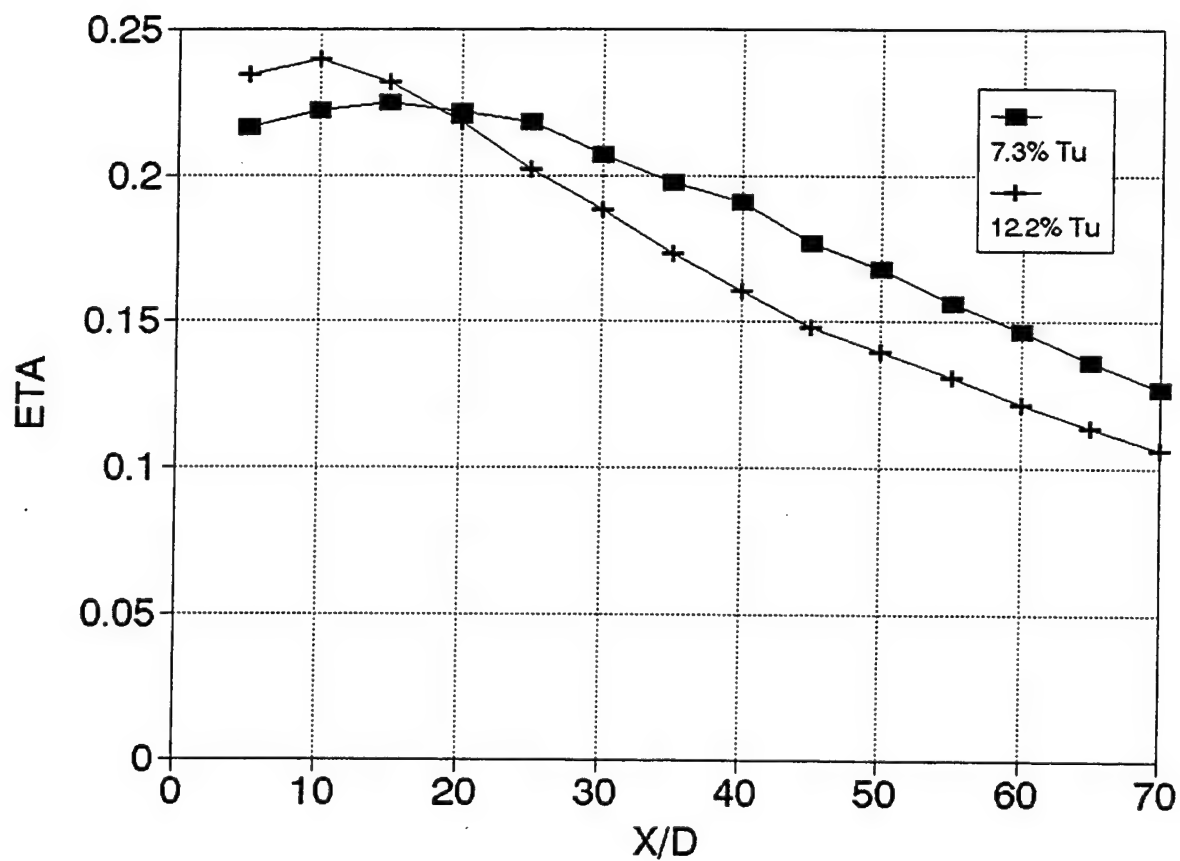


Figure 5-11. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.0$ , Free stream velocity at injection = 37 m/s



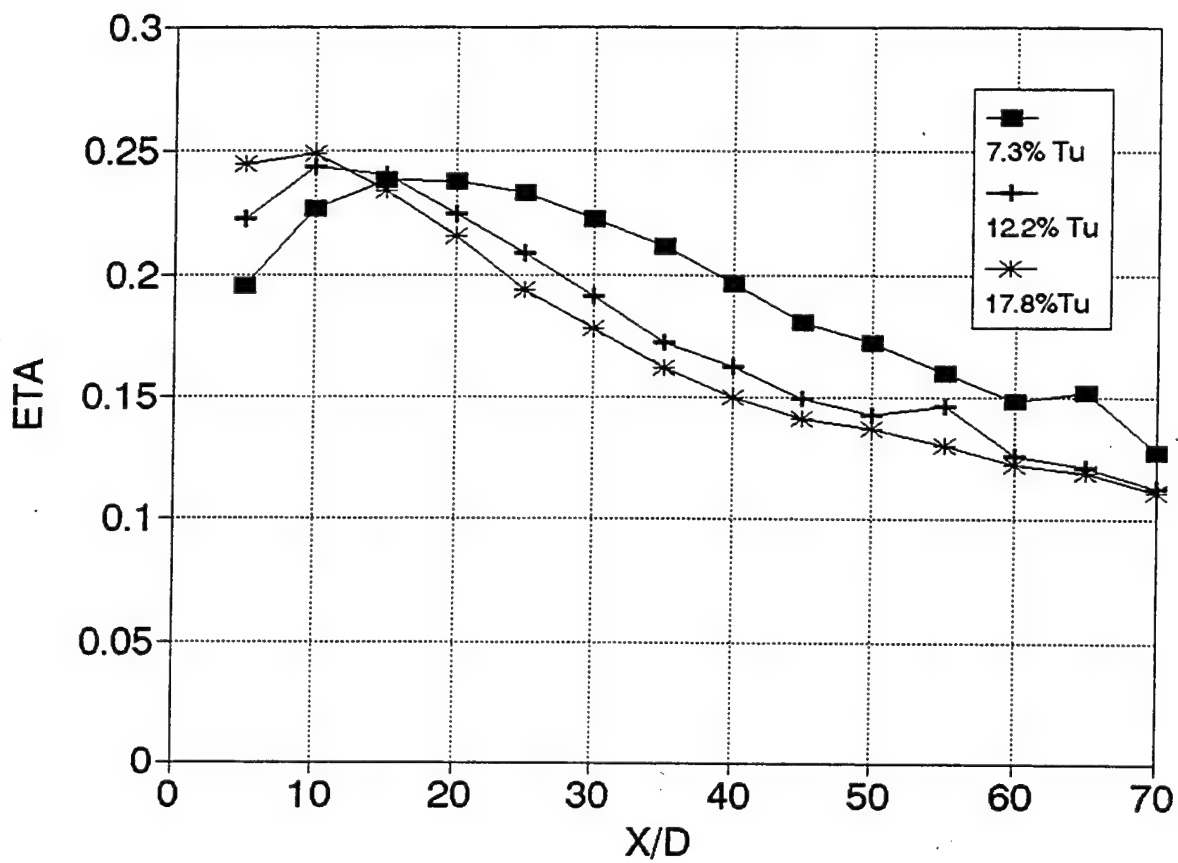


Figure 5-12. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.25$ , Free stream velocity at injection = 10 m/s

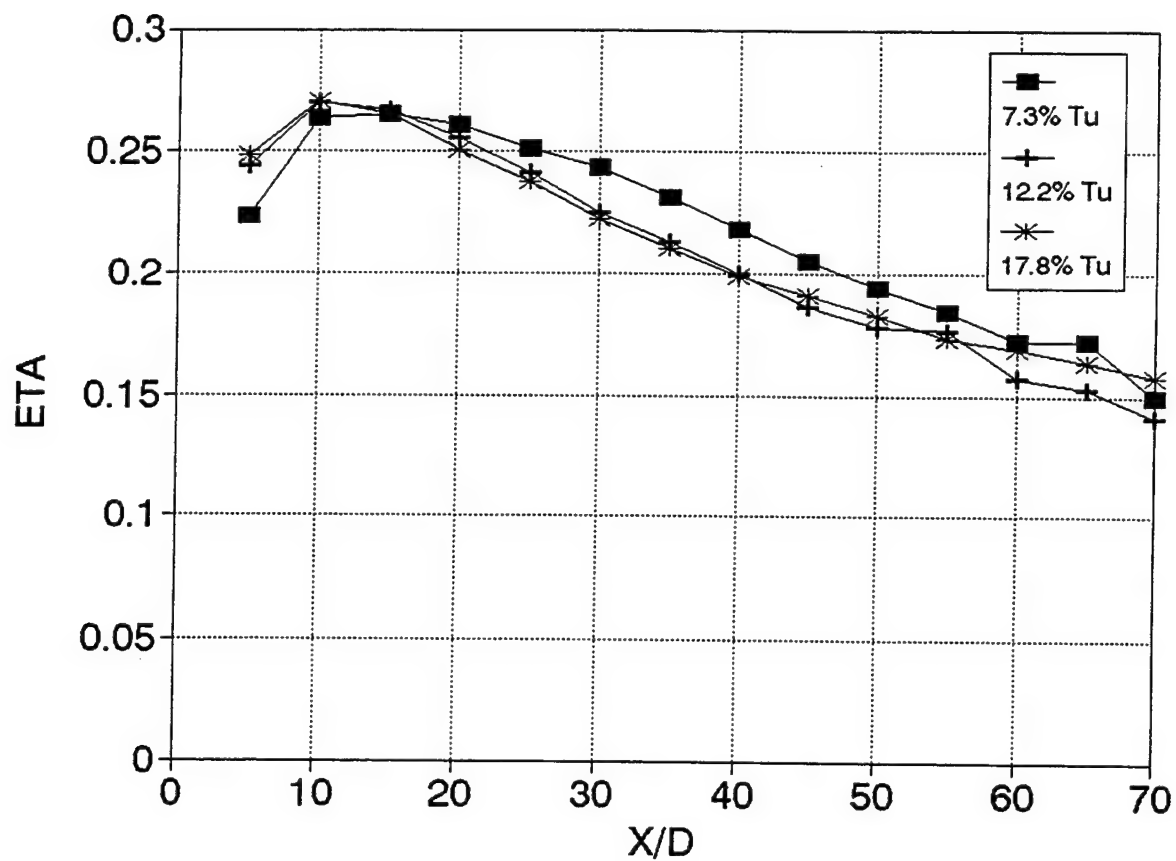


Figure 5-13. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 2.0$ , Free stream velocity at injection = 10 m/s

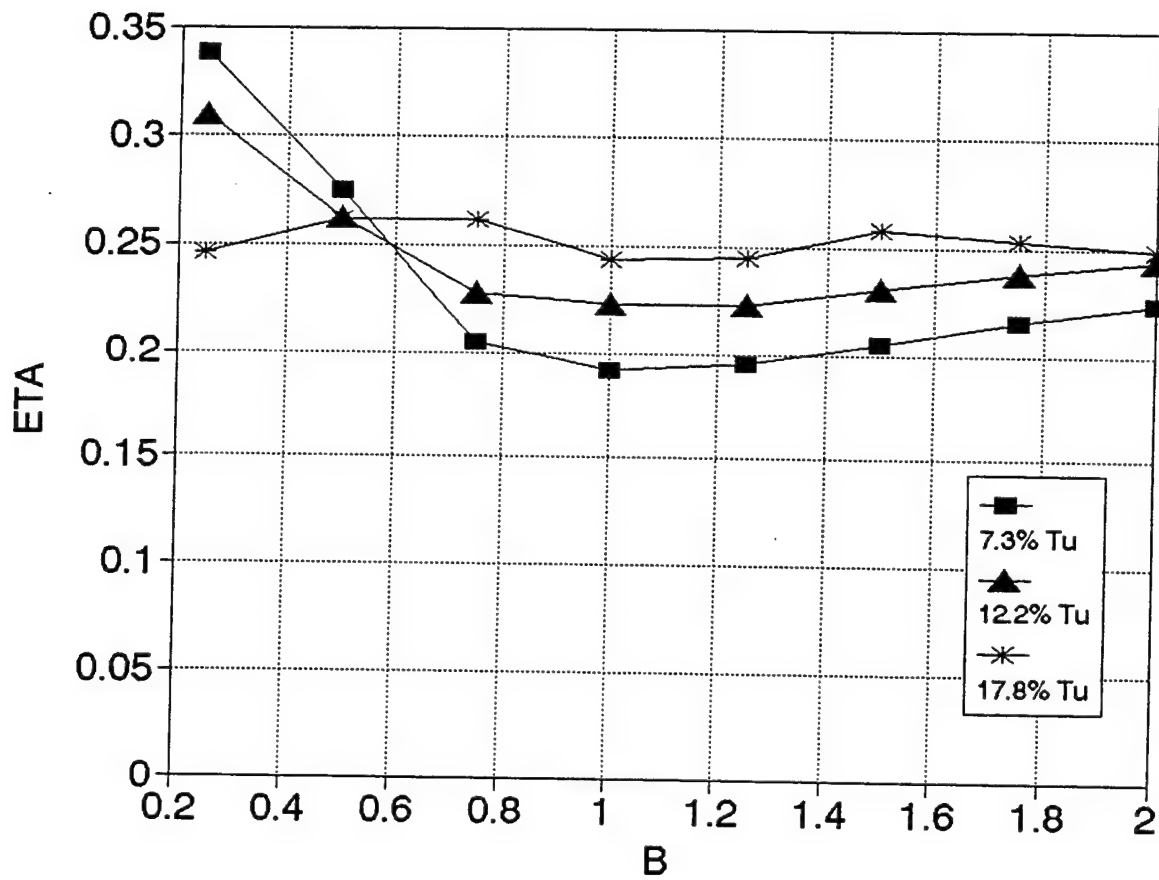


Figure 5-14. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 5$ , Free stream velocity at injection = 10 m/s

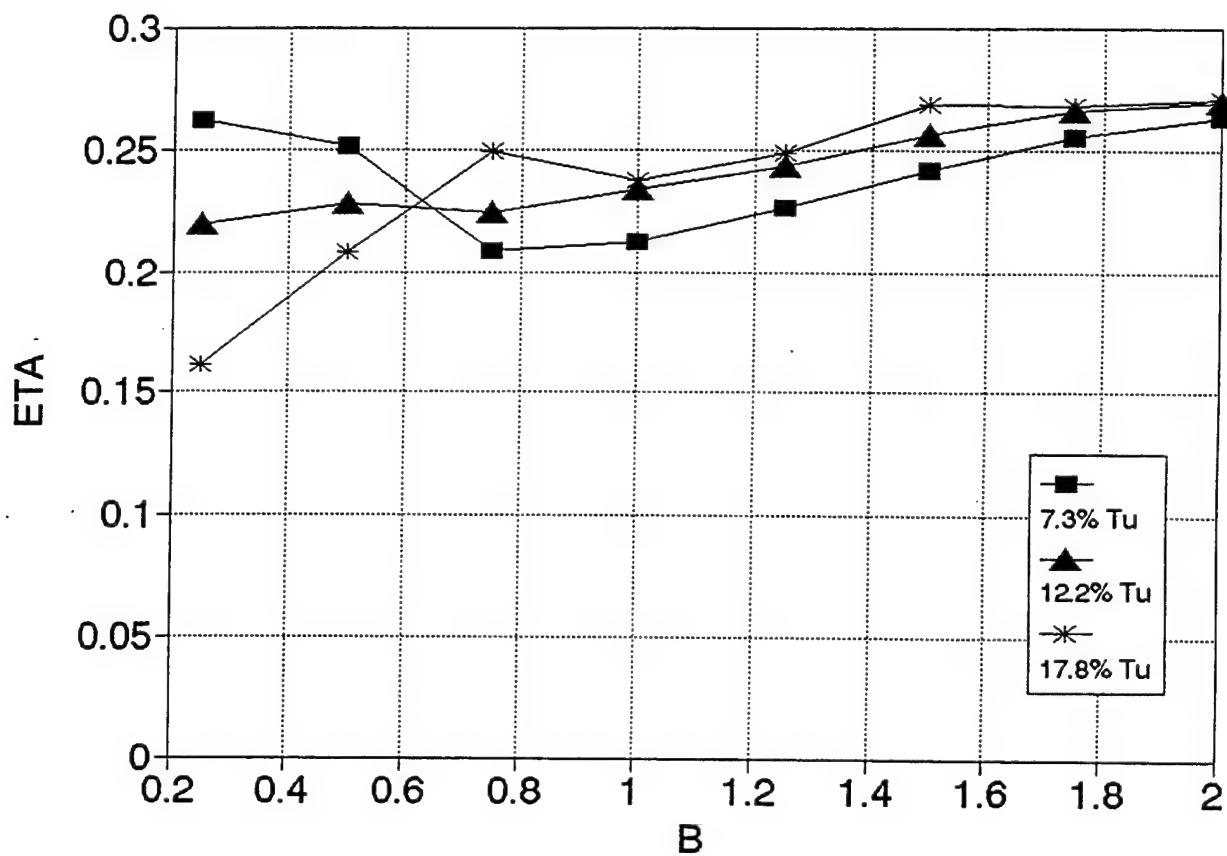


Figure 5-15. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 10$ , Free stream velocity at injection = 10 m/s

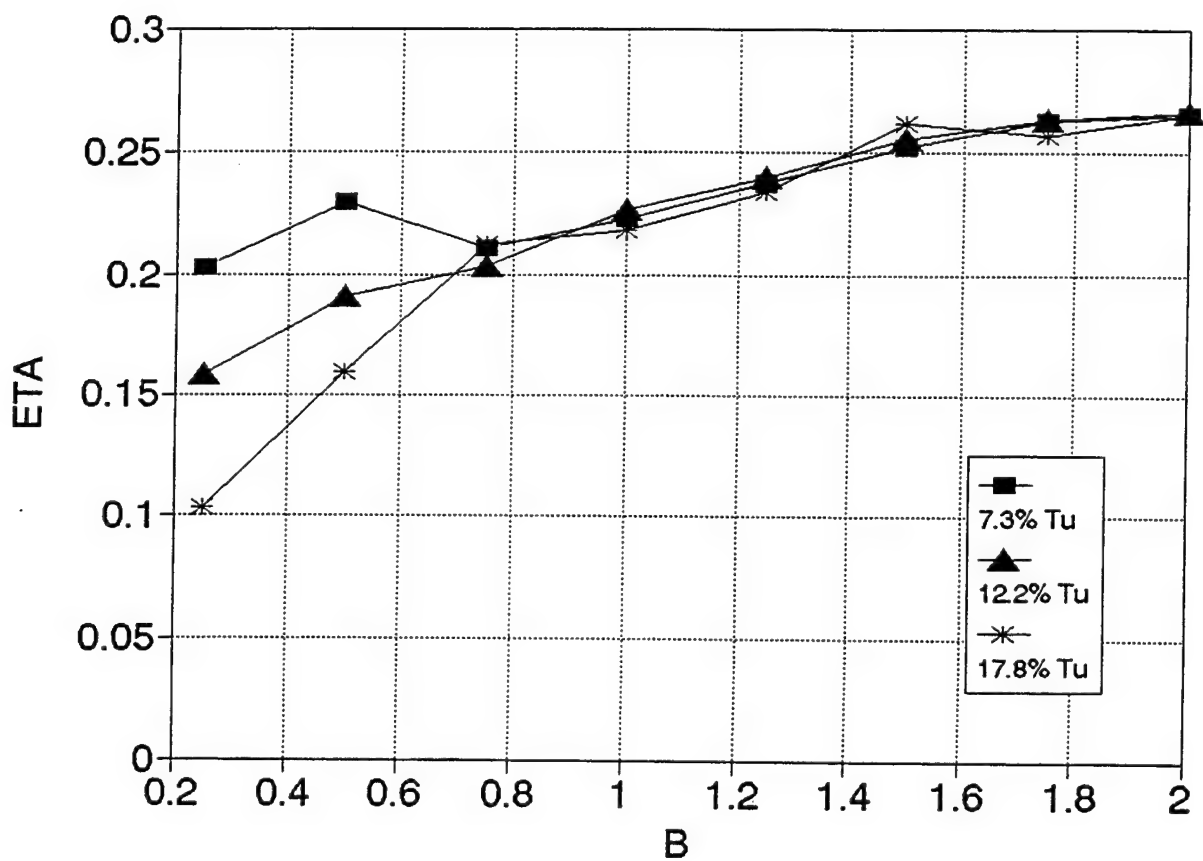


Figure 5-16. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 15$ , Free stream velocity at injection = 10 m/s

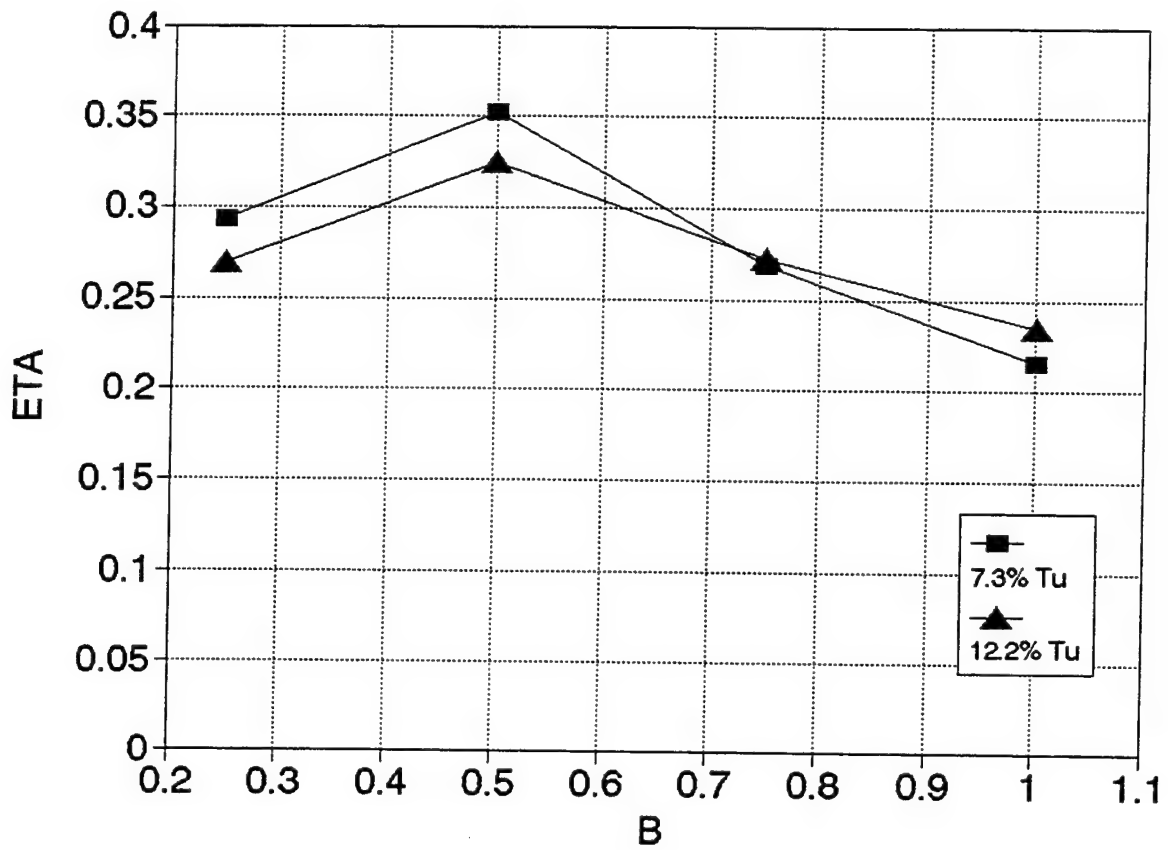


Figure 5-17. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 5$ , Free stream velocity at injection = 37 m/s

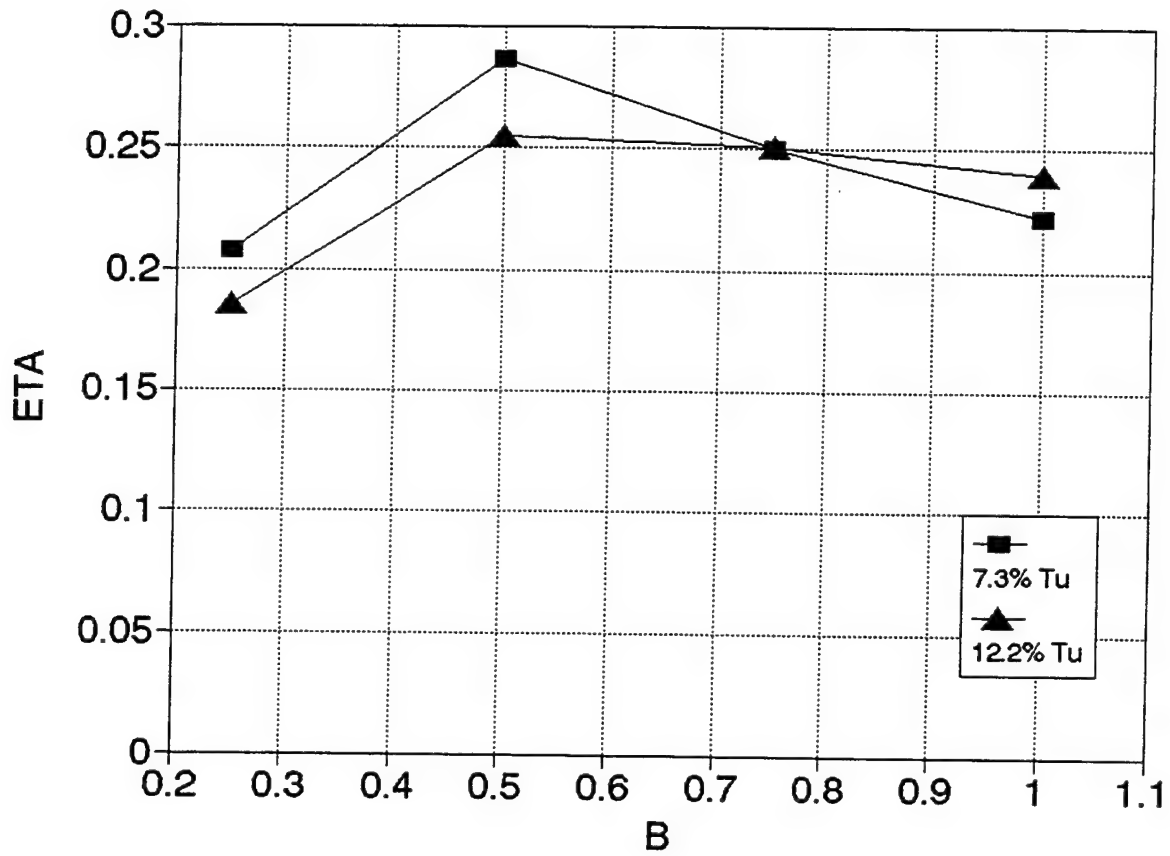


Figure 5-18. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 10$ , Free stream velocity at injection = 37 m/s

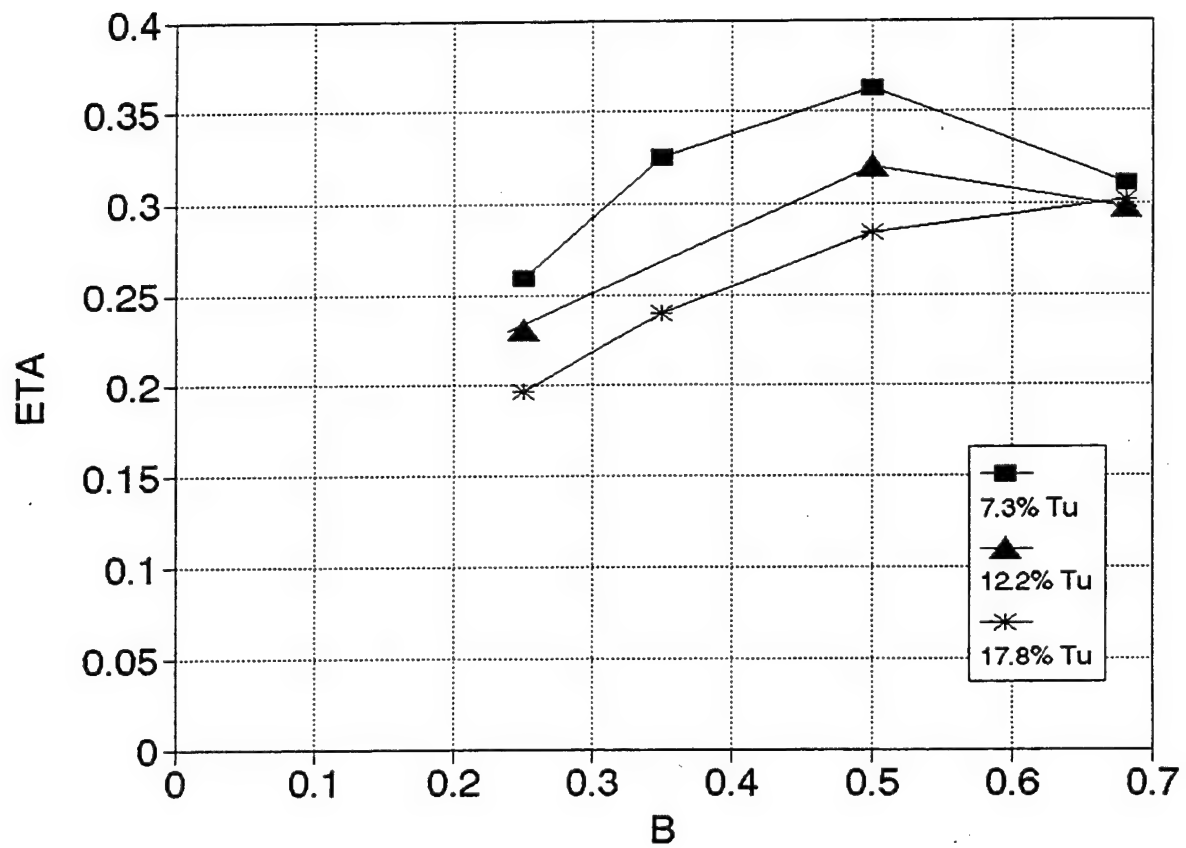


Figure 5-19. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 5$ , Free stream velocity at injection = 60 m/s



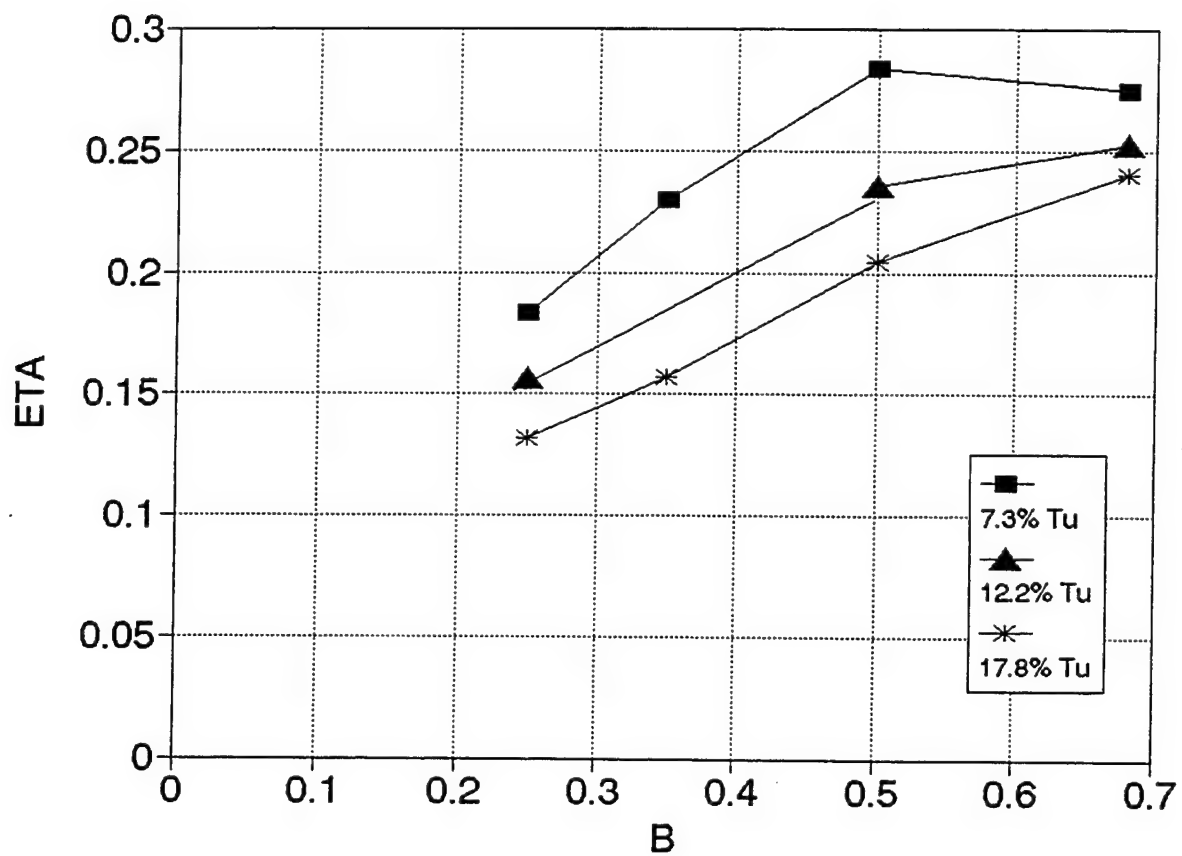


Figure 5-20. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 10$ , Free stream velocity at injection = 60 m/s

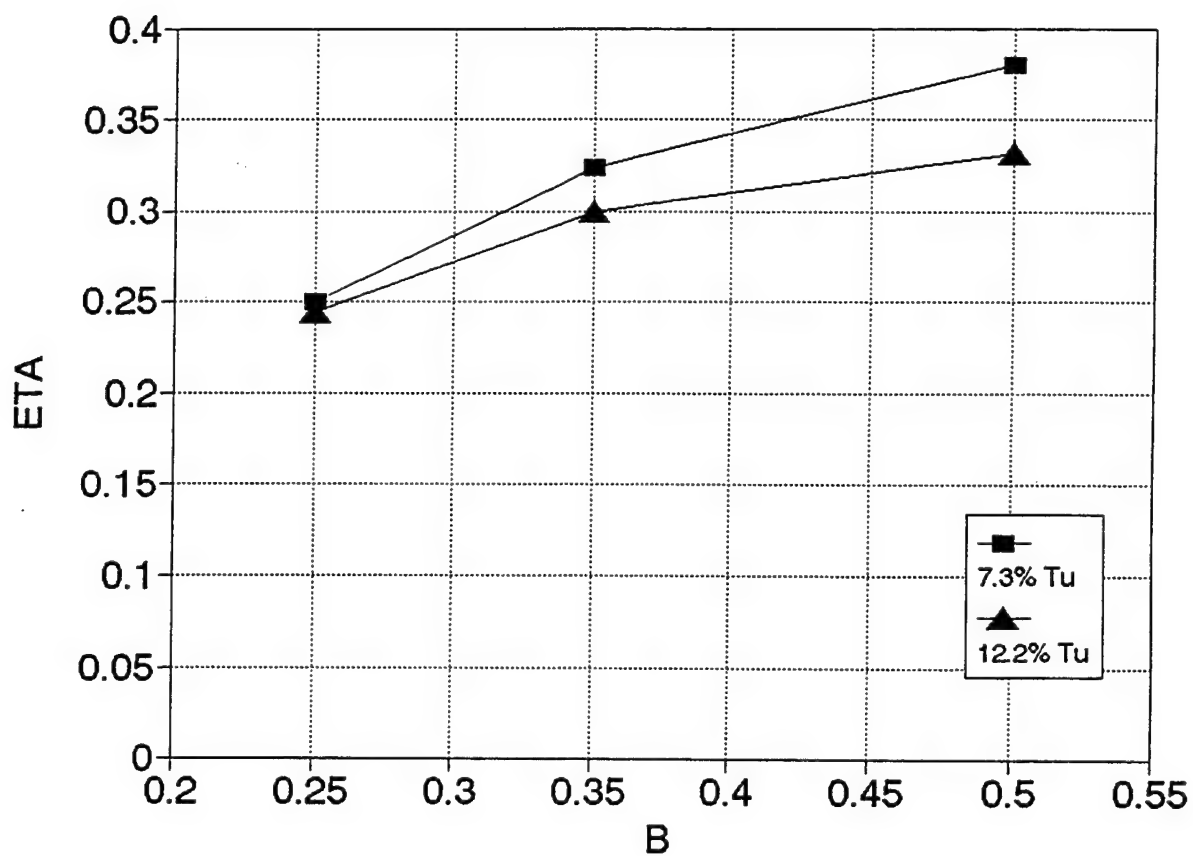


Figure 5-21. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 5$ , Free stream velocity at injection = 85 m/s

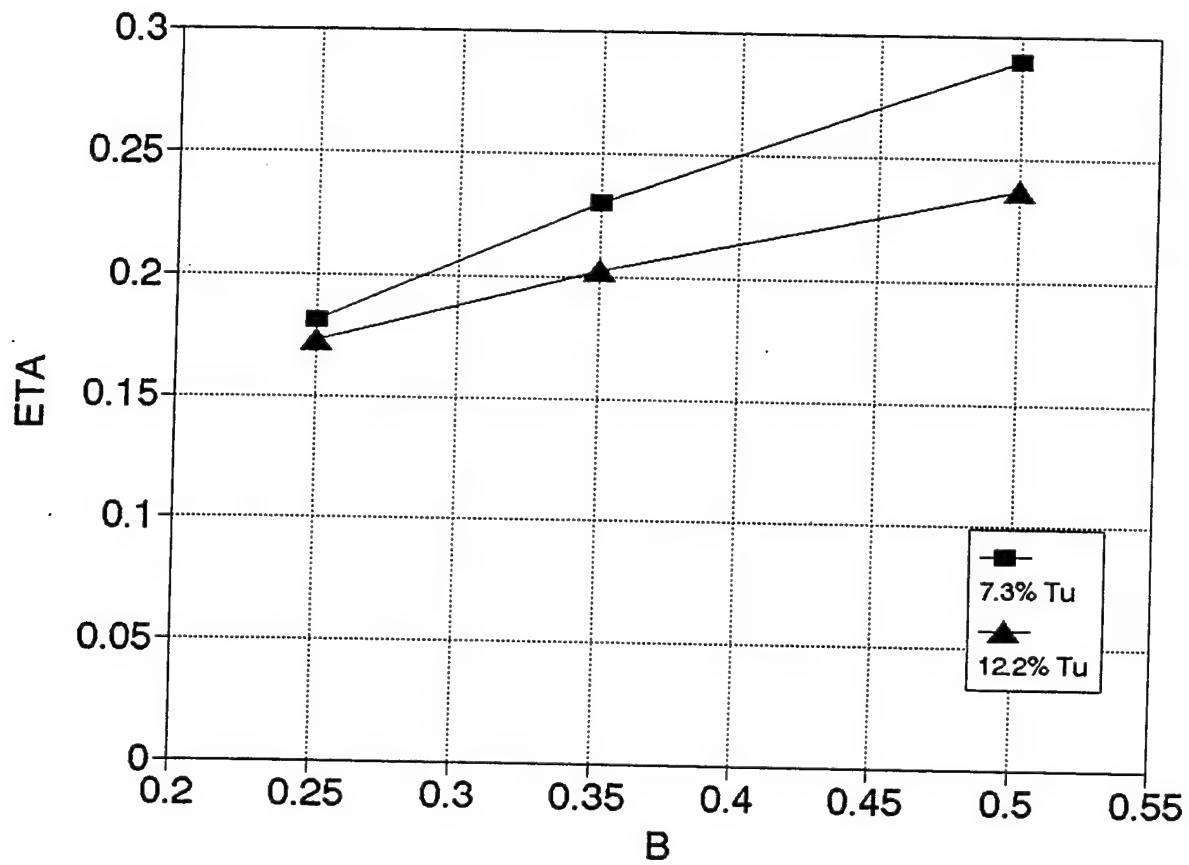


Figure 5-22. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 10$ , Free stream velocity at injection = 85 m/s

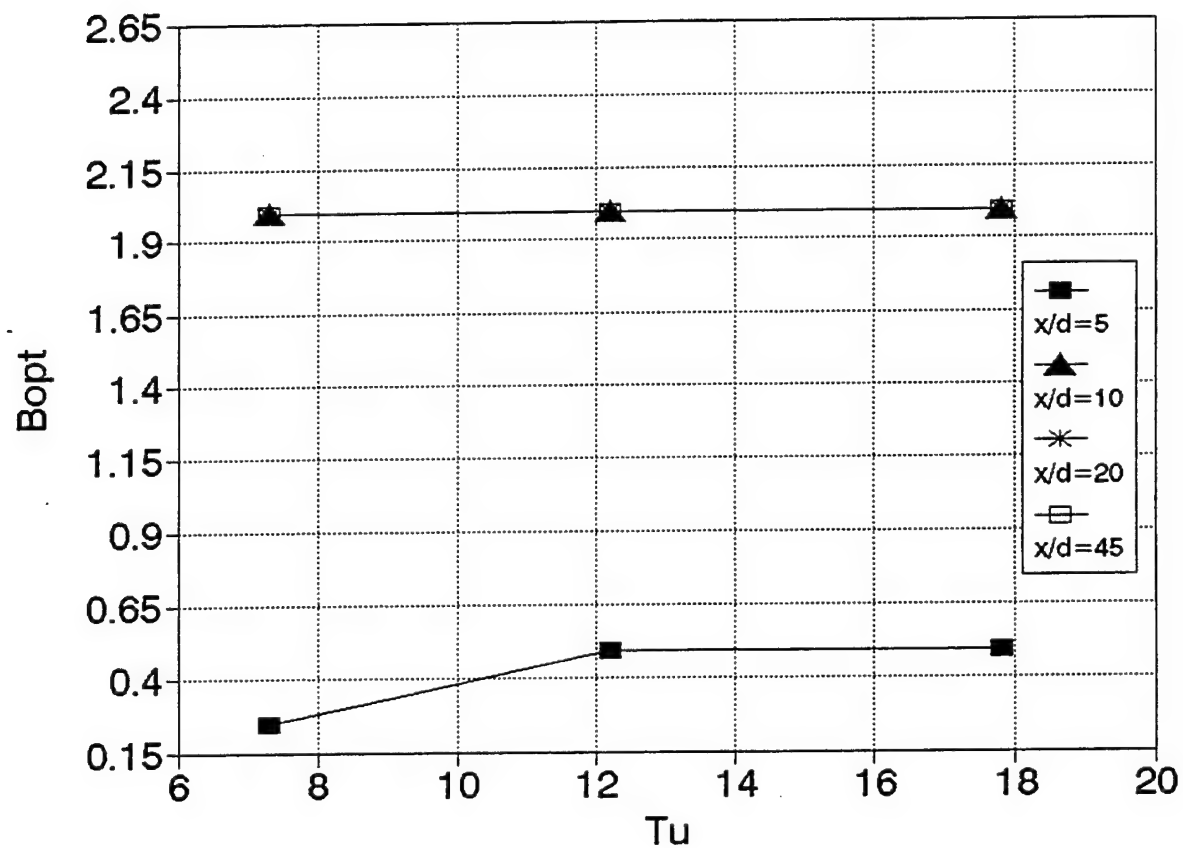


Figure 5-23. Effect of Free Stream Turbulence on Optimum Blowing Ratio  
Free stream velocity at injection = 18 m/s

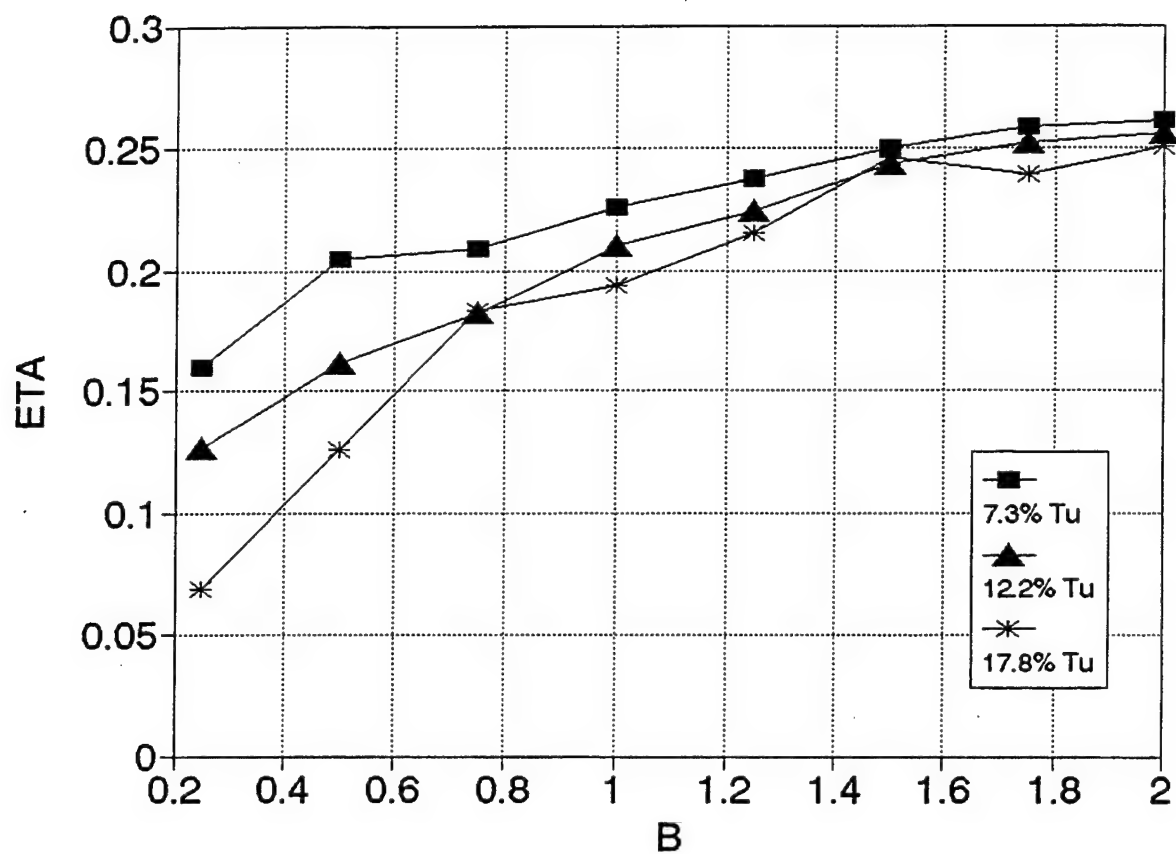


Figure 5-24. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 20$ , Free stream velocity at injection = 10 m/s

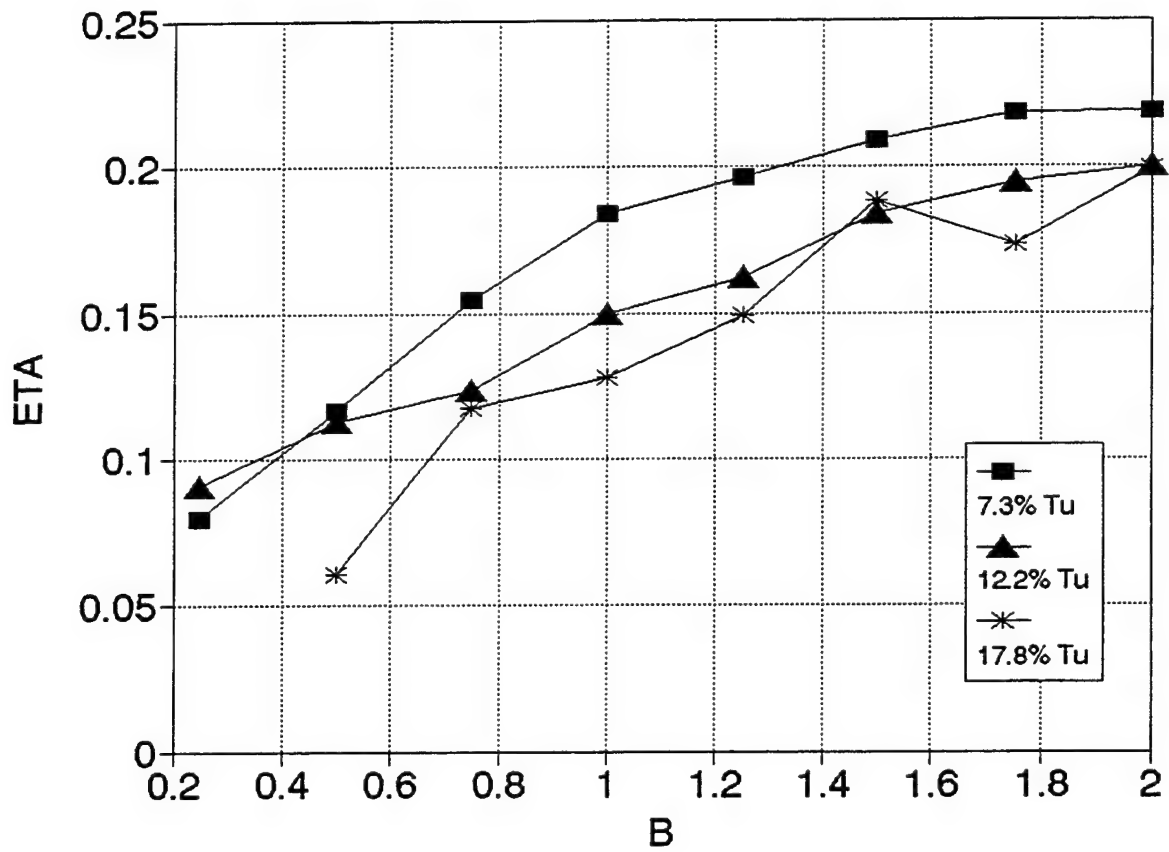


Figure 5-25. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 40$ , Free stream velocity at injection = 10 m/s

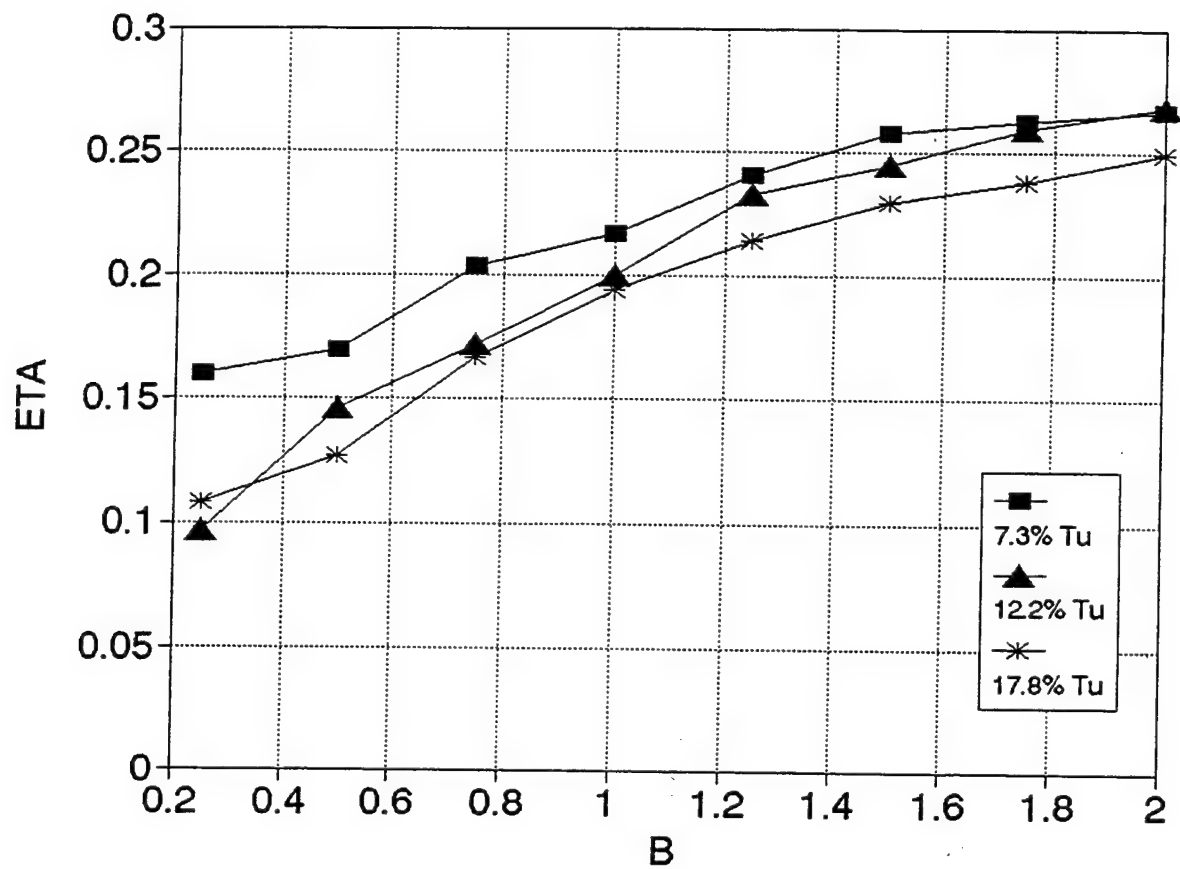


Figure 5-26. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 20$ , Free stream velocity at injection = 18 m/s

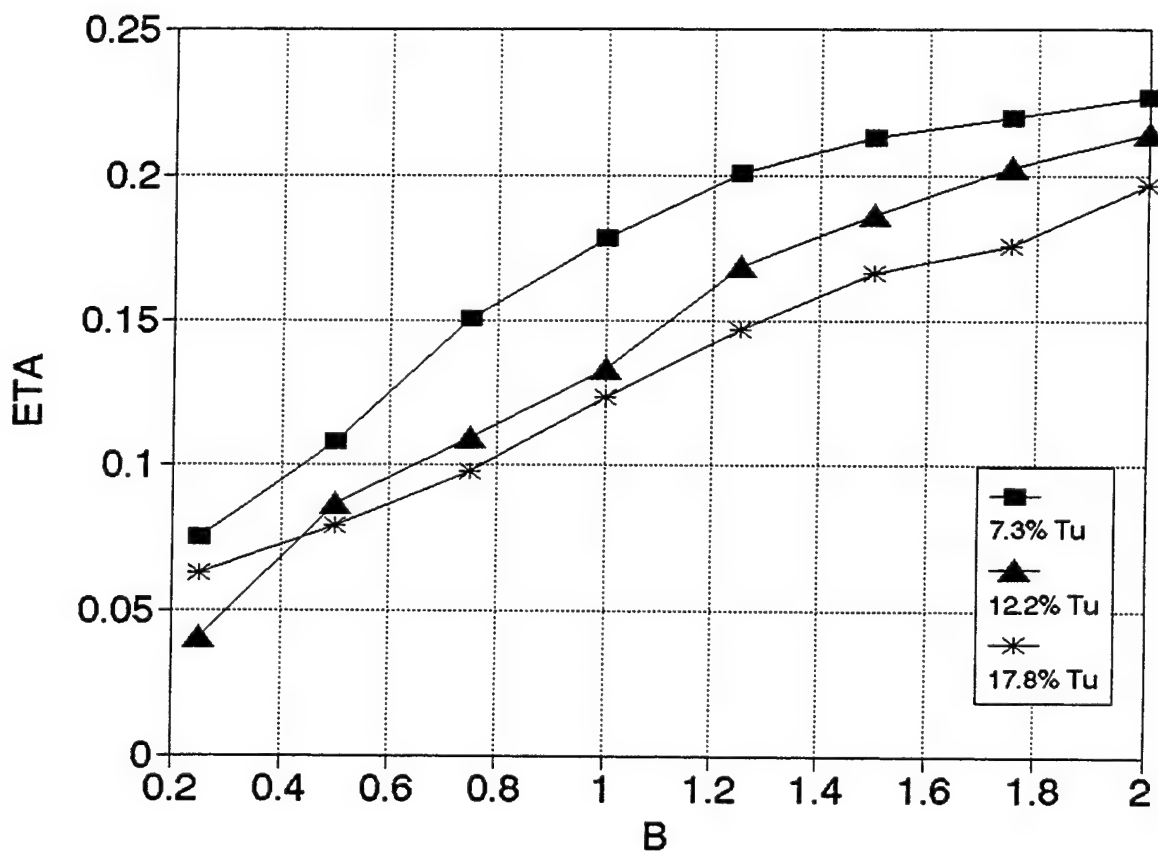


Figure 5-27. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 40$ , Free stream velocity at injection = 18 m/s



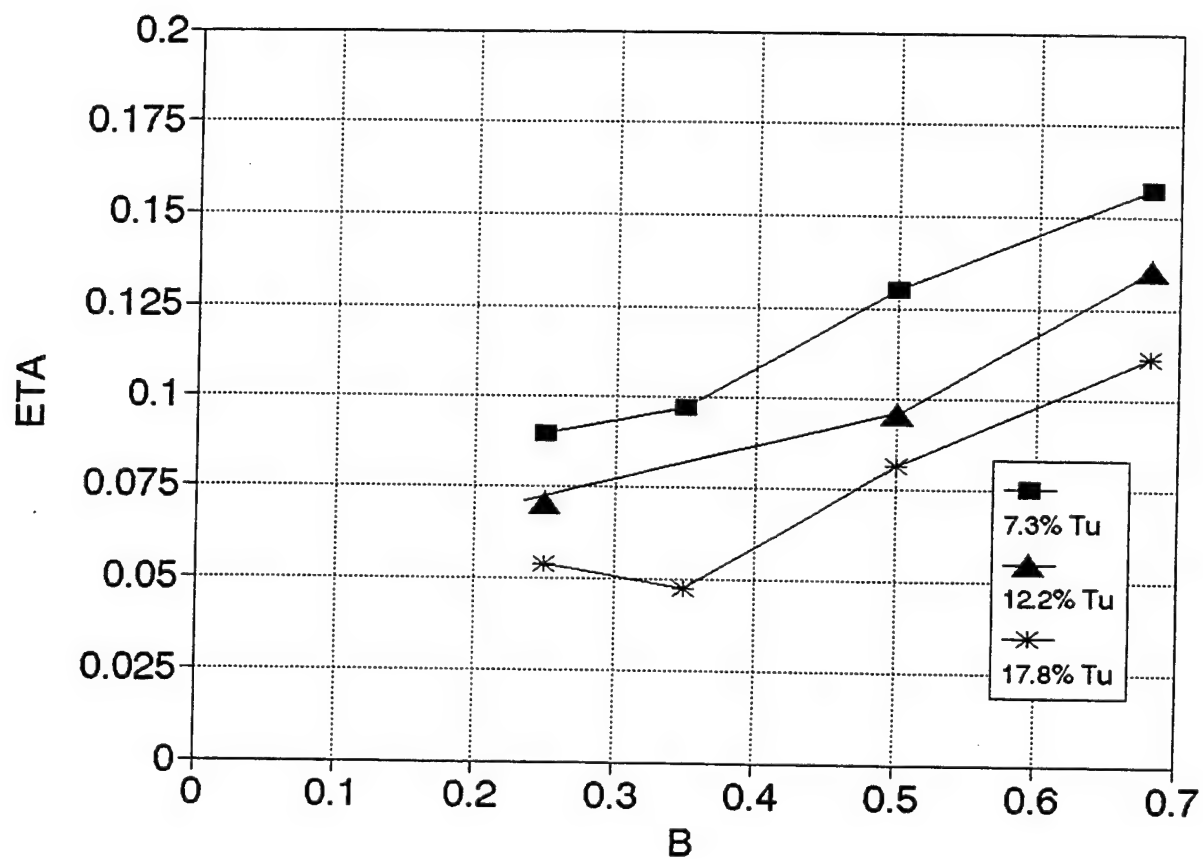


Figure 5-28. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 30$ , Free stream velocity at injection = 60 m/s

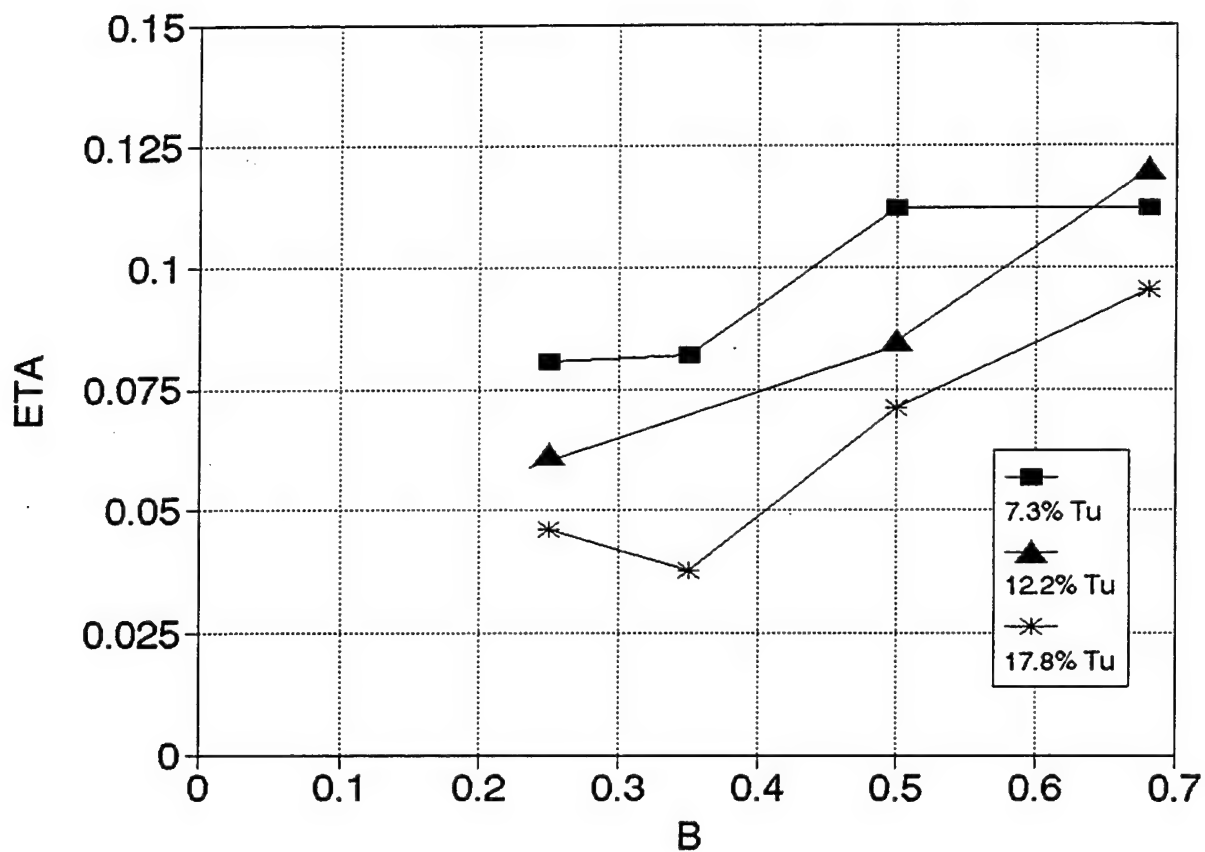


Figure 5-29. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 35$ , Free stream velocity at injection = 60 m/s

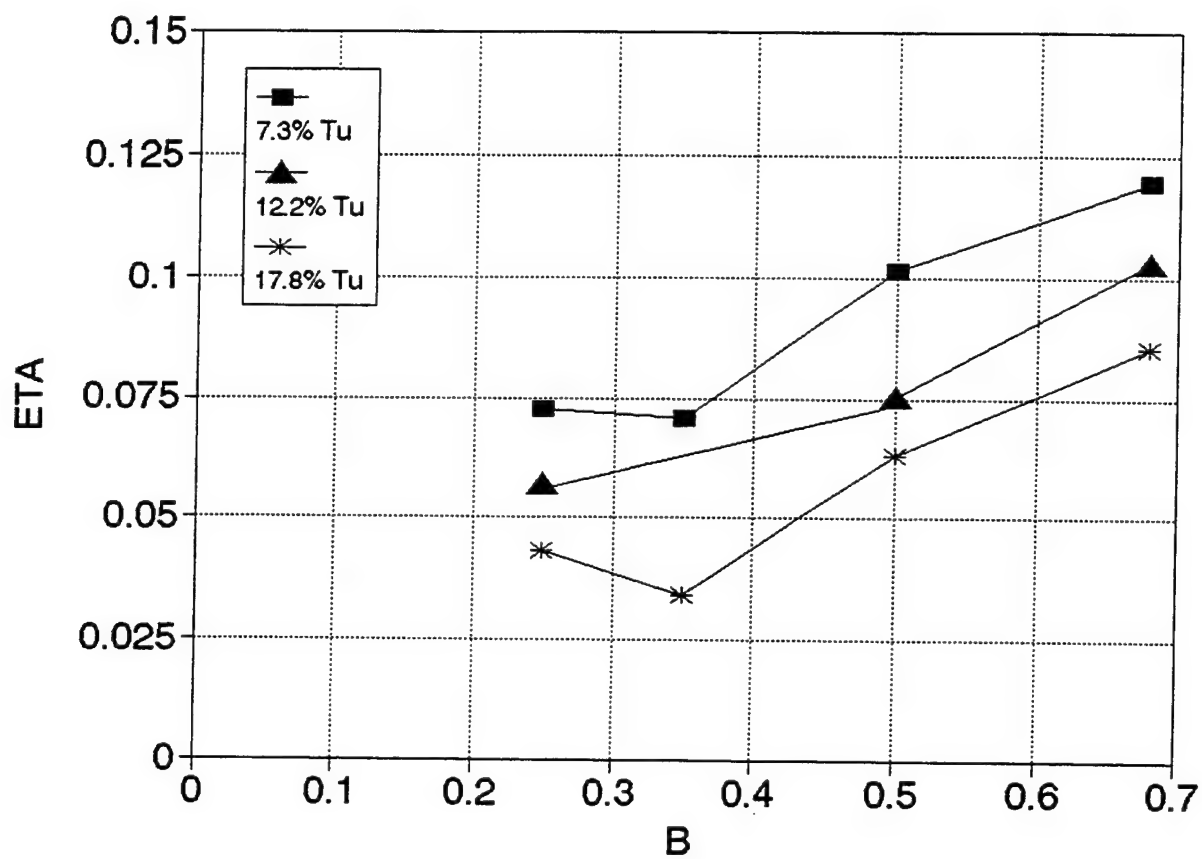


Figure 5-30. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 40$ , Free stream velocity at injection = 60 m/s

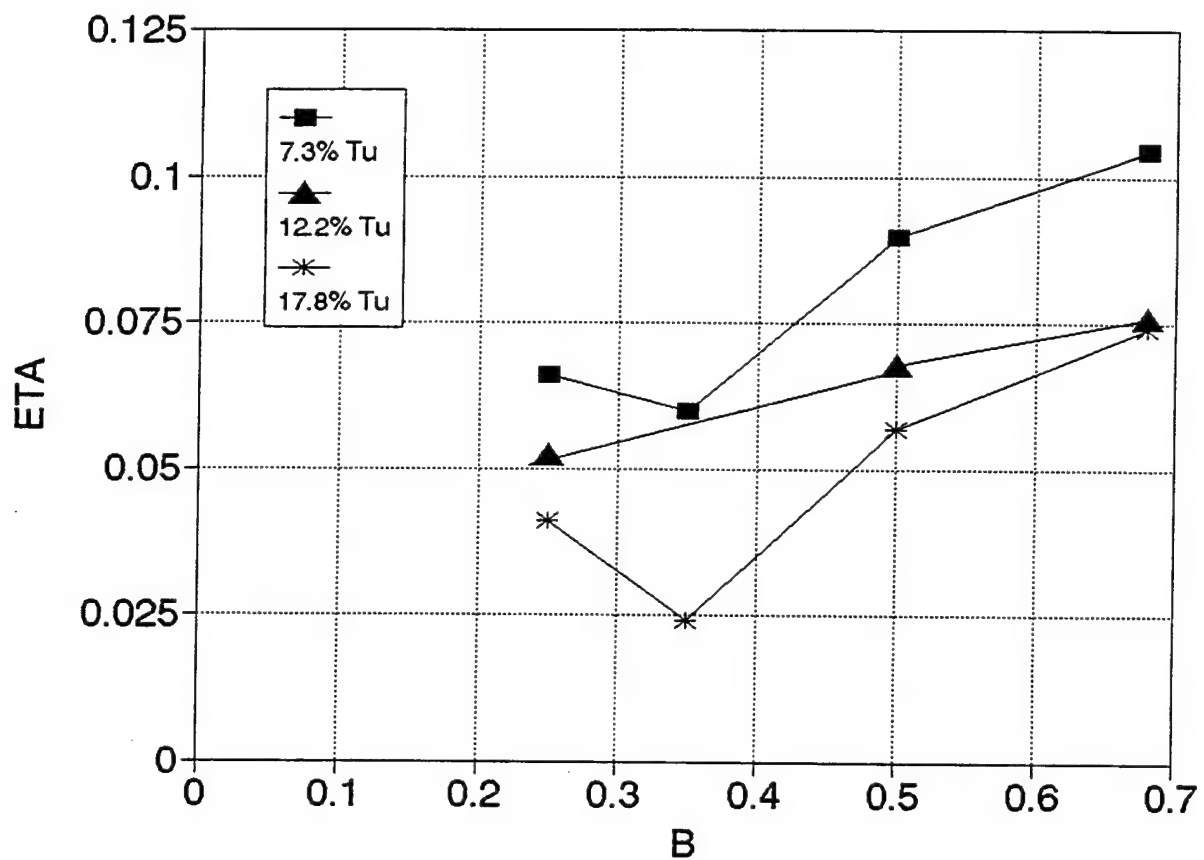


Figure 5-31. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 45$ , Free stream velocity at injection = 60 m/s

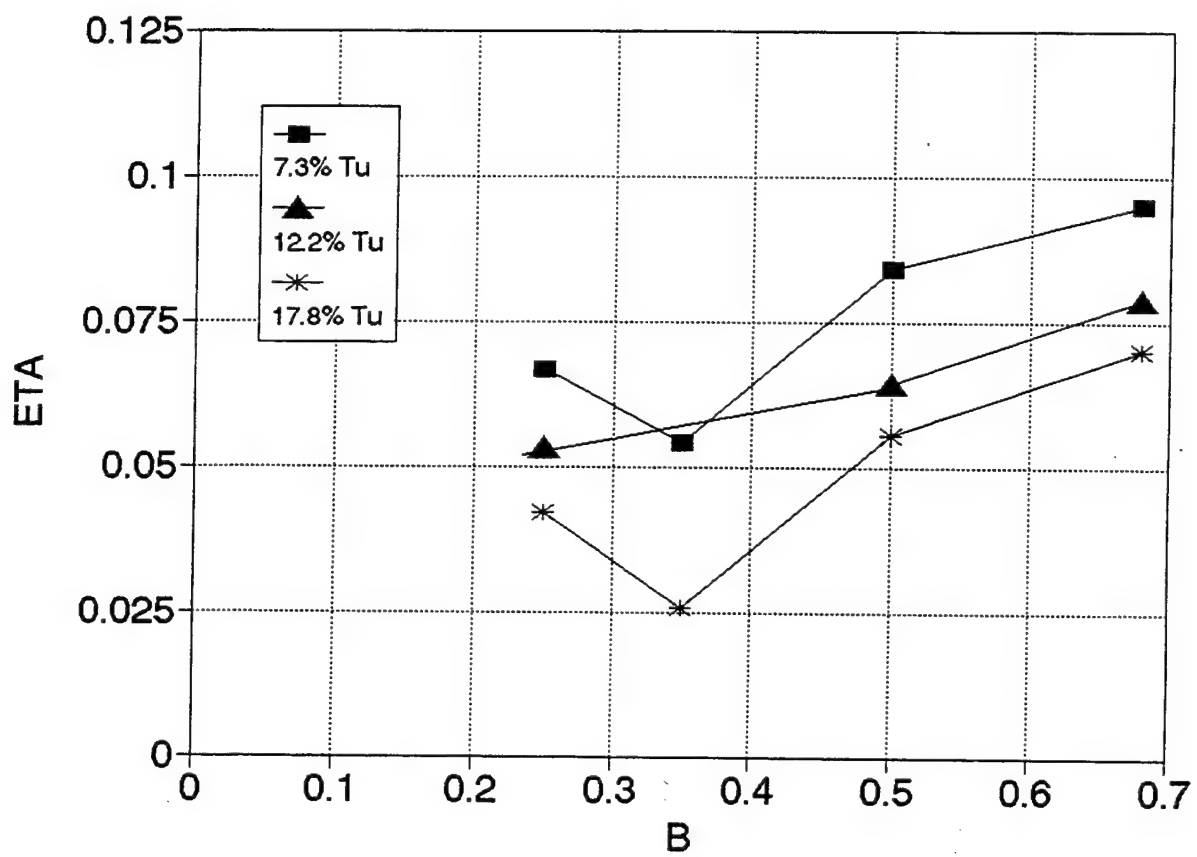


Figure 5-32. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 50$ , Free stream velocity at injection = 60 m/s

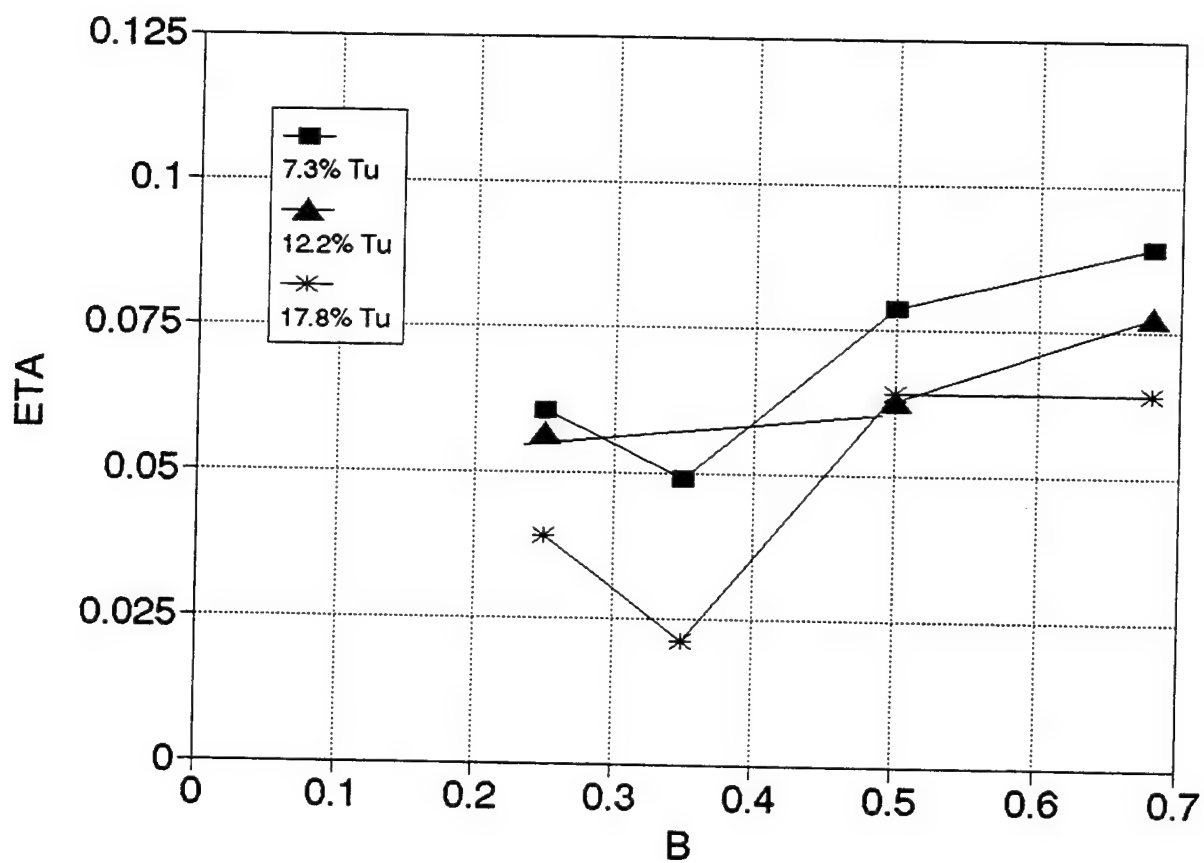


Figure 5-33. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 55$ , Free stream velocity at injection = 60 m/s

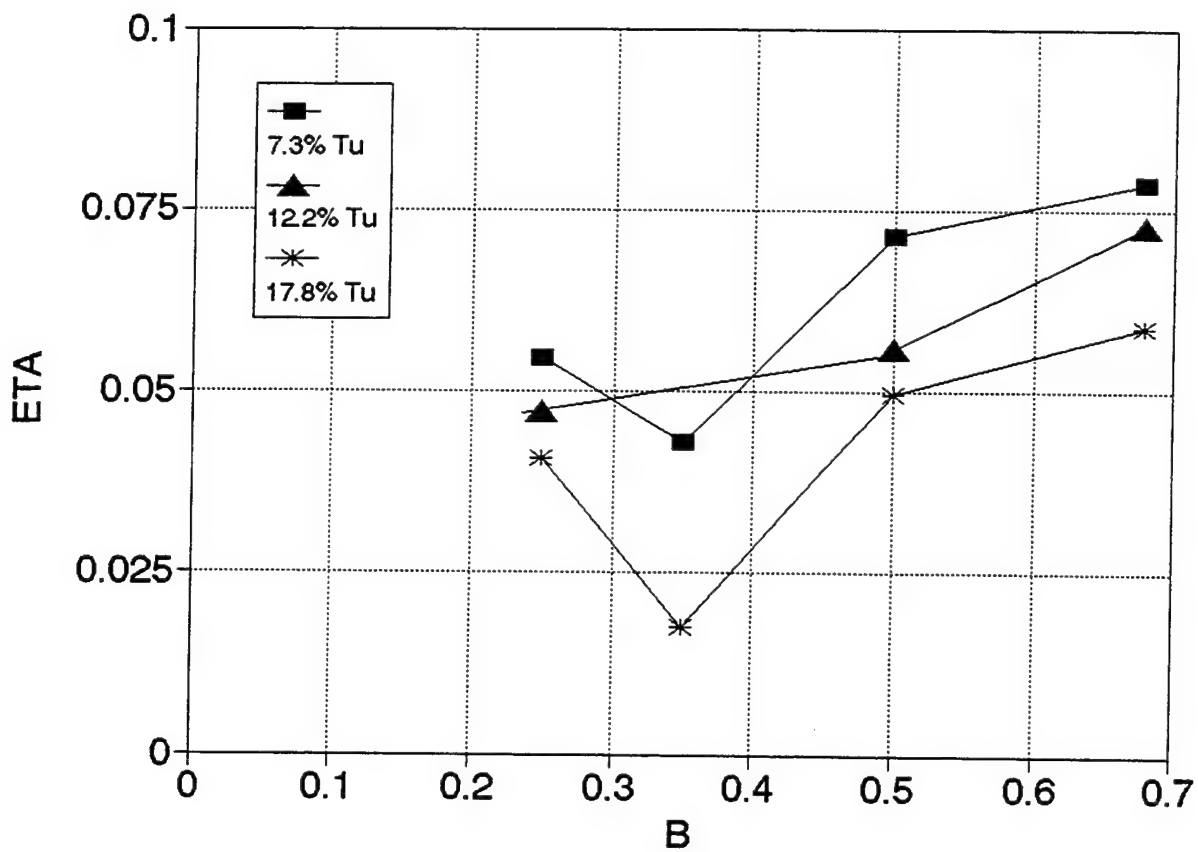


Figure 5-34. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 60$ , Free stream velocity at injection = 60 m/s

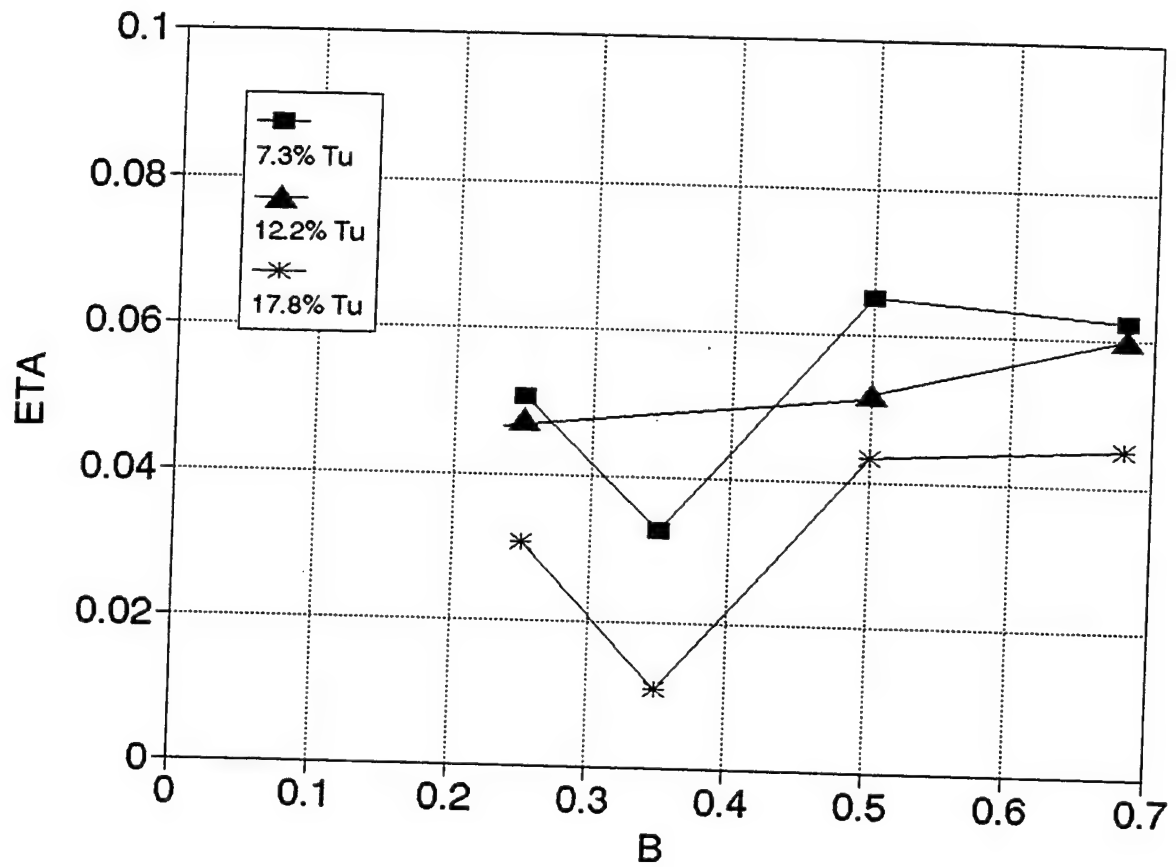


Figure 5-35. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 70$ , Free stream velocity at injection = 60 m/s



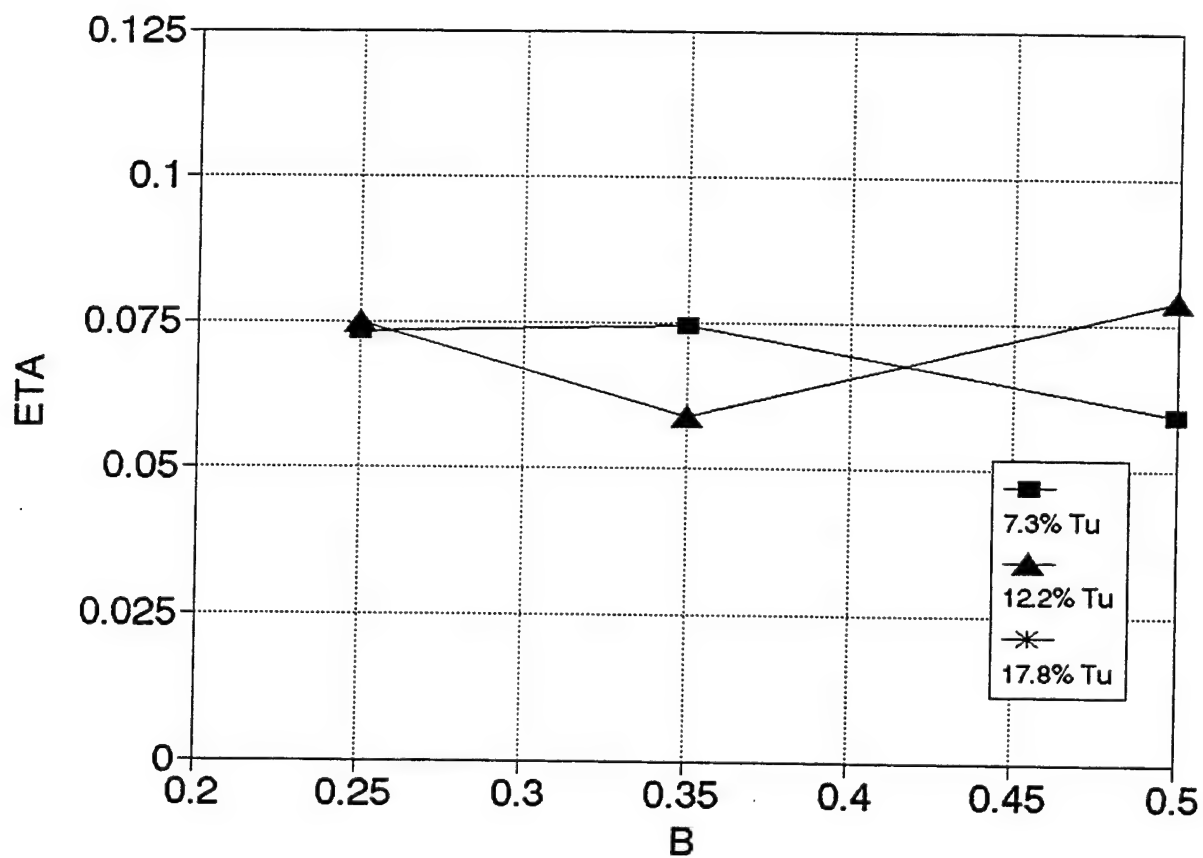


Figure 5-36. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 65$ , Free stream velocity at injection = 85 m/s

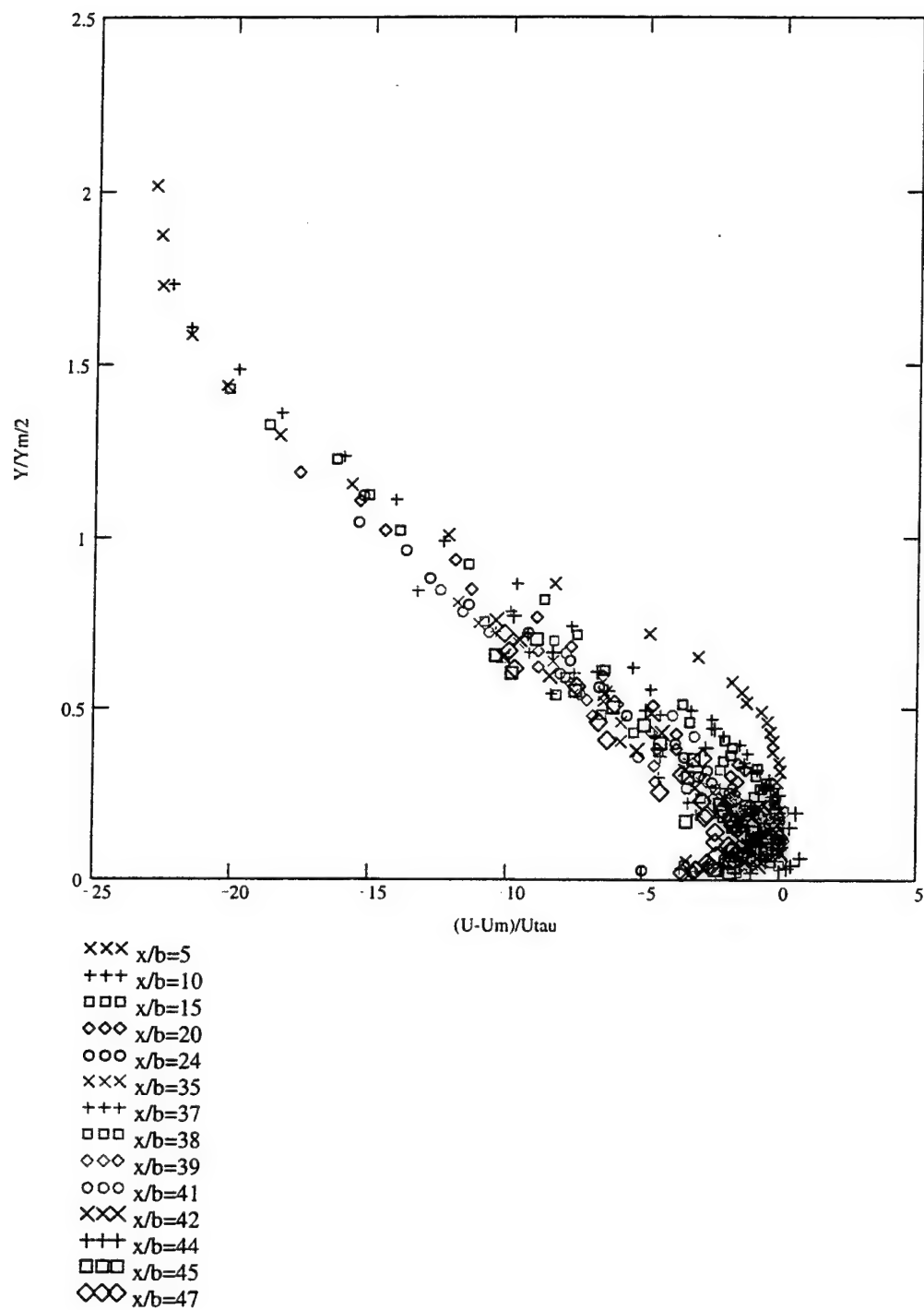


Figure 5-37. Velocity Profile Scaling Using Outer Law Variables

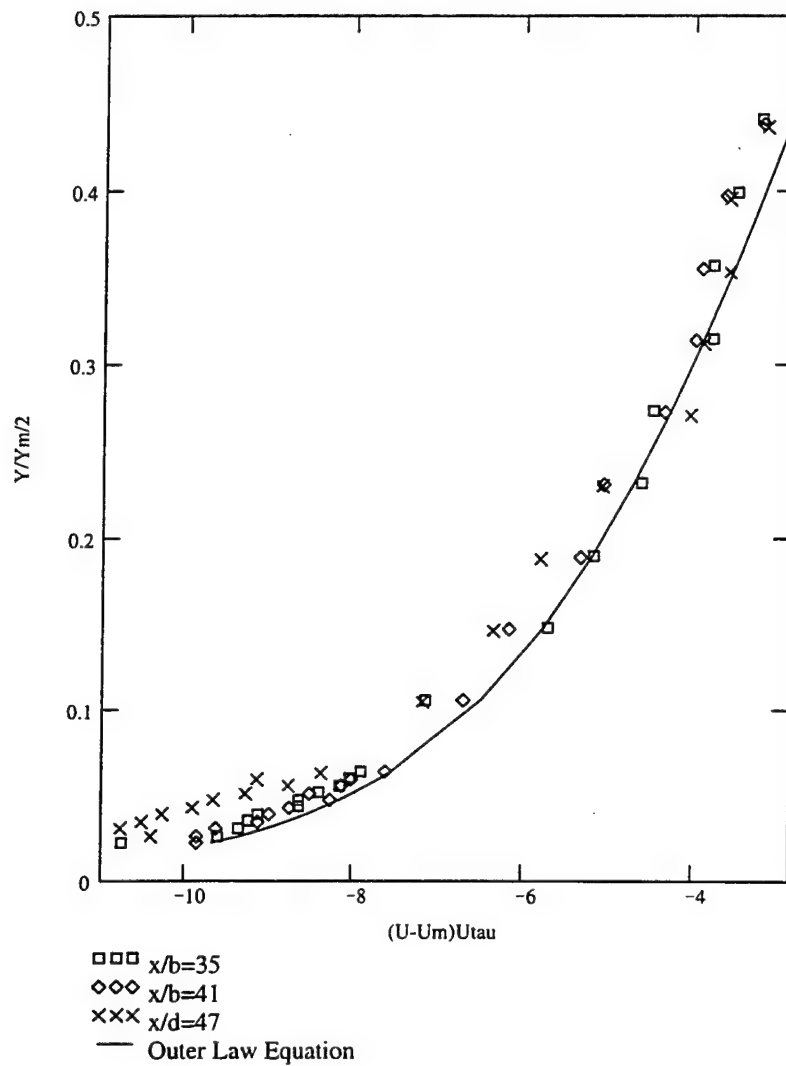


Figure 5-38. Near-Wall Velocity Profile Scaling Using Outer Law Variables

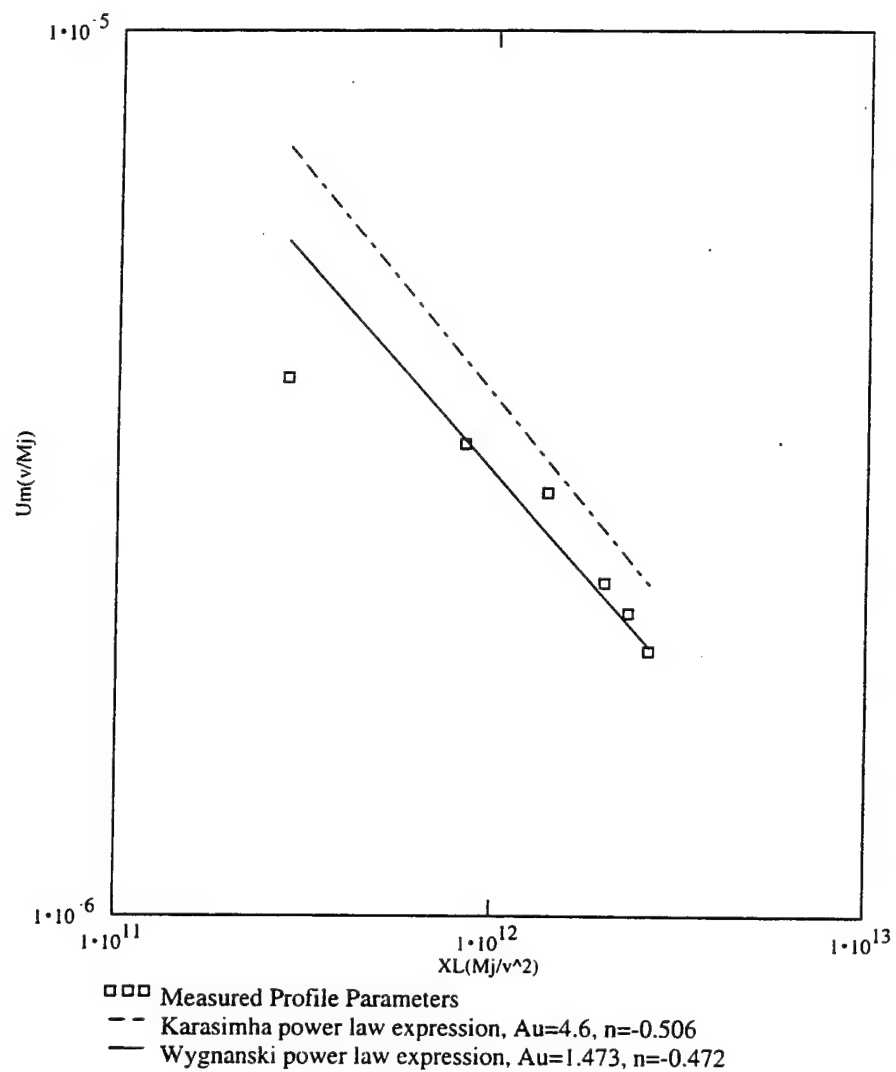


Figure 5-39. Wall Jet Momentum Scaling

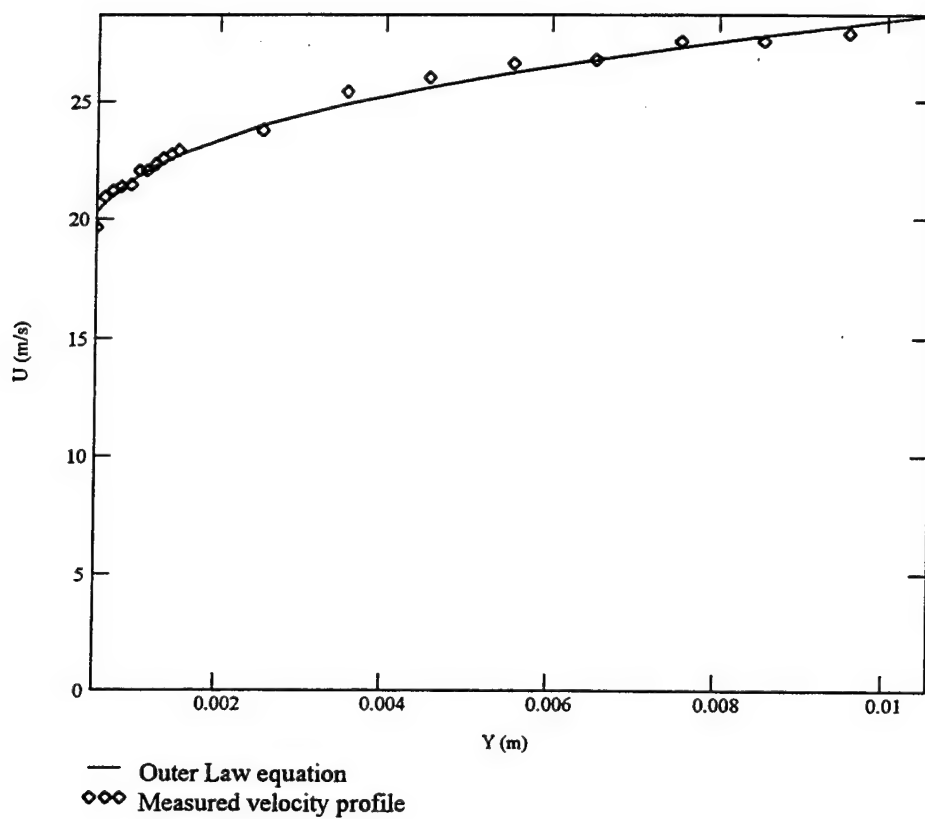


Figure 5-40. Comparison of Velocity Profile Generated with Outer Law Equation and Measured Velocity Profile ( $x/b = 35$ ,  $U_{inf} = 55.8$  m/s)

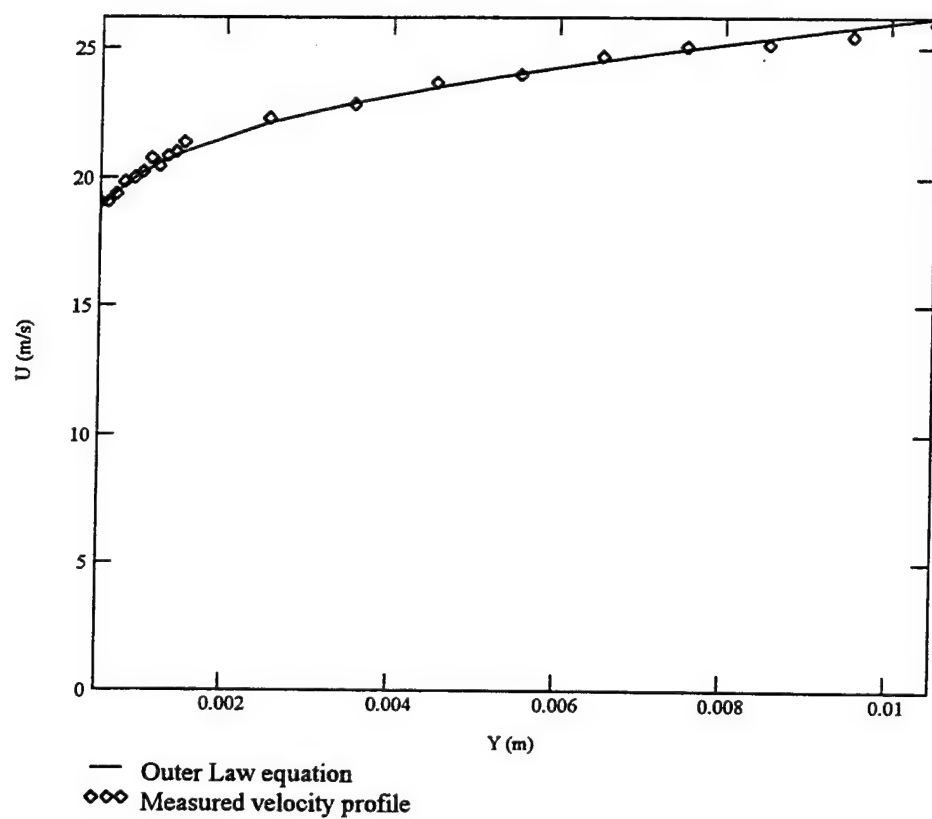


Figure 5-41. Comparison of Velocity Profile Generated with Outer Law Equation and Measured Velocity Profile ( $x/b = 41$ ,  $U_{inf} = 55.8$  m/s)

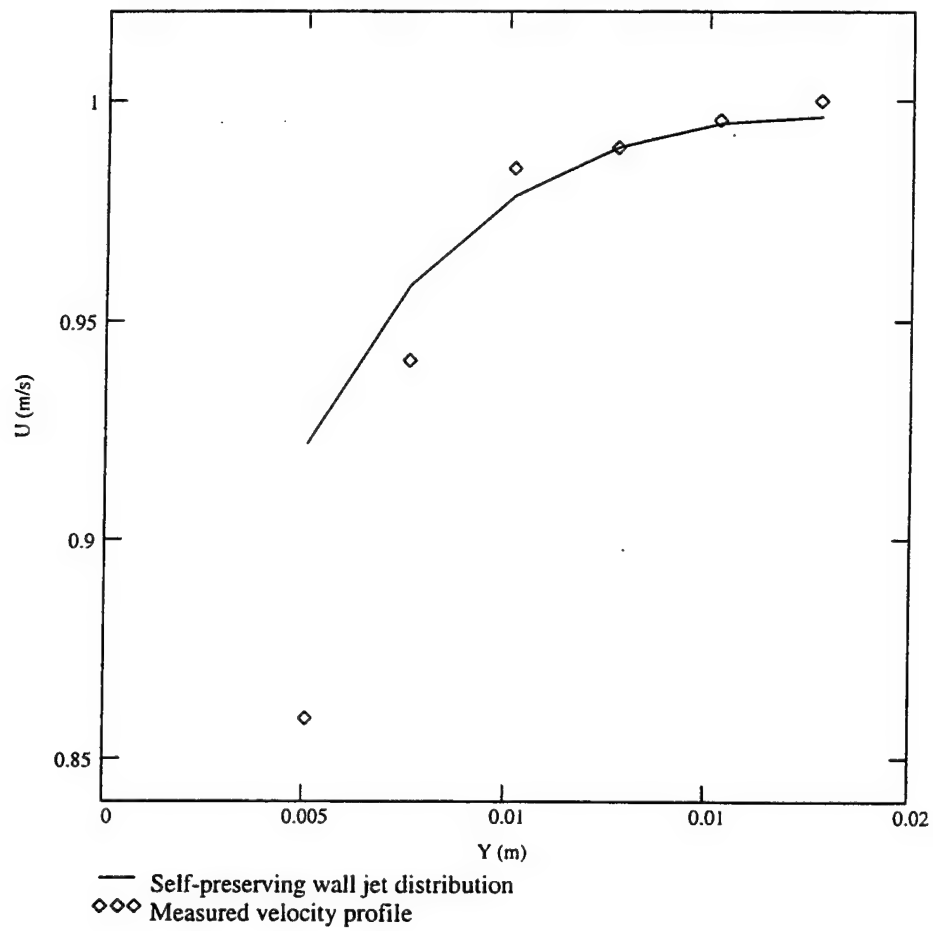


Figure 5-42 Velocity Profile Distribution Fit ( $x/b = 5.03$ ,  $U_{inf} = 55.8 \text{ m/s}$ )

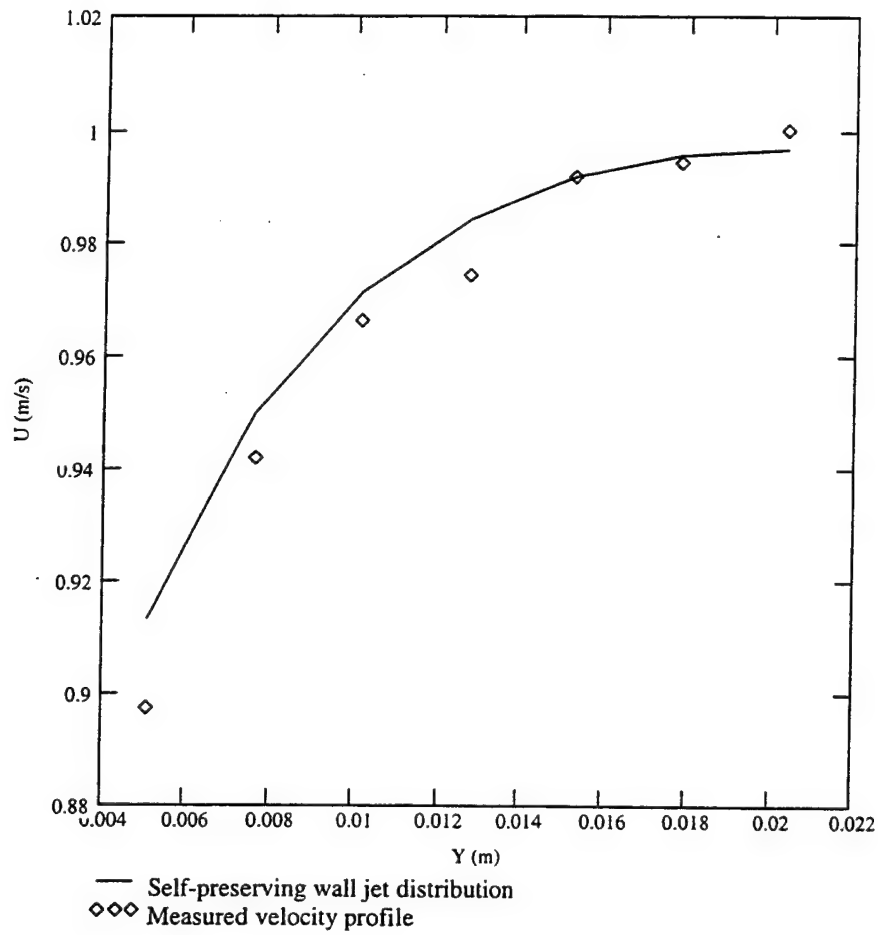


Figure 5-43. Velocity Profile Distribution Fit ( $x/b = 10.06$ ,  $U_{inf} = 55.8$  m/s)



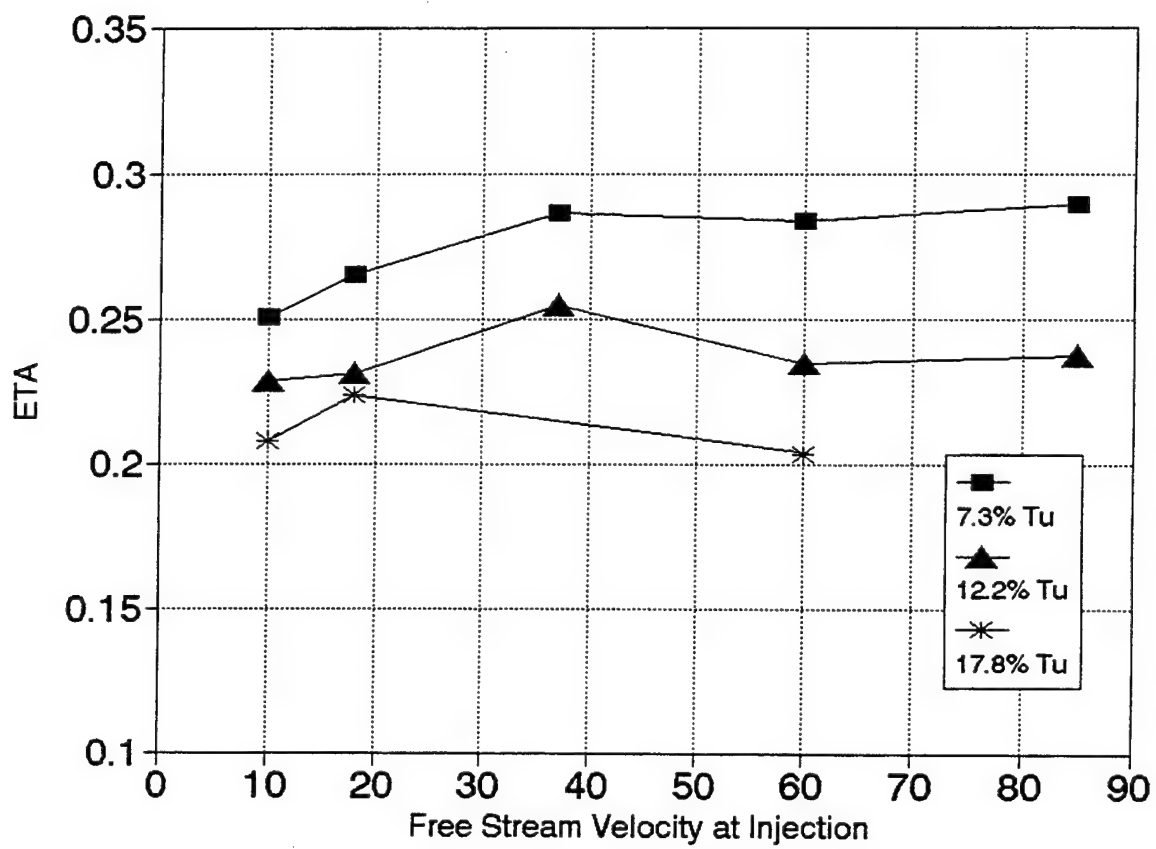
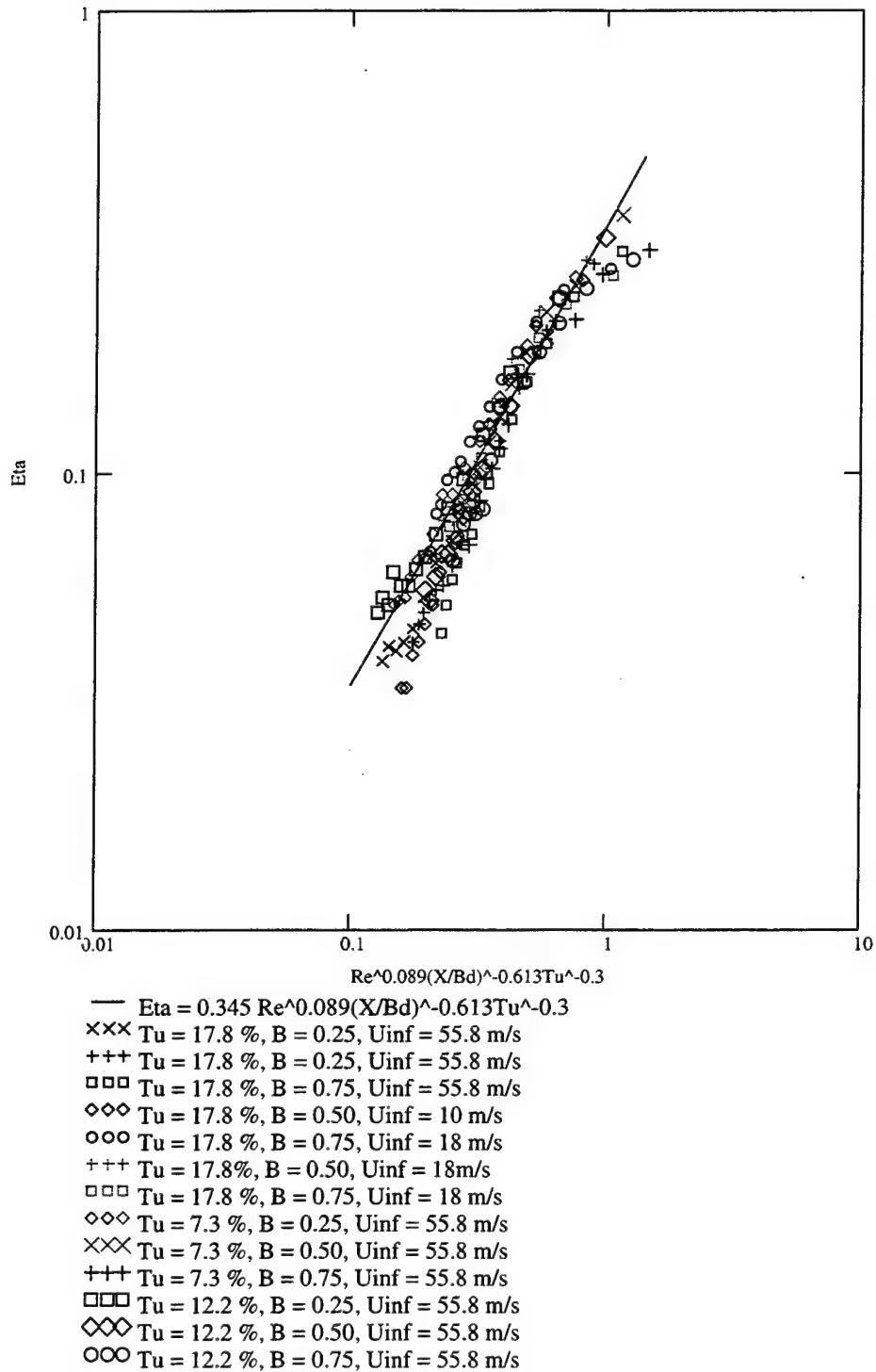
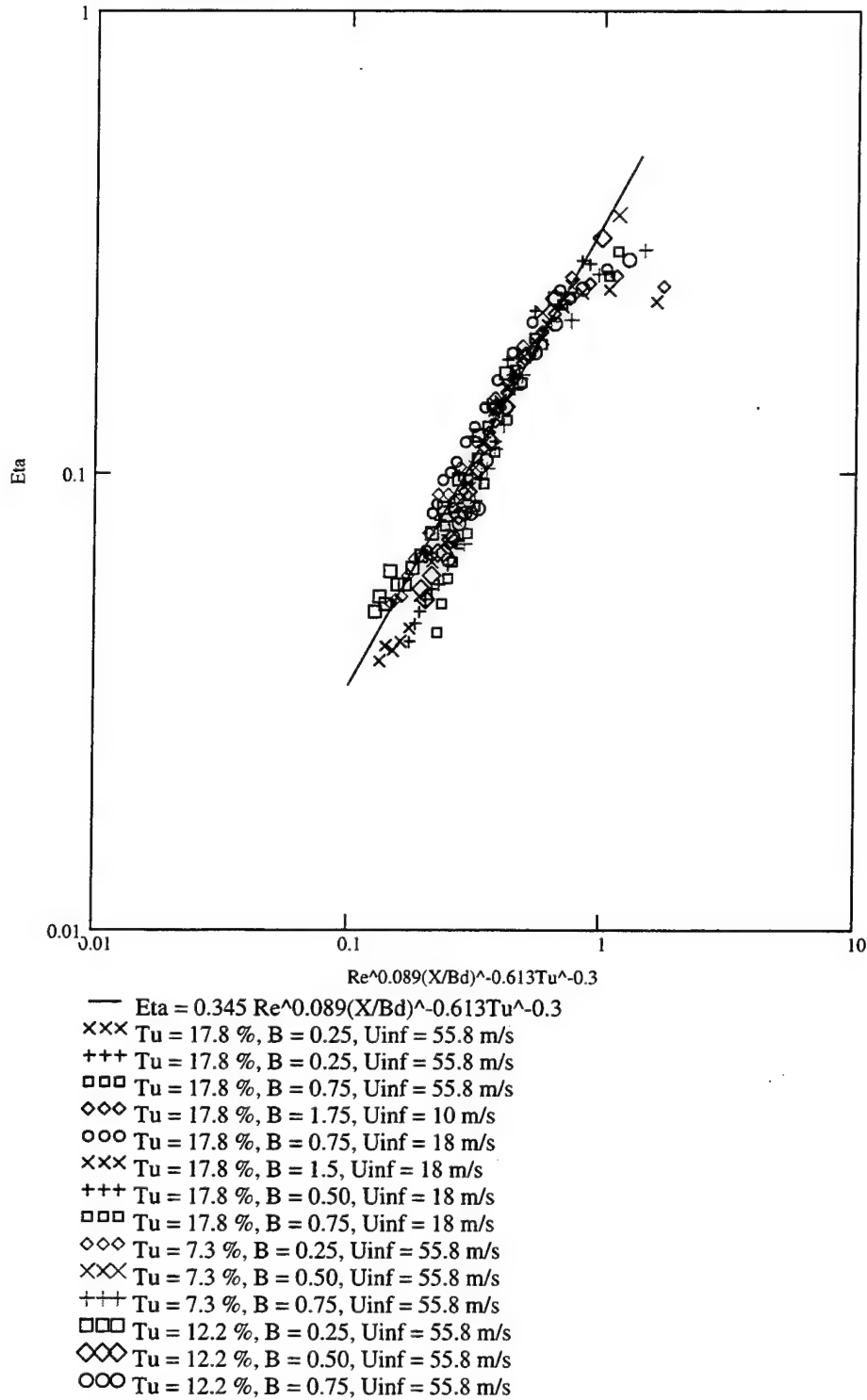


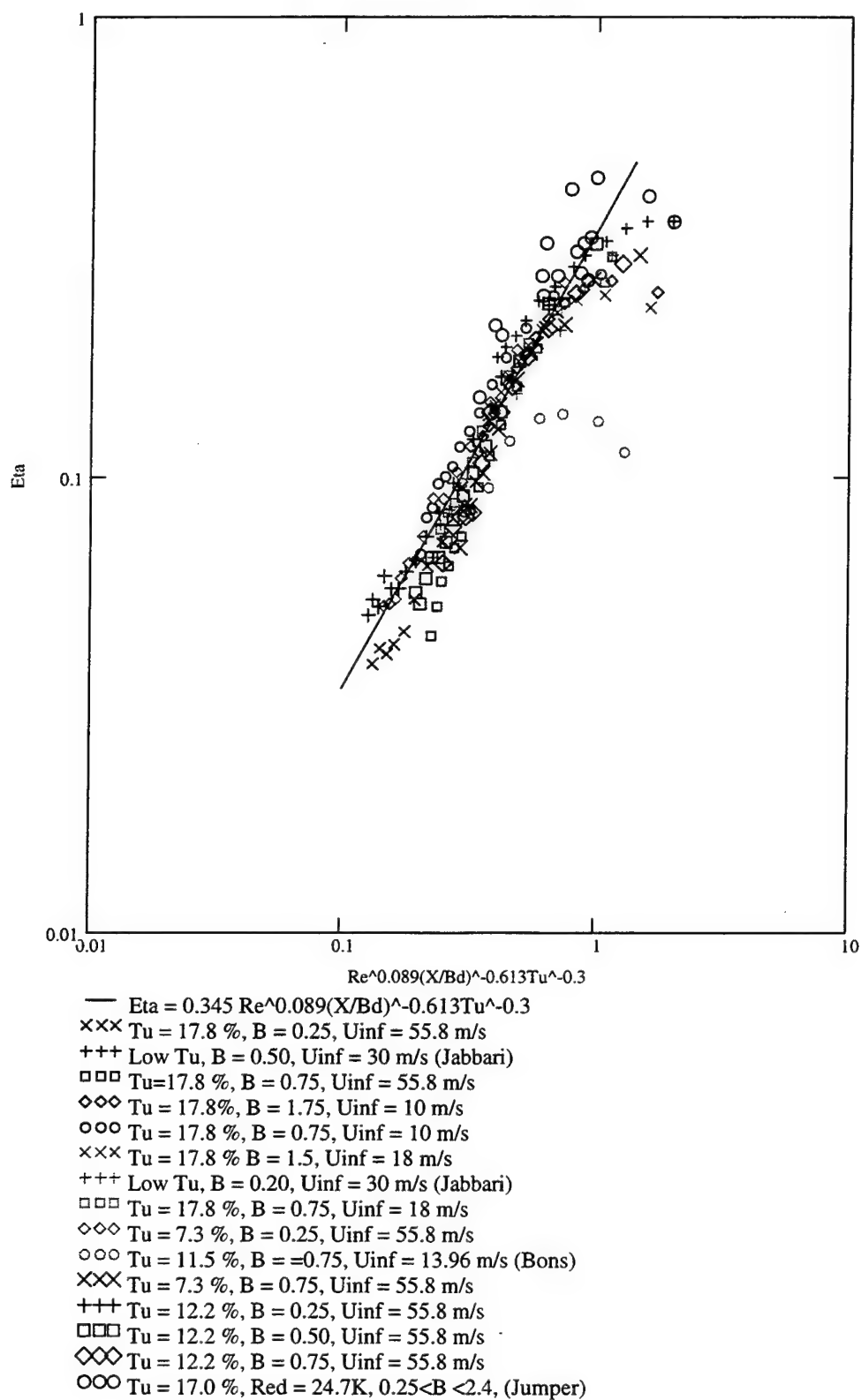
Figure 5-44. Effect of Free Stream Velocity on Film Cooling Effectiveness  
 $X/D = 10$ ,  $B = 0.50$



**Figure 5-45. Effectiveness Correlation with Non-dimensional Grouping of Film Cooling Parameters  $0.25 < B < 0.75$**



**Figure 5-46. Effectiveness Correlation with Non-dimensional Grouping of Film Cooling Parameters  $0.25 < B < 1.75$**



**Figure 5-47. Effectiveness Correlation Comparison Between Very Low and High Free Stream Tu and One and Two Rows of Holes**  
**0.20 < B < 2.4**

## Appendix A : Derivation of the Corrected Effectiveness Equation

To obtain the adiabatic effectiveness, surface temperature measurements were taken on a plate with no power applied. This set up produces a near-adiabatic condition. The term near-adiabatic is used because the plate will experience loss of energy through conduction and radiation despite the insulation placed under the plate and the low emissivity of the plate material.

A correction to account for the non-adiabatic effects was calculated and applied to each effectiveness value obtained. The correction was typically 1 to 5 % of the final effectiveness value with the largest percentages obtained towards the end of the plate.

From a one-dimensional heat balance on an element of the surface one obtains

$$q_{rad} + q_{cond} + q_{conv} = 0 \quad (A-1)$$

where  $q_{rad}$  is the radiated heat from the surface to the environment,  $q_{cond}$  is the conducted heat flux through the surface, and  $q_{conv}$  is the convective heat flux from the surface to the fluid. The above equation can be expressed in terms of the respective temperatures as

$$\varepsilon\sigma(T_w^4 - T_\infty^4) + h_o(T_w - T_\phi) + k\left(\frac{T_w - T_{bs}}{l}\right) = 0 \quad (A-2)$$

where  $T_w$  is the measured surface temperature,  $T_\infty$  is the free stream temperature,  $T_\phi$  is the temperature resulting from a mixture of the free stream temperature and the coolant temperature ( i.e., the temperature  $T_w$  would be if the plate was truly adiabatic),  $T_{bs}$  is the temperature at the bottom surface of the plate, and  $h_o$  is the heat transfer coefficient with injection. Defining  $\Delta T = T_w - T_\phi$  and solving yields

$$\Delta T = - \left( \frac{\varepsilon \sigma (T_w^4 - T_\infty^4) + k \left( \frac{T_w - T_{bs}}{l} \right)}{h_o} \right) \quad (A-3)$$

The adiabatic wall temperature  $T_{aw}$  can be expressed as the combination of the measured temperature and  $\Delta T$ ,  $T_{aw} = T_w - \Delta T$ . Substituting this expression in the adiabatic effectiveness equation

$$\eta = \frac{(T_{aw} - T_\infty)}{(T_c - T_\infty)} \quad (A-4)$$

gives

$$\eta = \frac{(T_w - \Delta T - T_\infty)}{(T_c - T_\infty)} \quad (A-5)$$

Finally substituting equation A-3 into A-5 yields the corrected effectiveness

$$\eta = \frac{(T_w - T_\infty)}{(T_c - T_\infty)} + \frac{(q_{rad} + q_{cond})}{h_o (T_c - T_\infty)} \quad (A-6)$$

where

$$q_{rad} = \varepsilon \sigma (T_w^4 - T_\infty^4)$$

$$q_{cond} = k \frac{(T_w - T_{bs})}{\ell}$$

The heat transfer coefficient in equation A-6 was obtained from a Stanton number approximation due to Kays and Crawford (1980)

$$St_o Pr^{0.4} = 0.0287 Re_x^{-0.2} \quad (A-7)$$

This approximation was adjusted for the appropriate level of free stream turbulence using a relation by Simonich (1978)

$$\frac{St}{St_o} = 1 + 5Tu \quad (A-8)$$

## Appendix B : Description of the Uncertainty

The uncertainty present in the calculation of effectiveness was described by a method for single-sample experiments presented by Kline and McClintock (1953). In this method the uncertainty interval for the result (in this case effectiveness) is

$$w_R = \left[ \left( \frac{\partial R}{\partial v_1} w_1 \right)^2 + \left( \frac{\partial R}{\partial v_2} w_2 \right)^2 + \dots + \left( \frac{\partial R}{\partial v_n} w_n \right)^2 \right]^{1/2} \quad (B-1)$$

where the result  $R$  is a function of  $n$  independent variables,  $v_1, v_2, \dots, v_n$ , and  $w_1, w_2, \dots, w_n$  are the uncertainty intervals for the independent variables. This equation is referred to as the second-power equation.

The function  $R$  for this experiment is the effectiveness equation expressed in terms of directly measured parameters as:

$$\eta = \frac{T_w - \left[ T_{\tan k} - \frac{(r-1)}{2C_p} \left( \frac{T_{hw}}{P_{bar}} - \frac{110.42 P_{kiel}^{3/2} T_{hw}}{P_{bar}^2} \right)^2 \right]}{T_{\tan kcool} - \frac{R_{air} (r-1) CD^2 P_{\tan kcool} T_{\tan kcool}}{C_p (P_{\tan kcool} + P_{bar})} - \left[ T_{\tan k} - \frac{(r-1)}{2C_p} \left( \frac{T_{hw}}{P_{bar}} - \frac{110.42 P_{kiel}^{3/2} T_{hw}}{P_{bar}^2} \right)^2 \right]}$$

where

$T_w$  is the temperature measured at the surface of the plate

$T_{\tan k}$  is the total temperature measured in the air supply tank

$T_{\tan kcool}$  is the total temperature measured in the coolant tank

$T_{bw}$  is the temperature measured in the free stream at the maximum velocity point



$P_{\text{tankcool}}$  is the total pressure measured in the coolant tank

$P_{\text{bar}}$  is the ambient barometric pressure

$P_{\text{kiel}}$  is the pressure measured at the maximum velocity point

CD is the coolant coefficient of discharge

Table B-1 presents the respective uncertainty intervals for each measured variable. The uncertainty intervals were estimated base on knowledge acquired through experience using instrumentation devices.

TABLE B-1. Uncertainty intervals for the independent variables in the effectiveness equation

Variable	Uncertainty Interval
$T_w$	$\pm 0.1^\circ\text{K}$
$T_{\text{tank}}$	$\pm 0.1^\circ\text{K}$
$T_{\text{tankcool}}$	$\pm 0.1^\circ\text{K}$
$T_{\text{hw}}$	$\pm 0.1^\circ\text{K}$
$P_{\text{tankcool}}$	$\pm 260 \text{ Pa}$
$P_{\text{bar}}$	$\pm 130 \text{ Pa}$
$P_{\text{kiel}}$	$\pm 260 \text{ Pa}$

The uncertainty interval for the effectiveness was calculated for three different combinations of  $U_\infty$ , Tu, B, and x/d. Uncertainty intervals along with the respective combination of parameters are presented in the following table:

TABLE B-2. Uncertainty intervals for the effectiveness at specific locations and conditions

$U_\infty$ (m/s)	Tu (%)	B	x/d	Uncertainty interval
18	17.8	1.75	10	0.008
37	12.2	0.25	50	0.015
10	7.3	0.25	5	0.021

## **Appendix C : Data Reduction Program**

This Appendix contains the Q-Basic program used to perform cooling jet calculations, main flow calculations and flow calculations at nozzle exit

```

DEC! ARE SUB Tempss (filename$, Pbar!, Thw!, PrAvg!(), Temp!(), Qconv!())
DECLARE SUB FC (filename$, Pbar!, Tinf!, Tmpf!, TankTempF!, CoolTankTempF!, Temp!(), PrAvg!(), Cmnt2$, Twavg!, Tstat!, Tstatcool!,
DECLARE FUNCTION Cpl (Temp!)
DECLARE SUB StaAvg (filename$, Temp!(), chan!(), NTC3497%, NTC3852%)
'Created by Dino Ishikura
Data reduction for TC
LS : COLOR 11, 0
CONST Rair = 287.074          'Gas constant for air J/(kg.K)
CONST HgtoPa = 3386.4        'convert pressure from in_Hg to Pa
CONST H2OtoPa = 248.7        'convert pressure from in_H2O to Pa
CONST rec = .862
OPTION BASE 1
n% = 150          ' Max number of thermocouple
DIM chan(n%), Temp(n%), X(n%), Y(n%), Z(n%), Ttype(n%), Status(n%)
DIM Qrad(n%), Qconv(n%)
DIM PrVAvg(3), PrAvg(3)
DIM StaTback(16), StaTsum(16), StaTavg(16), StaTcount%(16), StaTcen(16)
INPUT "Enter directory name <c:\filmcool\>: ", dirname$
IF dirname$ = "" THEN
    'dir$ = "c:\filmcool\"
    dir$ = "c:\filmcool\v10\"
ELSE
    dir$ = dirname$
END IF
dir$ = "c:\filmcool\3ht60_60\"

INPUT "Enter initial file number: ", init%
INPUT "Enter final file number: ", final%
CLS : COLOR 14, 0
PRINT "Main Menu"
PRINT " 1 - Process temperature for spreadsheet      "
PRINT " 2 - Process station temperature average      "
PRINT " 3 - Process Film Cooling data                  "
INPUT "Select option #: ", opt%
'opt% = 3
INPUT "Enter output file name: ", outname$
ou1% = FREEFILE
OPEN dir$ + outname$ + ".dat" FOR OUTPUT AS #ou1%
ou2% = FREEFILE
OPEN dir$ + "f" + outname$ + ".dat" FOR OUTPUT AS #ou2%
'sta% = 15
sta% = 0
FOR k% = init% TO final%
    'sta% = sta% - 1
    sta% = sta% + 1
    ans$ = LTRIM$(STR$(k%))
    filename$ = dir$ + ans$
    filein$ = filename$ + ".hftcd"
    Unit% = FREEFILE
    OPEN filein$ FOR INPUT AS #Unit%
    DO
        IF NOT EOF(Unit%) THEN
            INPUT #Unit%, Title1$
            WRITE Title1$
            INPUT #Unit%, Ftype%, Filnum, Id$, TIME1$, DATE1$, Cmnt1$, Cmnt2$
            WRITE Ftype%, Filnum, Id$, TIME1$, DATE1$, Cmnt1$, Cmnt2$
        '
        ' /Exprm/ Experiment conditions
        '
        INPUT #Unit%, Pbar, Tinf, Jetdp, CoolJetdp, Tmpf, Probetemp, TankTempF, CoolTankTempF
        WRITE Pbar, Tinf, Jetdp, CoolJetdp, Tmpf, Probetemp, TankTempF, CoolTankTempF
        '
        ' /PlateT/, /Table/ Heated plate data
        '
        INPUT #Unit%, NStations%, NperStation%, NTCtotal%, NTC3497%, NTC3852%
        WRITE NStations%, NperStation%, NTCtotal%, NTC3497%, NTC3852%
        '
        ' /TC Temperature Data/
        '
        FOR indx% = 1 TO NTCtotal%
            INPUT #Unit%, chan(indx%), Temp(indx%), X(indx%), Y(indx%), Z(indx%), Ttype(indx%), Status(indx%), Qrad(indx%), Qconv(i
            WRITE #Unit1%, chan(indx%), Temp(indx%), X(indx%), Y(indx%), Z(indx%), Ttype(indx%), Status(indx%), Qrad(indx%), Qconv
        NEXT indx%
        '
        ' /Pressure Data/
        '
        INPUT #Unit%, Title2$
        WRITE Title2$
        FOR i% = 1 TO 3
            INPUT #Unit%, PrVAvg(i%), PrAvg(i%)
            WRITE PrVAvg(i%), PrAvg(i%)
        NEXT i%
        '
    '

```

```

' /Heated table parameters/
INPUT #Unit%, Title3$
WRITE Title3$

INPUT #Unit%, PlateV, PlateArea, RShunt, ShuntV, TPlateAvg
WRITE PlateV, PlateArea, RShunt, ShuntV, TPlateAvg
ELSE
EXIT DO
END IF
LOOP
CLOSE #Unit%
SELECT CASE opt%
CASE 1
CALL Tempss(filename$, Pbar, Tmpf, PrAvg(), Temp(), Qconv())
CASE 2
CALL StaAvg(filename$, Temp(), chan(), NTC3497%, NTC3852%)
CASE 3
'CALL FC(filename$, Pbar, Tinf, Tmpf, TankTempF, CoolTankTempF, Temp(), PrAvg(), Cmnt2$, eta, sta%)
CALL FC(filename$, Pbar, Tinf, Tmpf, TankTempF, CoolTankTempF, Temp(), PrAvg(), Cmnt2$, Tw, Tinf, Tcool, eta, sta%)
'xd = ((sta% - 1) * 3.63 + 83.31) / 2.54
'IF sta% = 0 THEN eta = 0
'PRINT #ou1%, filename$, xd, eta
PRINT #ou1%, filename$, Tw, Tinf, Tcool, eta
PRINT #ou2%, sta%, Tinf
CASE ELSE: BEEP
END SELECT
NEXT k%
CLOSE #ou1%, #ou2
BEEP: PRINT : PRINT ">>>> DONE <<<<"
END

```

```

FUNCTION Cp (Temp)
' Curve fit using: Tmin = 250 K
' Tmax = 500 K
' Reference: Table A.1 (Kays/Crawford)
Cp = 1022! - .167572 * Temp + .000360722# * Temp * Temp
END FUNCTION

```

```

SUB FC (filename$, Pbar, Tinf, Tmpf, TankTempF, CoolTankTempF, Temp(), PrAvg(), Cmnt2$, Twavg, Tstat, Tstatcool, eta, sta%)

Ptankmain1 = PrAvg(1)
IF Ptankmain1 < 0 THEN Ptankmain1 = 0
PKiel1 = PrAvg(2)
IF PKiel1 < 0 THEN PKiel1 = 0
Ptankcool1 = PrAvg(3)
IF Ptankcool1 > .2 THEN
CD = .66 - .1532 / Ptankcool1 + .01634 / (Ptankcool1 * Ptankcool1)
ELSE
Ptankcool1 = 0
CD = 0
END IF

'Convert pressure from inH2O to Pascal
Pbar = Pbar * HgtoPa
Ptankmain = Ptankmain1 * H2OtoPa
PKiel = PKiel1 * H2OtoPa
Ptankcool = Ptankcool1 * H2OtoPa

'rename to mnemonic temperature names
Thw1 = Tmpf
Ttanklarge1 = TankTempF
Ttankcool1 = CoolTankTempF
Tceiling1 = Tinf

'Convert temperature from deg Farenheit to deg Kelvin
Thw = (Thw1 - 32) / 1.8 + 273.15
Ttanklarge = (Ttanklarge1 - 32) / 1.8 + 273.15
Ttankcool = (Ttankcool1 - 32) / 1.8 + 273.15
Tceiling = (Tceiling1 - 32) / 1.8 + 273.15

' Cooling jet calculation
Denstankcool = (Ptankcool + Pbar) / (Rair * Ttankcool)
VjetcoolTheo = SQR(2! * Ptankcool / Denstankcool)
Vjetcool = CD * VjetcoolTheo
Tstatcool = Ttankcool - Vjetcool * Vjetcool / (2 * Cp(Ttankcool))
Densjetcool = Pbar / (Rair * Tstatcool)
Gcool = Densjetcool * Vjetcool

' Main flow calculation
' n iterations over main flow static temperature
n% = 3
Dens = Pbar / (Rair * Thw)
Vel = SQR(2! * PKiel / Dens) 'first approach
Tstat = Thw - rec * Vel * Vel / (2 * Cp(Thw))

FOR i% = 1 TO n%

```

```

PkielSTP = Pkiel * (101325 / Pbar) * (Tstat / 288.15)
Dens = Pbar / (Rair * Tstat)
Vel = SQR(2 * PkielSTP / Dens) 'first approach
Tstat = Thw - rec * Vel * Vel / (2 * Cp(Tstat))
NEXT i%
Ginf = Dens * Vel
IF Ginf <> 0 THEN
  B = Gcool / Ginf
ELSE
  B = 0
END IF

' Flow calculation at nozzle exit
Denstanklarge = (Pbar + Ptankmain) / (Rair * Ttanklarge)
Velexit = SQR(2 * Ptankmain / Denstanklarge) 'first approach
INPUT "Enter TC station #: ", sta%
PRINT filename$
ou% = FREEFILE
OPEN filename$ + "fc.dat" FOR OUTPUT AS #ou%
IF sta% = 0 THEN
  WRITE #ou%, Pbar / HgtoPa, DATE$, TIME$
  PRINT #ou%, "          Lge Tank    Cool Tank    NozExit    Infinity    Cool jet    "
  PRINT #ou%, USING "Pressure (inH2O)  ###.##    ###.##    ###.##    ###.##    ###.##"; Ptankmain1; Ptankcool1; Pkiel1; Pt
  PRINT #ou%, USING "Temperature (K)    ###.##    ###.##    ###.##    ###.##    ###.##"; Ttanklarge; Ttankcool; Tstat; Tsta
  PRINT #ou%, USING "Density (kg/m^3)   ###.##    ###.##    ###.##    ###.##    ###.##"; Denstanklarge; Denstankcool; Dens
  PRINT #ou%, USING "Velocity (m/s)     ###.##    ###.##    ###.##    ###.##    ###.##"; Velexit; Vel; Vjetcool
  PRINT #ou%, USING "Discharge Coefficient = ###.## "; CD
  PRINT #ou%, USING "Gcool = ###.## kg.m/s    Ginf = ###.## kg.m/s "; Gcool; Ginf
  PRINT #ou%, USING "Blowing ratio = ###.## "; B
ELSE
  PRINT #ou%, Cmnt2$
  WRITE #ou%, Pbar / HgtoPa, DATE$, TIME$
  PRINT #ou%,
  PRINT #ou%, "Station Temperature in deg. Farenheit"
  Twavg = 0: n% = 0
  FOR i% = 1 TO 7
    PRINT #ou%, Temp((sta% - 1) * 8 + i%);
    T = Temp((sta% - 1) * 8 + i%)
    IF T > 32 AND T < 120 THEN
      Twavg = Twavg + Temp((sta% - 1) * 8 + i%)
      n% = n% + 1
    ELSE
      Temp((sta% - 1) * 8 + i%) = 0
    END IF
  NEXT i%
  PRINT #ou%,
  IF n% <> 0 THEN Twavg = (Twavg / n% + 459.95) / 1.8 'Kelvin
  PRINT #ou%, Cmnt2$
  WRITE #ou%, Pbar / HgtoPa, DATE$, TIME$
  PRINT #ou%,
  PRINT #ou%, "Station "; sta%; " - Avg wall temperature = "; Twavg
  PRINT #ou%, "Static Flow Temp = "; Tstat
  PRINT #ou%, "Static cool jet Temp = "; Tstatcool
  eta = (Twavg - Tstat) / (Tstatcool - Tstat)
  PRINT #ou%, USING "FC effectiveness = ###.### %"; eta * 100
  PRINT #ou%,
  PRINT #ou%, "; Lge; Tank; Cool; Tank; NozExit; Infinity; Cool; jet; "; ""
  PRINT #ou%, USING "Pressure (inH2O)  ###.##    ###.##    ###.##    ###.##    ###.##"; Ptankmain1; Ptankcool1; Pkiel1; Pt
  PRINT #ou%, USING "Temperature (K)    ###.##    ###.##    ###.##    ###.##    ###.##"; Ttanklarge; Ttankcool; Tstat; Tsta
  PRINT #ou%, USING "Density (kg/m^3)   ###.##    ###.##    ###.##    ###.##    ###.##"; Denstanklarge; Denstankcool; Dens
  PRINT #ou%, USING "Velocity (m/s)     ###.##    ###.##    ###.##    ###.##    ###.##"; Velexit; Vel; Vjetcool
  PRINT #ou%, USING "Discharge Coefficient = ###.## "; CD
END IF
END IF

CLOSE #ou%
END SUB

SUB StaAvg (filename$, Temp(), chan(), NTC3497%, NTC3852%)
DIM StaTback(16), StaTsum(16), StaTavg(16), StaTcount%(16), StaTcen(16)
fileout$ = filename$ + "hwt.dat"
Unit1% = FREEFILE
PRINT " >> processing ", filename$
OPEN fileout$ FOR OUTPUT AS #Unit1%

Tlow = 32
Thigh = 200
Xlimit = 100
NTCtotal% = NTC3497% + NTC3852%

FOR i% = 1 TO 16
  StaTcount%(i%) = 0
  StaTsum(i%) = 0
NEXT i%

'
' Calculate Station Average Temperature. First thermocouples were
' classified by station number. It was selected SELECT CASE command
' instead of formula in order to be more flexible to change HP

```

```

' channel number.
FOR indx% = 1 TO NTC3497% ' NTC3852%
  IF Temp(indx%) > Tlow AND Temp(indx%) < Thigh THEN
    SELECT CASE chan(indx%)
      CASE 0 TO 6: sta% = 1
      CASE 8 TO 14: sta% = 2
      CASE 16 TO 22: sta% = 3
      CASE 24 TO 30: sta% = 4
      CASE 32 TO 38: sta% = 5
      CASE 40 TO 46: sta% = 6
      CASE 48 TO 54: sta% = 7
      CASE 56 TO 62: sta% = 8
      CASE 64 TO 70: sta% = 9
      CASE 72 TO 78: sta% = 10
      CASE 80 TO 86: sta% = 11
      CASE 88 TO 94: sta% = 12
      CASE 7: StaTback(1) = Temp(indx%): GOTO ignore1
      CASE 15: StaTback(2) = Temp(indx%): GOTO ignore1
      CASE 23: StaTback(3) = Temp(indx%): GOTO ignore1
      CASE 31: StaTback(4) = Temp(indx%): GOTO ignore1
      CASE 39: StaTback(5) = Temp(indx%): GOTO ignore1
      CASE 47: StaTback(6) = Temp(indx%): GOTO ignore1
      CASE 55: StaTback(7) = Temp(indx%): GOTO ignore1
      CASE 63: StaTback(8) = Temp(indx%): GOTO ignore1
      CASE 71: StaTback(9) = Temp(indx%): GOTO ignore1
      CASE 79: StaTback(10) = Temp(indx%): GOTO ignore1
      CASE 87: StaTback(11) = Temp(indx%): GOTO ignore1
      CASE 95: StaTback(12) = Temp(indx%): GOTO ignore1
      CASE ELSE: GOTO ignore1
    END SELECT
    StaTcount%(sta%) = StaTcount%(sta%) + 1
    StaTsum(sta%) = StaTsum(sta%) + Temp(indx%)
  END IF
ignore1:
NEXT indx%

FOR indx% = NTC3497% + 1 TO NTCtotal%
  IF Temp(indx%) > Tlow AND Temp(indx%) < Thigh THEN
    SELECT CASE chan(indx%)
      CASE 0 TO 6: sta% = 13
      CASE 8 TO 14: sta% = 14
      CASE 16 TO 19, 100 TO 102: sta% = 15
      CASE 104 TO 110: sta% = 16
      CASE 7: StaTback(13) = Temp(indx%): GOTO ignore2
      CASE 15: StaTback(14) = Temp(indx%): GOTO ignore2
      CASE 103: StaTback(15) = Temp(indx%): GOTO ignore2
      CASE 111: StaTback(16) = Temp(indx%): GOTO ignore2
      CASE ELSE: GOTO ignore2
    END SELECT
    StaTcount%(sta%) = StaTcount%(sta%) + 1
    StaTsum(sta%) = StaTsum(sta%) + Temp(indx%)
  END IF
ignore2:
NEXT indx%

' PRINT "i%, StaTsum(i%), StaTcount%(i%), StaTavg(i%), StaTcen(i%), StaTback(i%)"
FOR i% = 1 TO 16
  StaTavg(i%) = StaTsum(i%) / StaTcount%(i%)
  StaTcen(i%) = Temp((i% - 1) * 8 + 4)
  'WRITE i%, StaTsum(i%), StaTcount%(i%), StaTavg(i%), StaTcen(i%), StaTback(i%)
  'WRITE #Unit1%, i%, StaTsum(i%), StaTcount%(i%), StaTavg(i%), StaTcen(i%), StaTback(i%)
  'WRITE #Unit1%, i%, StaTavg(i%), StaTcen(i%), StaTback(i%)
NEXT i%
CLOSE #Unit1%
END SUB

SUB Tempss (filename$, Pbar, Thw, PrAvg(), Temp(), Qconv())
  Pkiel = PrAvg(3) * H2OtoPa
  Pbar = Pbar * HgtoPa
  Thw = (Thw - 32) / 1.8 + 273.15

  fileout$ = filename$ + "hft.prn"
  Unit1% = FREEFILE
  LOCATE 10, 1: PRINT " >> processing ", filename$

  OPEN fileout$ FOR OUTPUT AS #Unit1%

  Tlow = 32
  Thigh = 200
  Xlimit = 100
  rint Temperature to a file in 16 columns
  R i% = 1 TO 8
  'PRINT i% - 4;
  PRINT #Unit1%, i% - 4;
  FOR j% = 1 TO 16
    indx% = (j% - 1) * 8 + i%
    IF Temp(indx%) > Tlow AND Temp(indx%) < Thigh THEN
      'PRINT Temp(indx%); ", ";

```

```

        PRINT #Unit1%, Temp(indx%); ", ";
    ELSE
        'PRINT CHR$(34); CHR$(34); ", ";
        PRINT #Unit1%, CHR$(34); CHR$(34); ", ";
    END IF
NEXT j%
'PRINT
PRINT #Unit1%,
NEXT i%
PRINT #Unit1%,

'print Qconv to a file in 16 columns
FOR i% = 1 TO 8
    'PRINT i% - 4;
    PRINT #Unit1%, i% - 4;
    FOR j% = 1 TO 16
        indx% = (j% - 1) * 8 + i%
        IF Temp(indx%) > Tlow AND Temp(indx%) < Thigh THEN
            'PRINT Temp(indx%); ", ";
            PRINT #Unit1%, Qconv(indx%); ", ";
        ELSE
            'PRINT CHR$(34); CHR$(34); ", ";
            PRINT #Unit1%, CHR$(34); CHR$(34); ", ";
        END IF
    NEXT j%
    'PRINT
    PRINT #Unit1%,
NEXT i%
Dens = Pbar / (Rair * Thw)
Vel = SQR(2 * Pkiel / Dens)      'first approach
Tstat = Thw - rec * Vel * Vel / (2 * Cp(Thw))
PkielSTP = Pkiel * (101325 / Pbar) * (Tstat / 288.15)
Dens = Pbar / (Rair * Tstat)
Vel = SQR(2 * PkielSTP / Dens)
Tstat = Thw - rec * Vel * Vel / (2 * Cp(Tstat))
WRITE #Unit1%, Vel, Tstat

CLOSE #Unit1%
END SUB

```

## Appendix D : Free stream recovery temperature reconstruction

The free stream recovery temperature,  $T_{r\infty}$ , was reconstructed using the total temperature (plenum temperature) and free stream velocity values. This reconstruction was necessary because the thermocouple used to measure the free stream recovery temperature was placed too close to the coolant stream and the temperature registered was not that of the coolant stream but of a mixture of free stream air and coolant.

The velocity term contribution was subtracted from the total temperature measured in the plenum and later added after multiplying the term by the recovery factor as follows:

$$T_{r\infty(new)} = T_{tan k} - \frac{U_{\infty}^2}{2C_p} + r \frac{U_{\infty}^2}{2C_p} \quad (D-1)$$

where

$T_{tan k}$  is the temperature measured inside the air supply plenum chamber

$U_{\infty}$  is the free stream velocity at each station

$r$  is the recovery factor (0.862)

Effectiveness was recalculated with the new free stream recovery temperatures for every combination of  $T_u$ ,  $B$  and  $U_{\infty}$ . The new effectiveness values were more in the order of what was expected based on other research seen, than the effectiveness obtained with the measured  $T_{r\infty}$  values.



## **Appendix E : Plots of Effectiveness with Film Cooling Parameters**

This Appendix contains figures of effectiveness with  $X/D$  and effectiveness with  $B$  for the combinations of  $Tu$ ,  $B$  and  $U_\infty$  not directly referenced in the main document. Figures are arranged by ascending order of free stream velocity at injection value first, and blowing ratio or station location second.

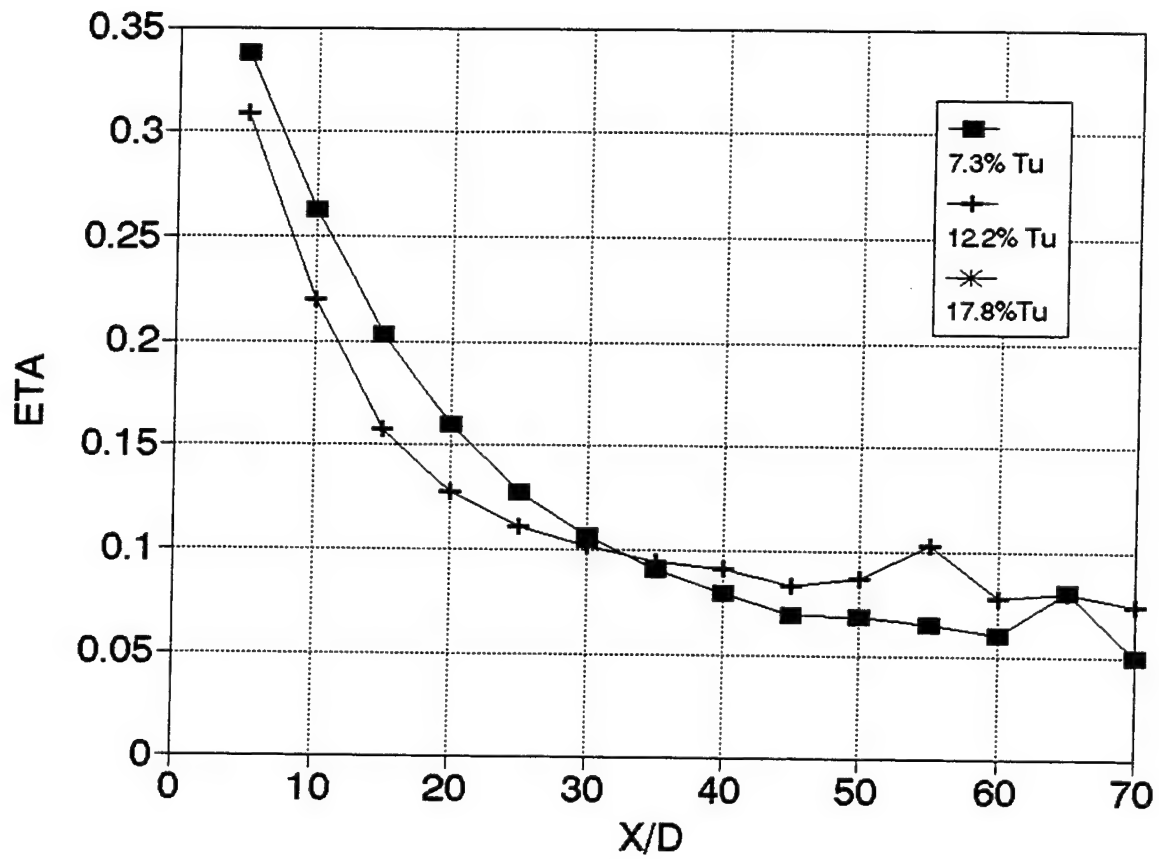


Figure E-1. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.25$ , Free stream velocity at injection = 10 m/s

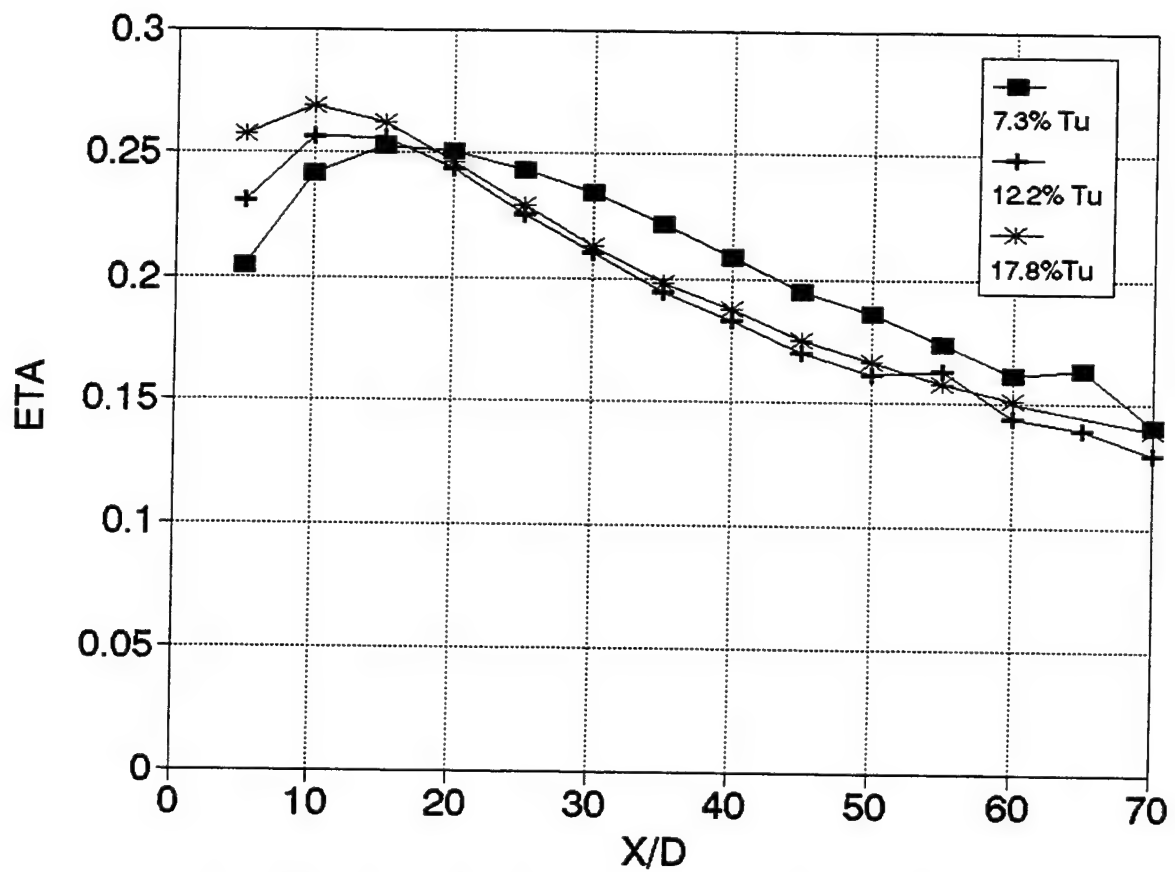


Figure E-2. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.50$ , Free stream velocity at injection = 10 m/s

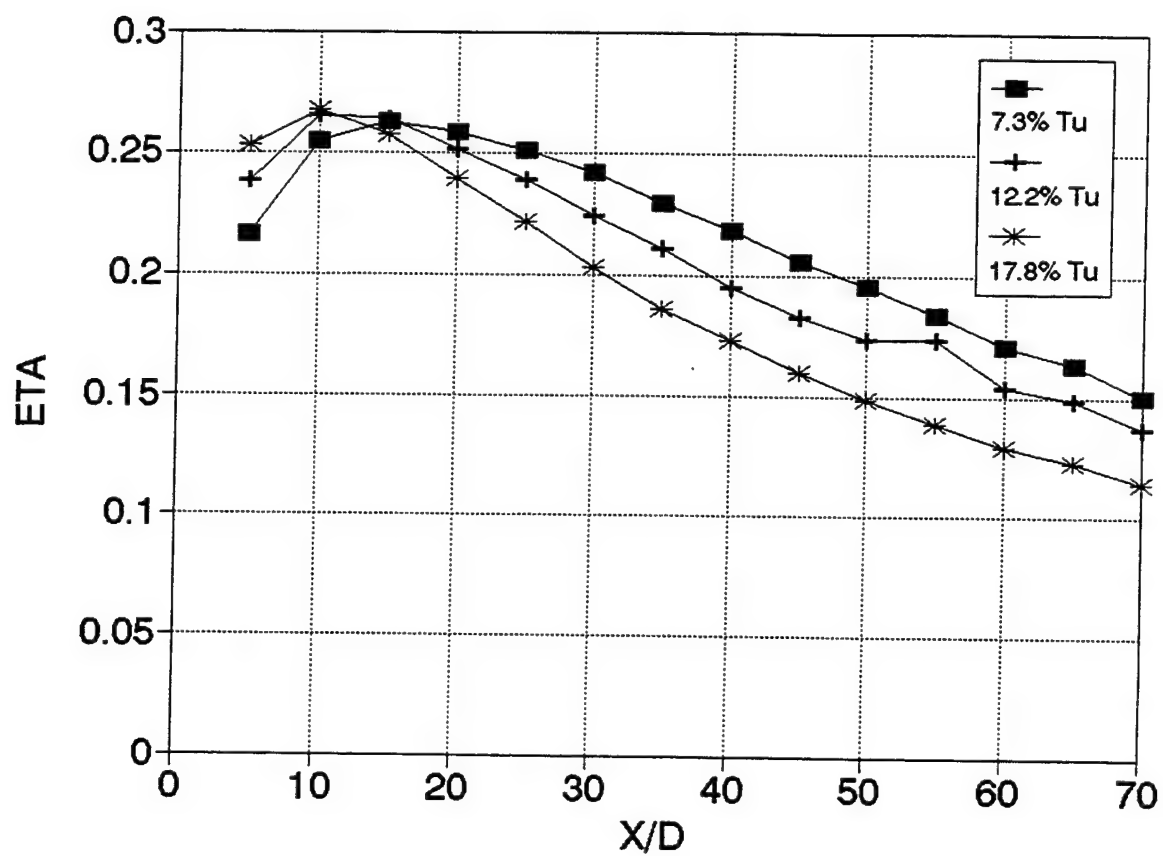


Figure E-3. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.75$ , Free stream velocity at injection = 10 m/s

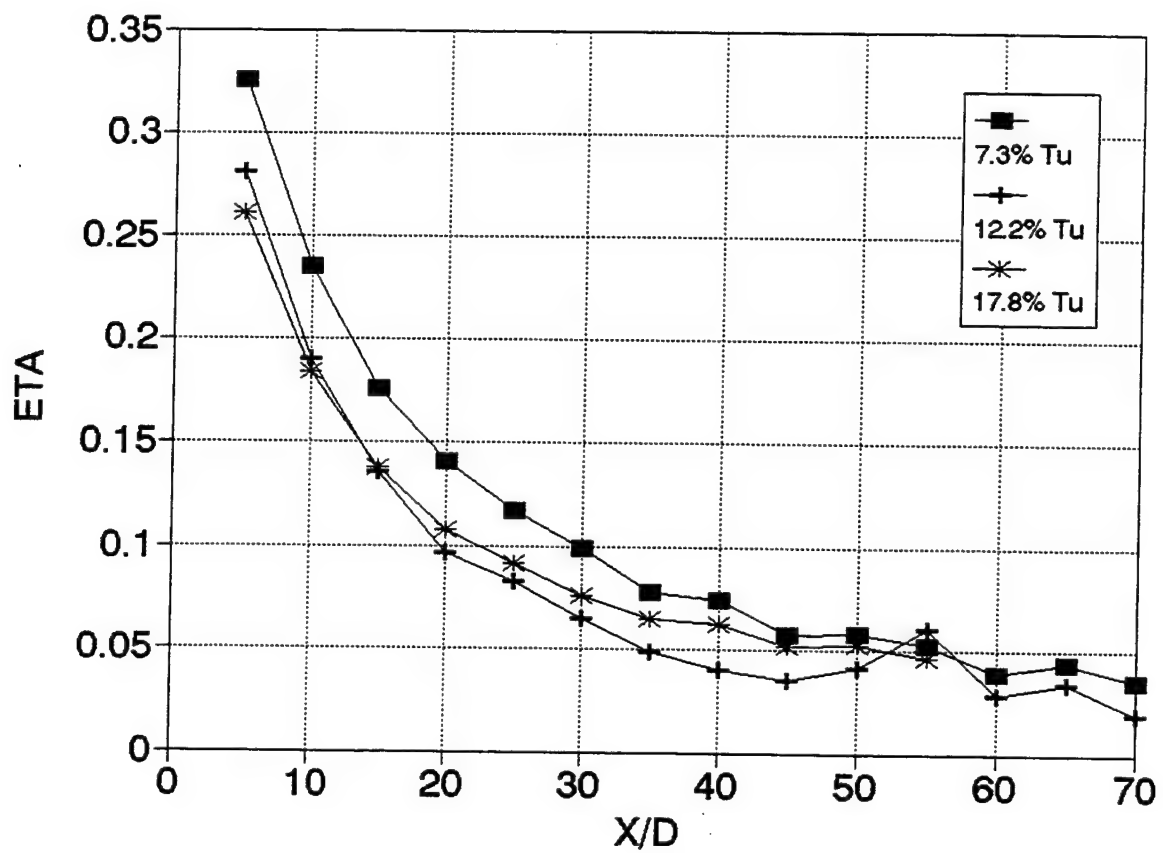


Figure E-4. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.25$ , Free stream velocity at injection = 18 m/s

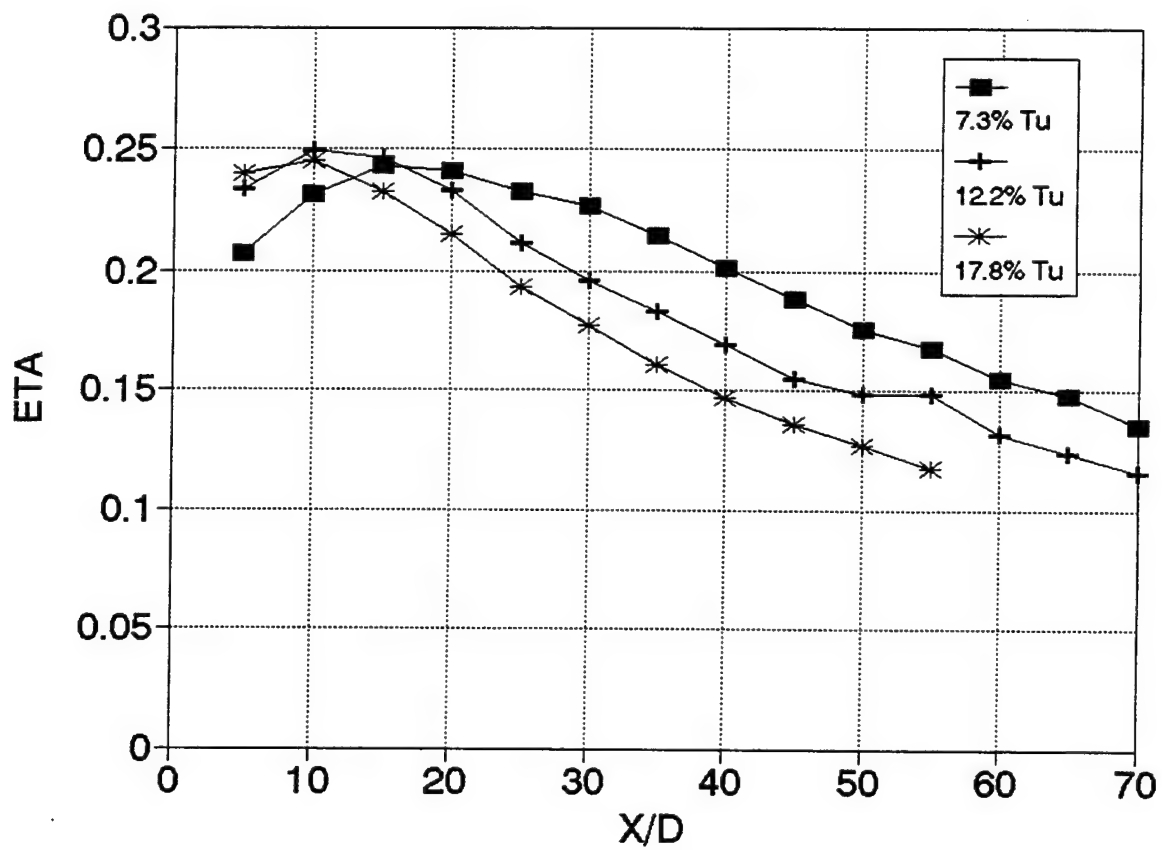


Figure E-5. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.25$ , Free stream velocity at injection = 18 m/s

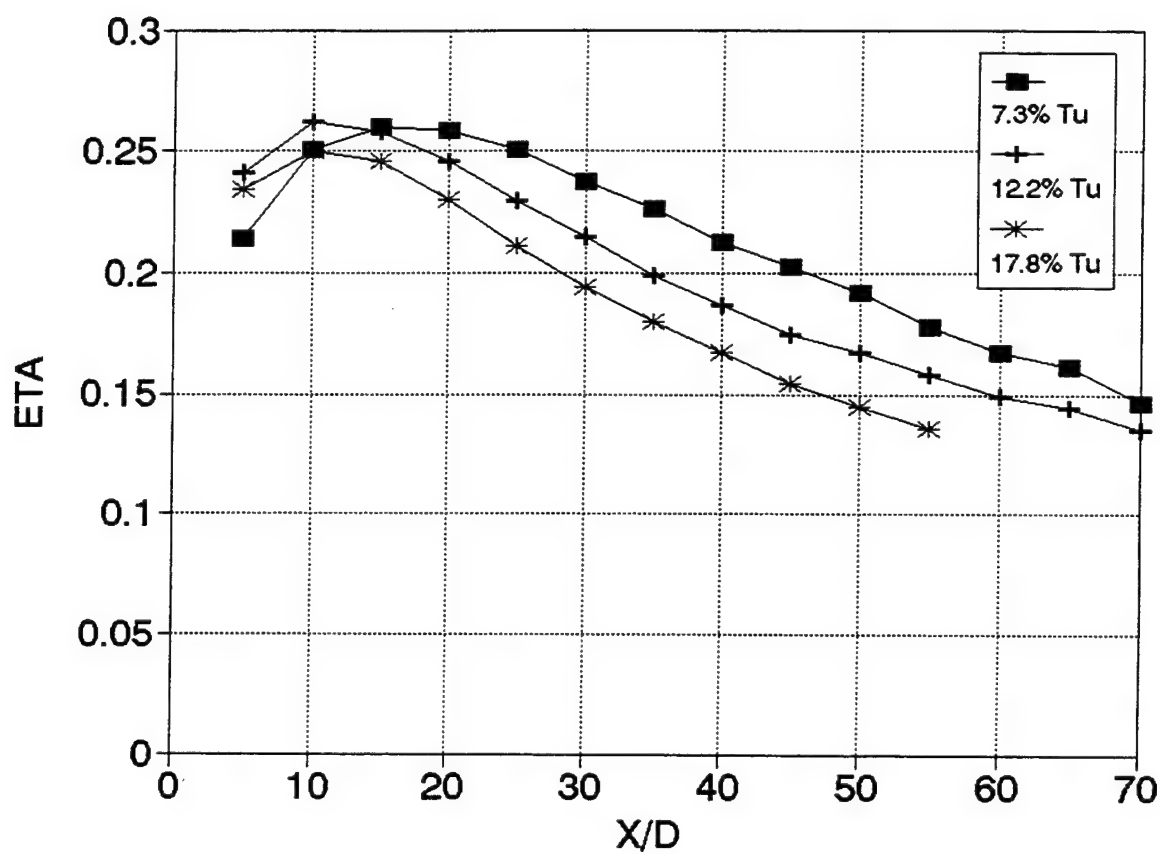


Figure E-6. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.50$ , Free stream velocity at injection = 18 m/s

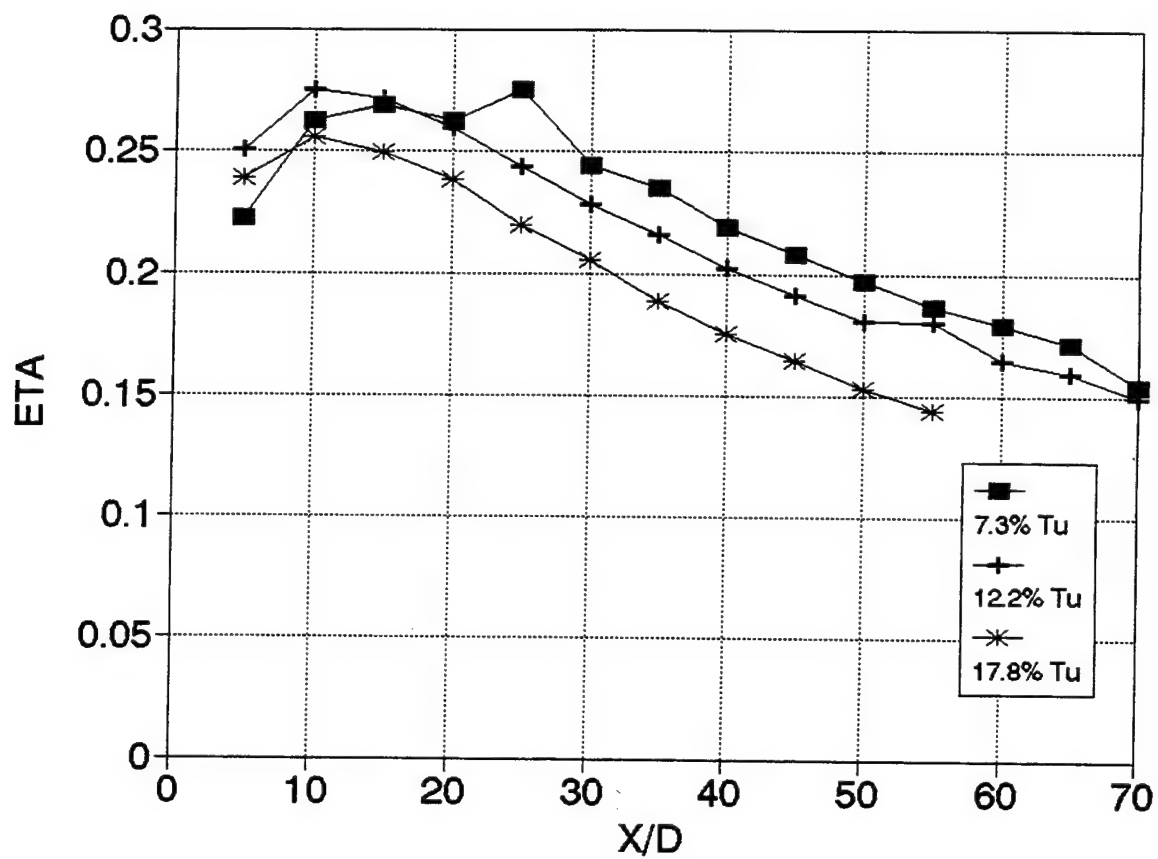


Figure E-7. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 1.75$ , Free stream velocity at injection = 18 m/s



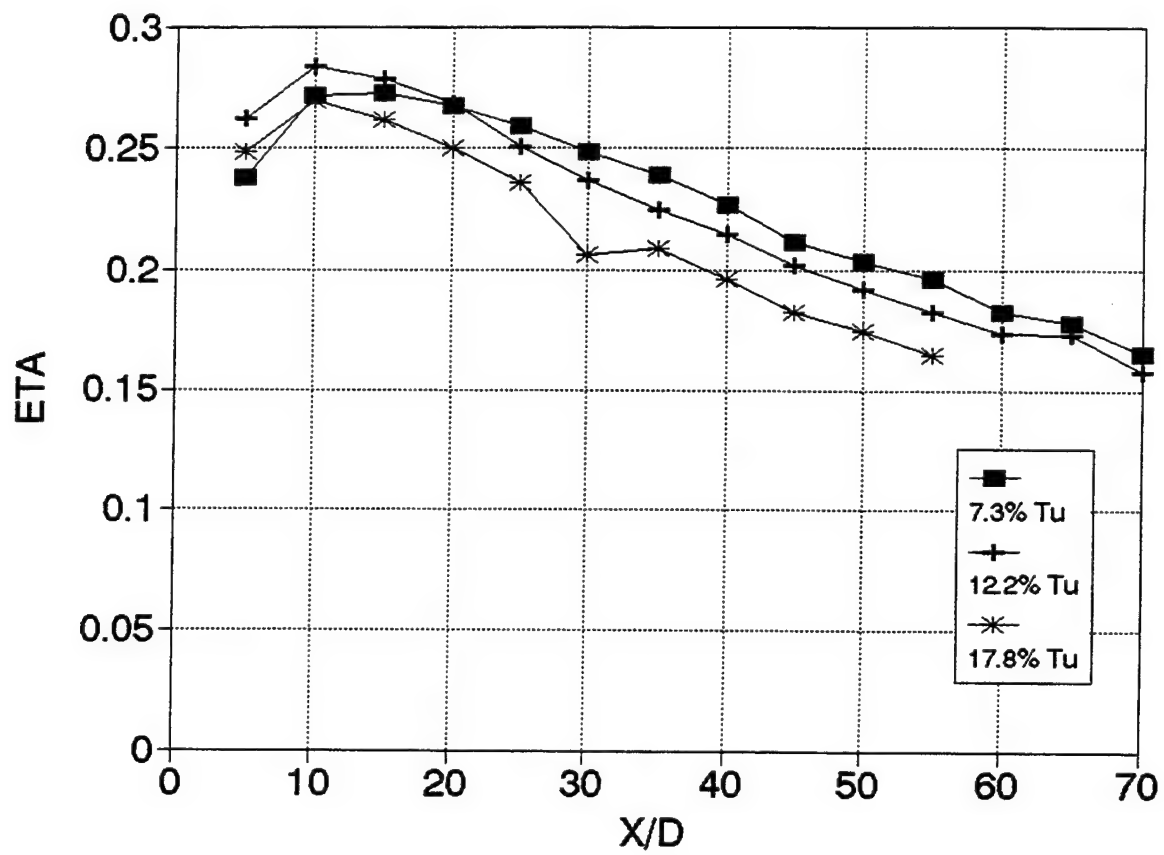


Figure E-8. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 2.0$ , Free stream velocity at injection = 18 m/s

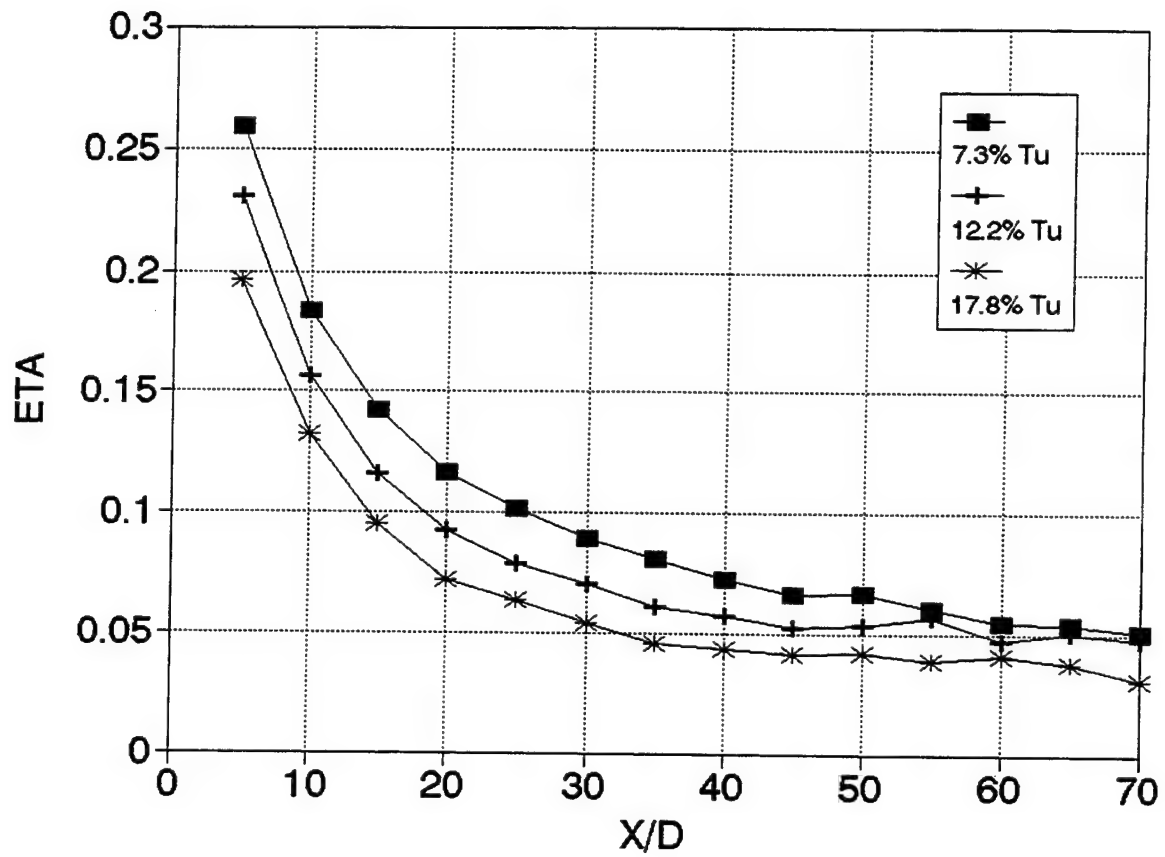


Figure E-9. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.25$ , Free stream velocity at injection = 60 m/s

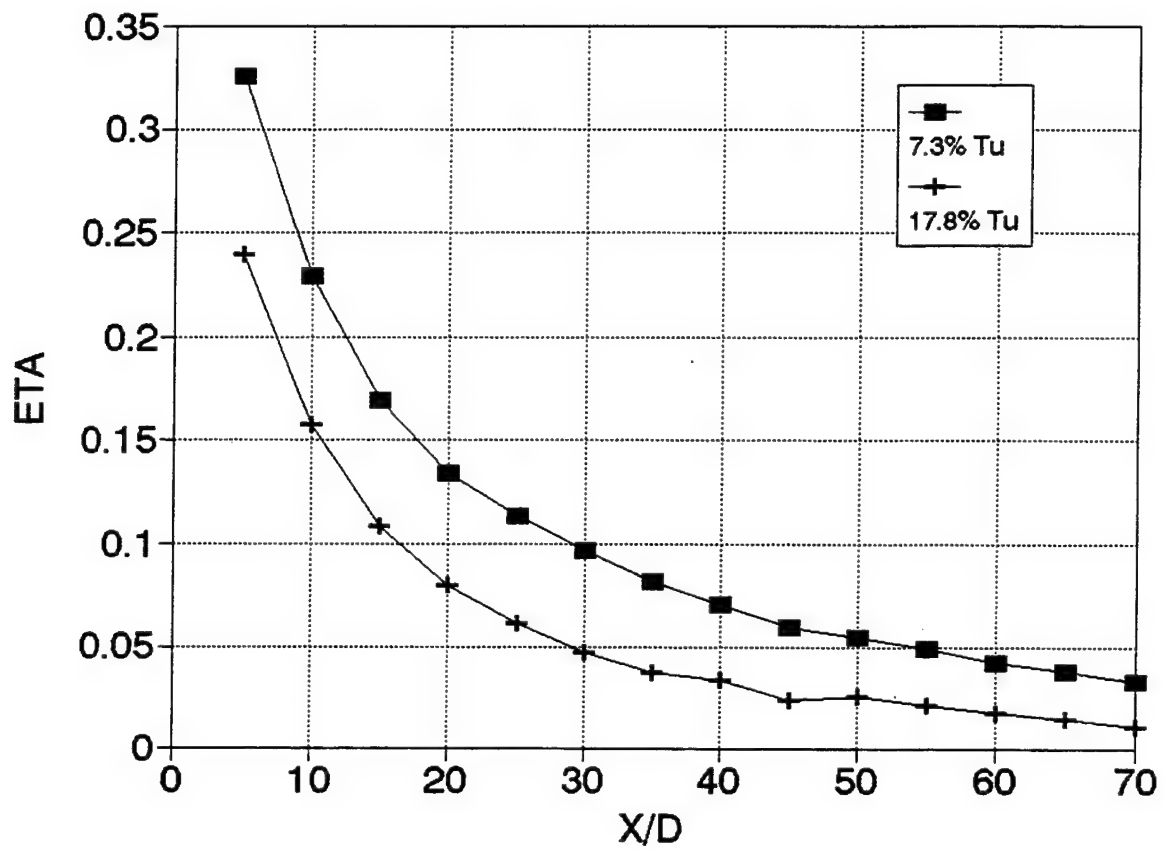


Figure E-10. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.35$ , Free stream velocity at injection = 60 m/s

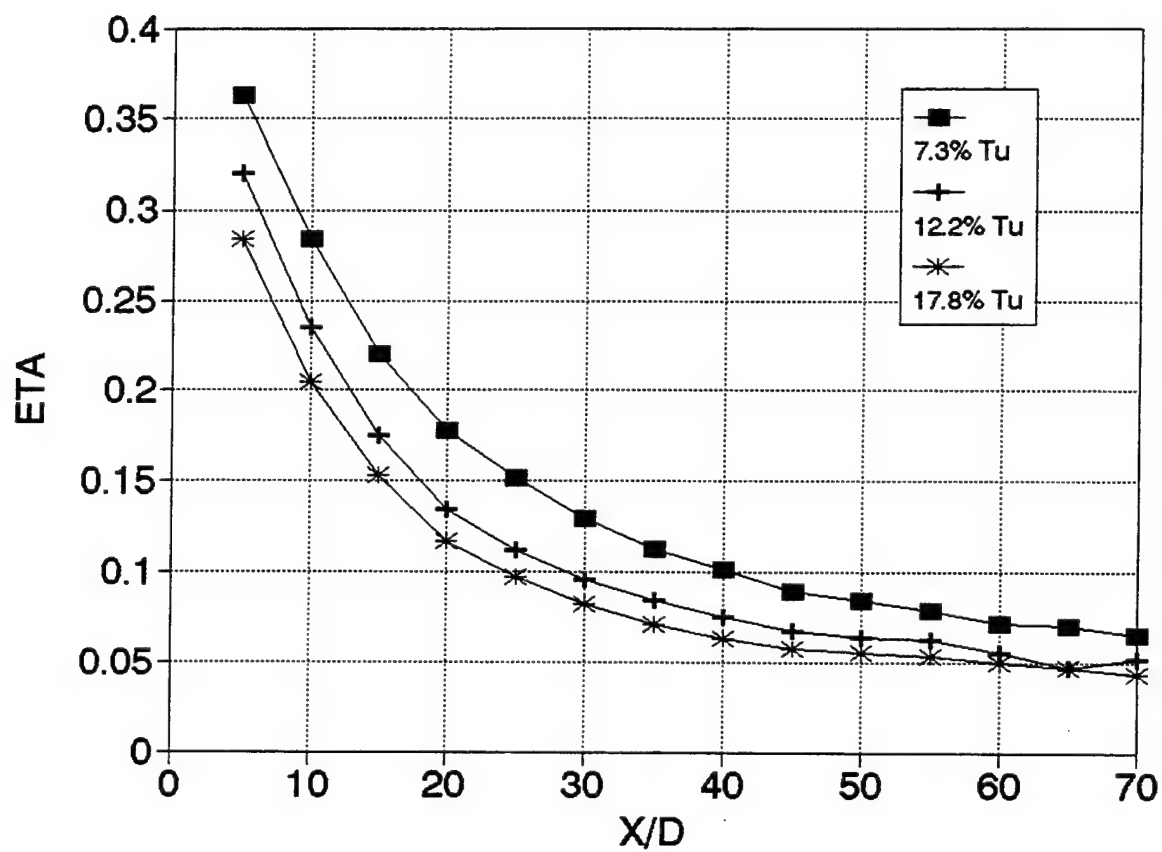


Figure E-11. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.50$ , Free stream velocity at injection = 60 m/s

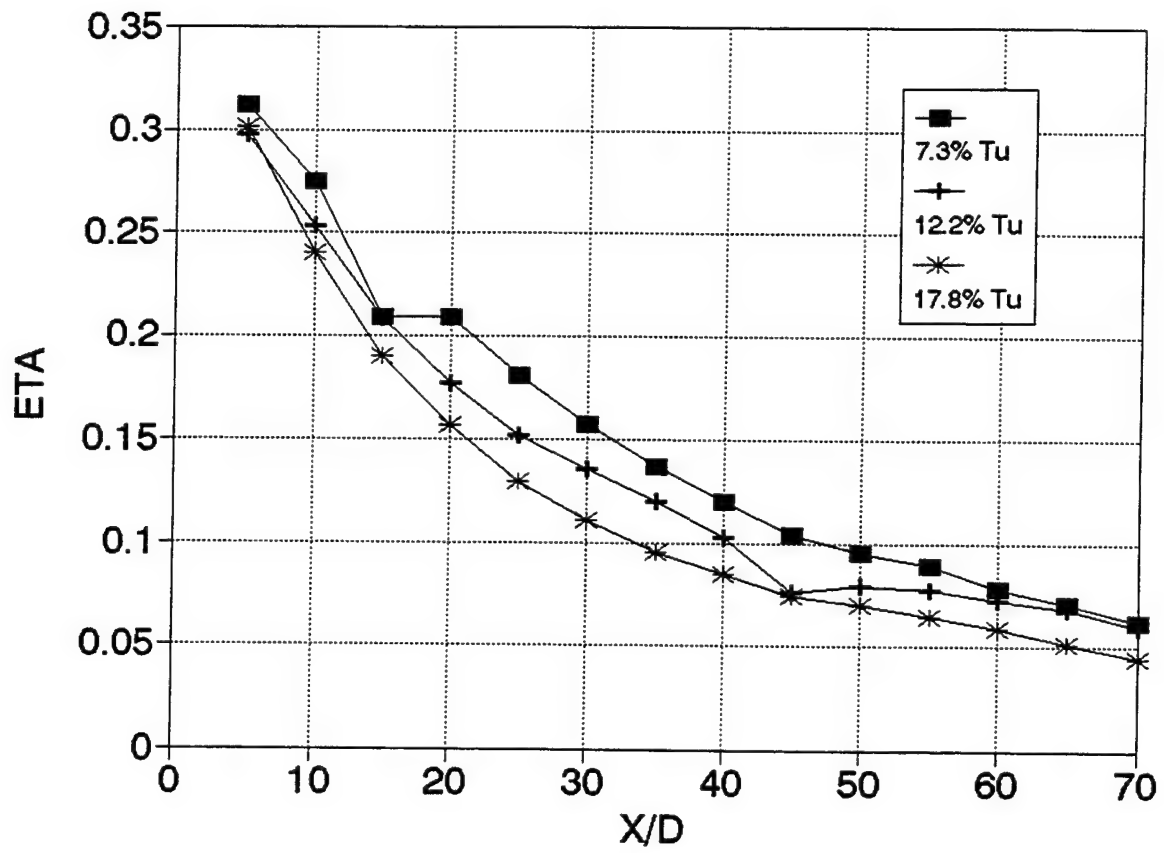


Figure E-12. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.68$ , Free stream velocity at injection = 60 m/s

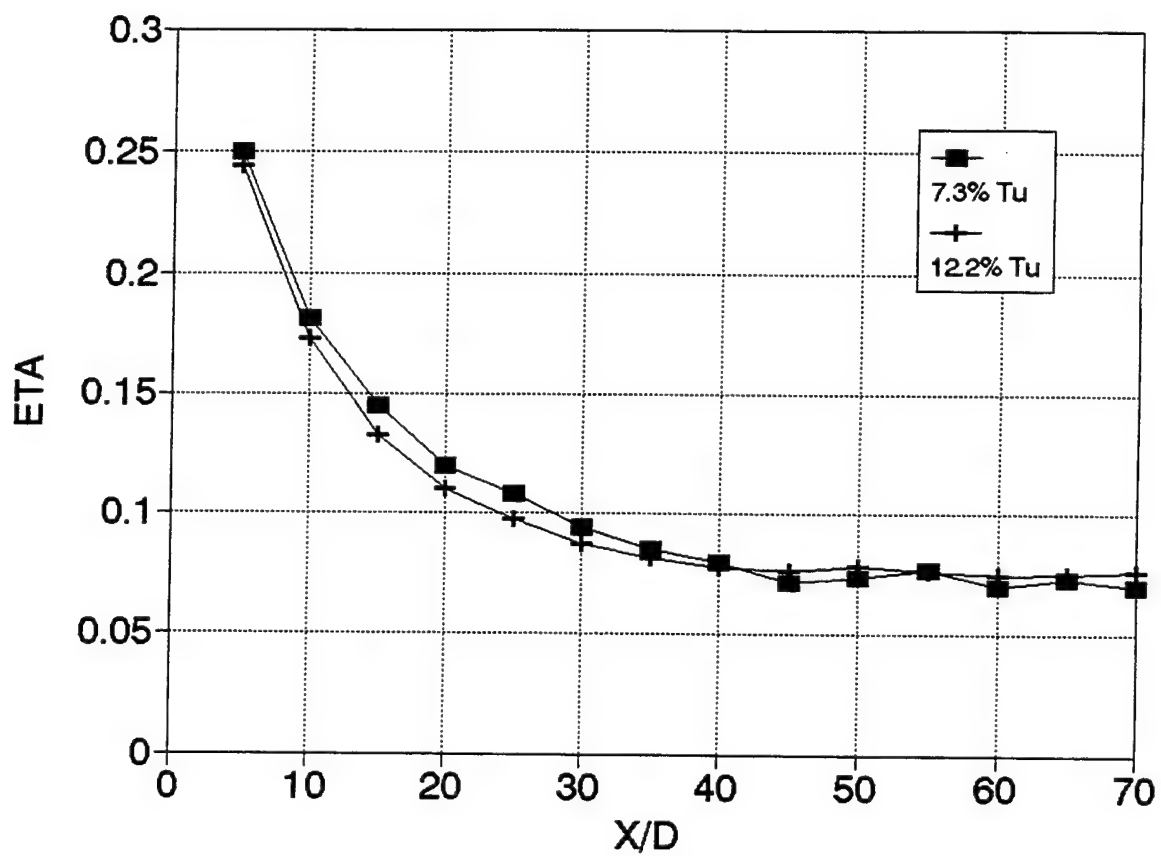


Figure E-13. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.25$ , Free stream velocity at injection = 85 m/s

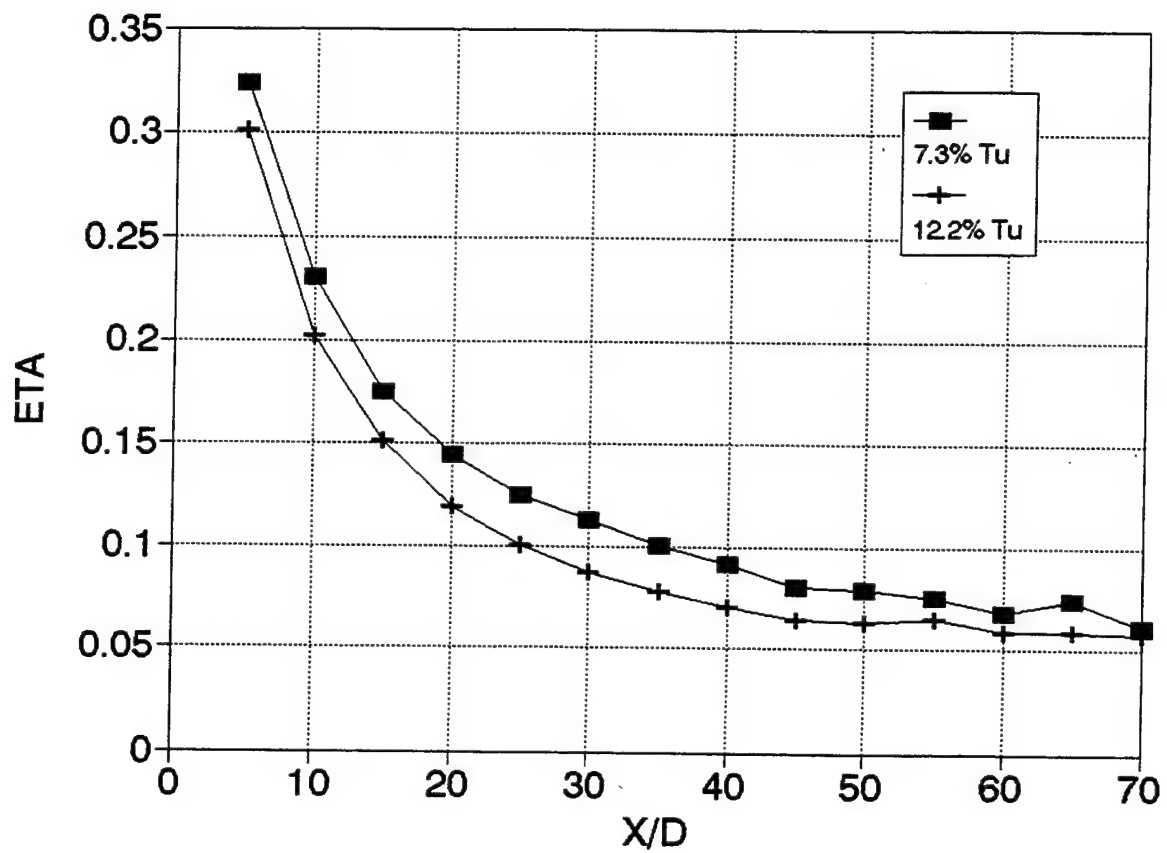


Figure E-14. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $B = 0.35$ , Free stream velocity at injection = 85 m/s

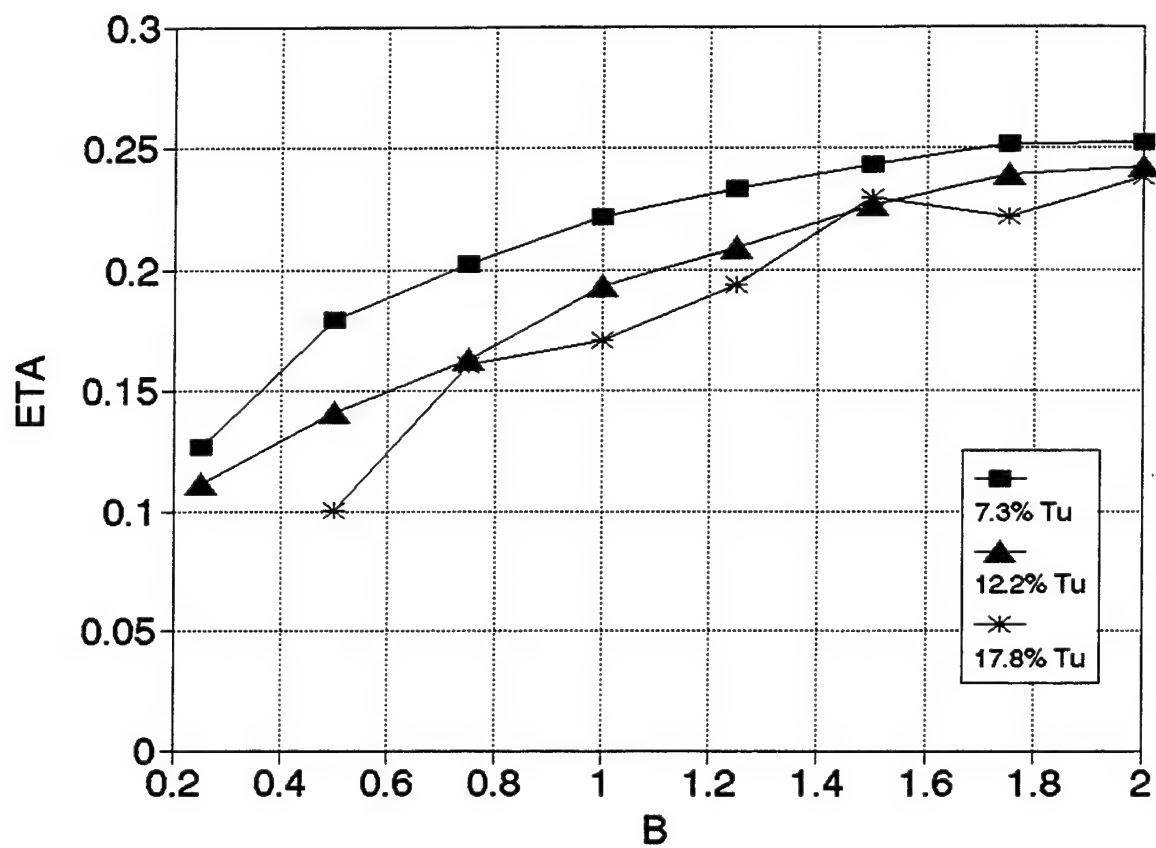


Figure E-15. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 25$ , Free stream velocity at injection = 10 m/s



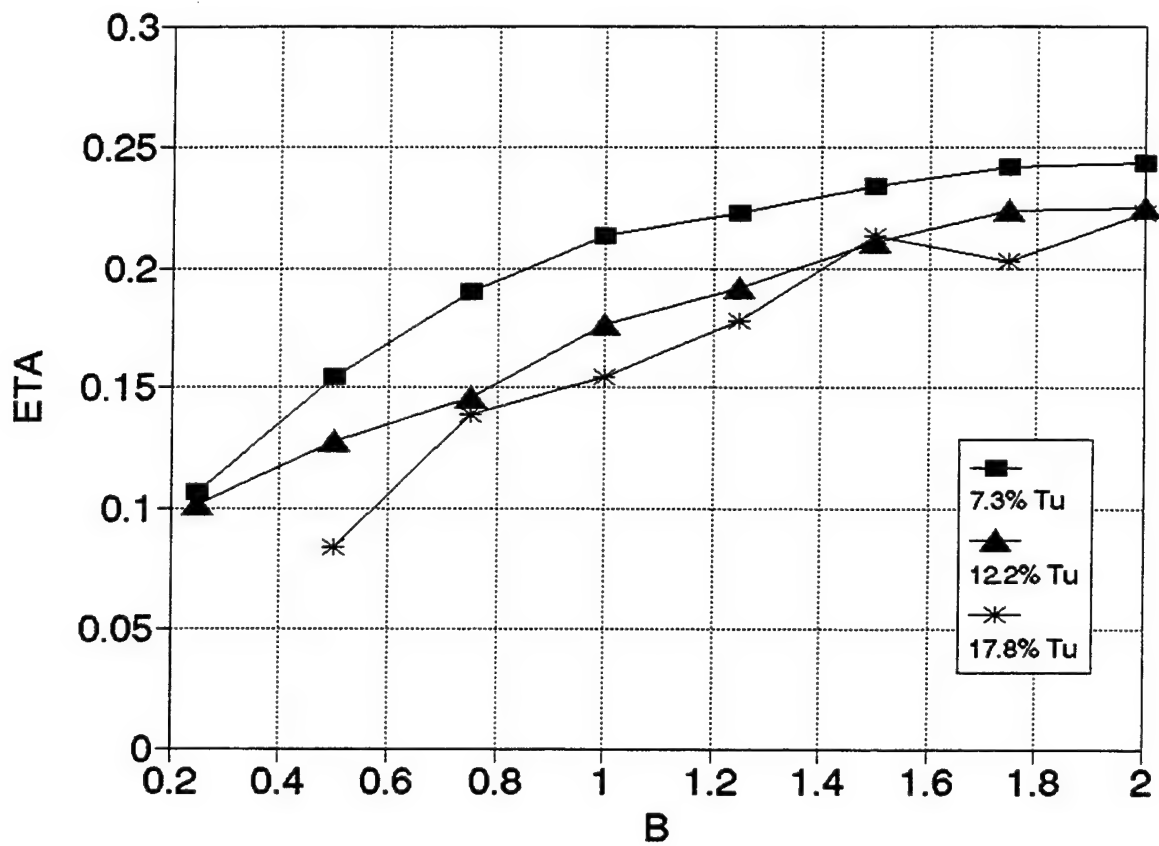


Figure E-16. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 30$ , Free stream velocity at injection = 10 m/s

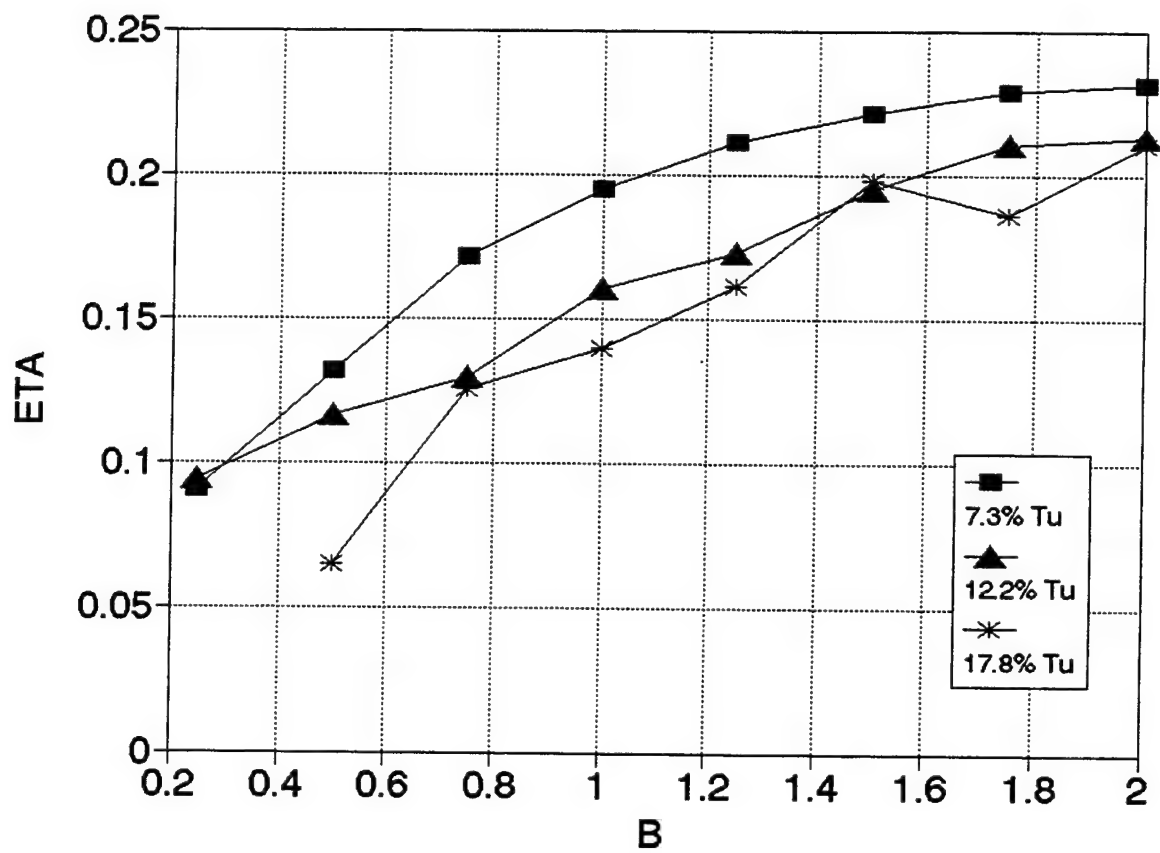


Figure E-17. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 35$ , Free stream velocity at injection = 10 m/s

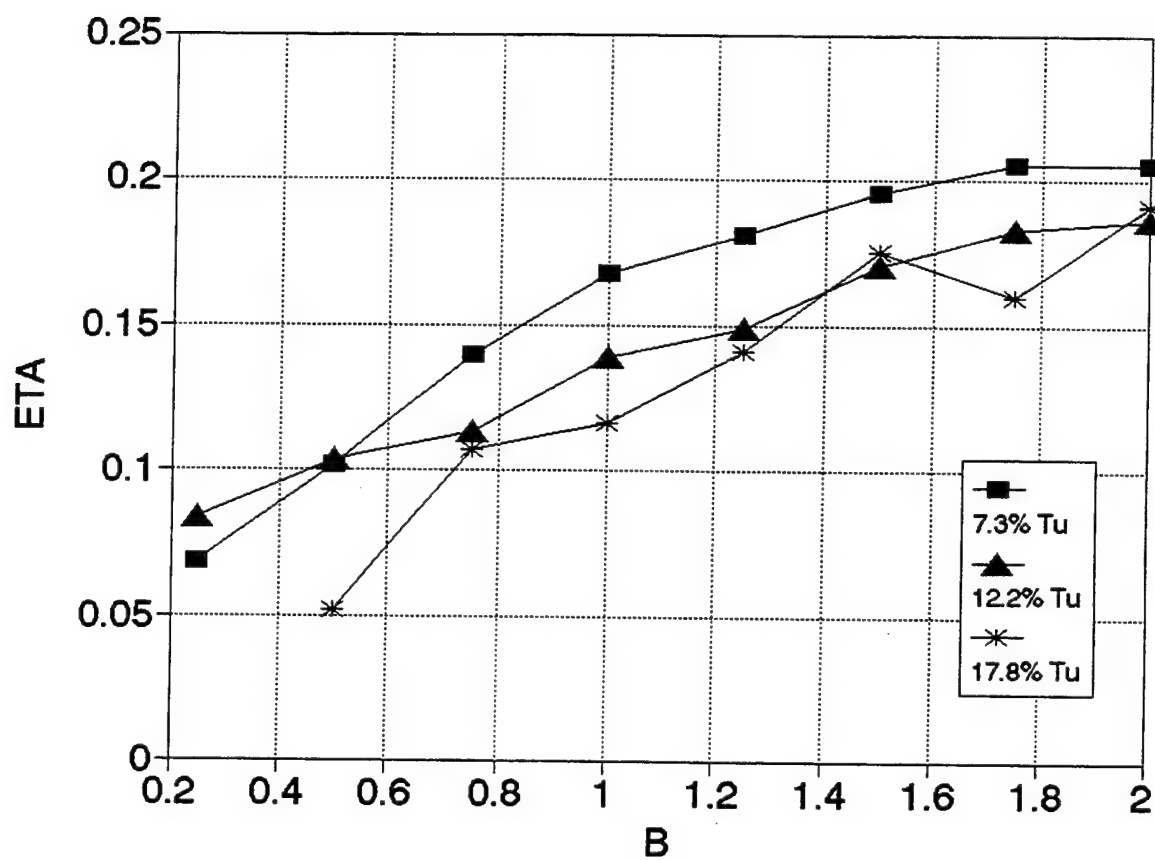


Figure E-18. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 45$ , Free stream velocity at injection = 10 m/s

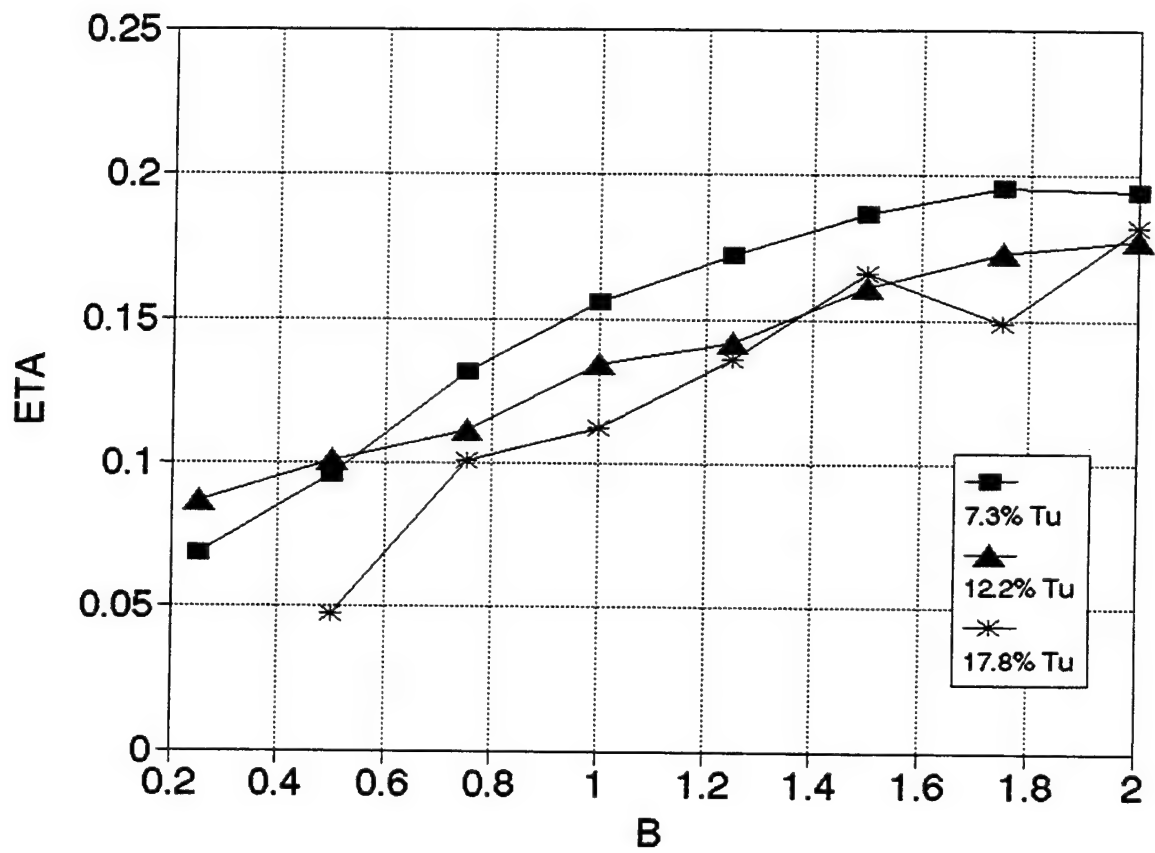


Figure E-19. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 50$ , Free stream velocity at injection = 10 m/s

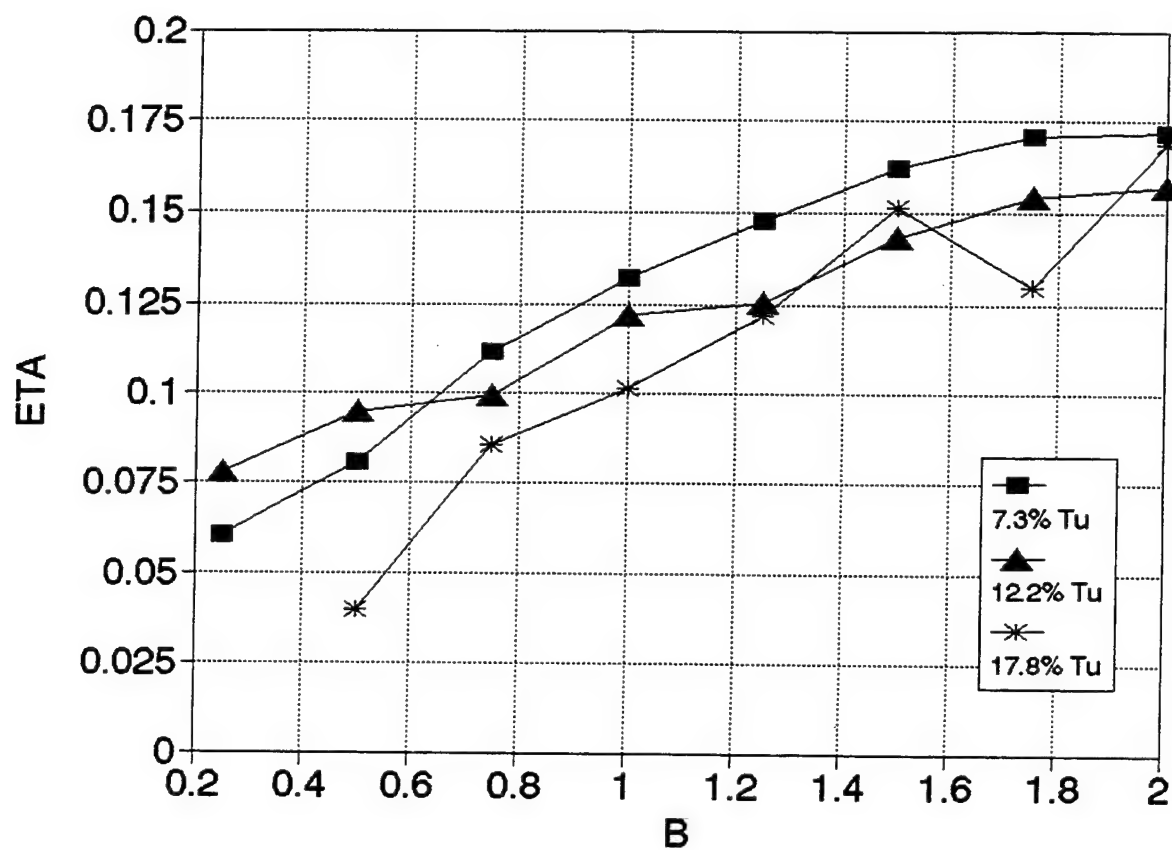


Figure E-20. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 55$ , Free stream velocity at injection = 10 m/s

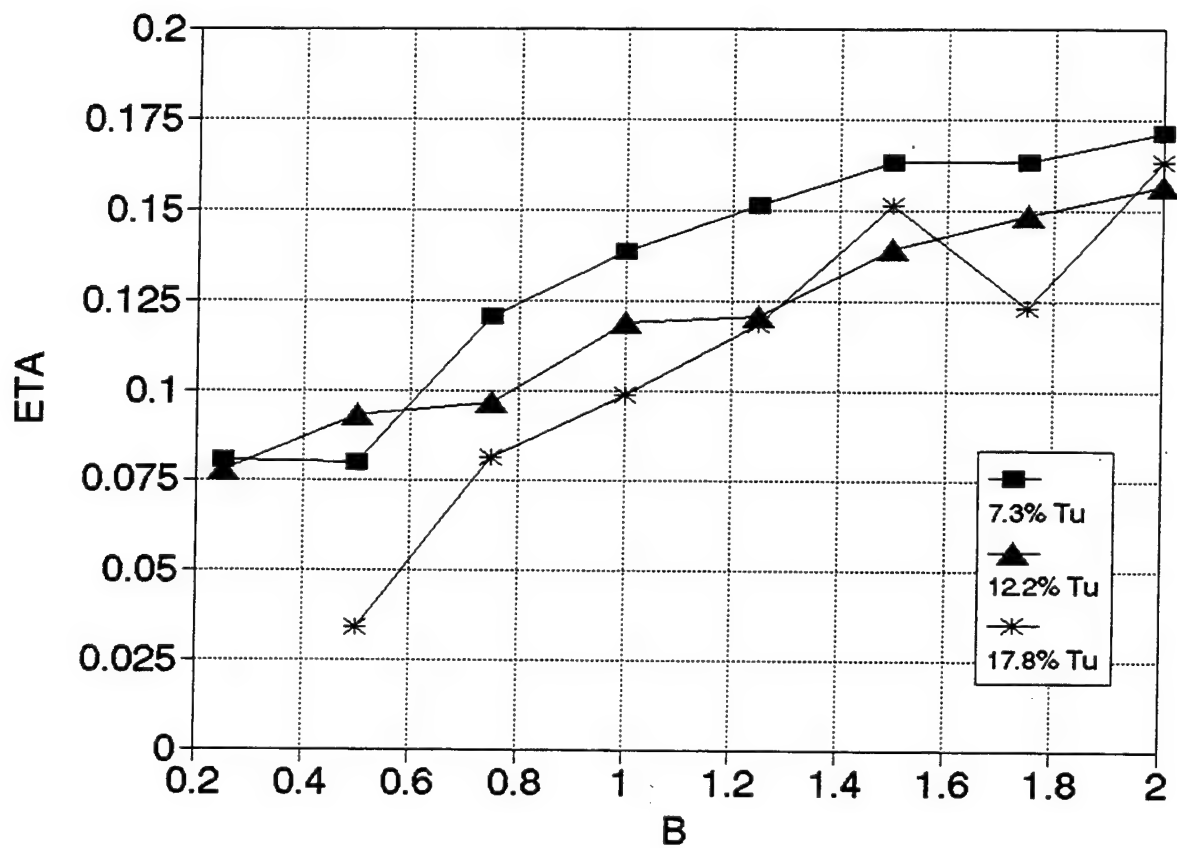


Figure E-21. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 60$ , Free stream velocity at injection = 10 m/s

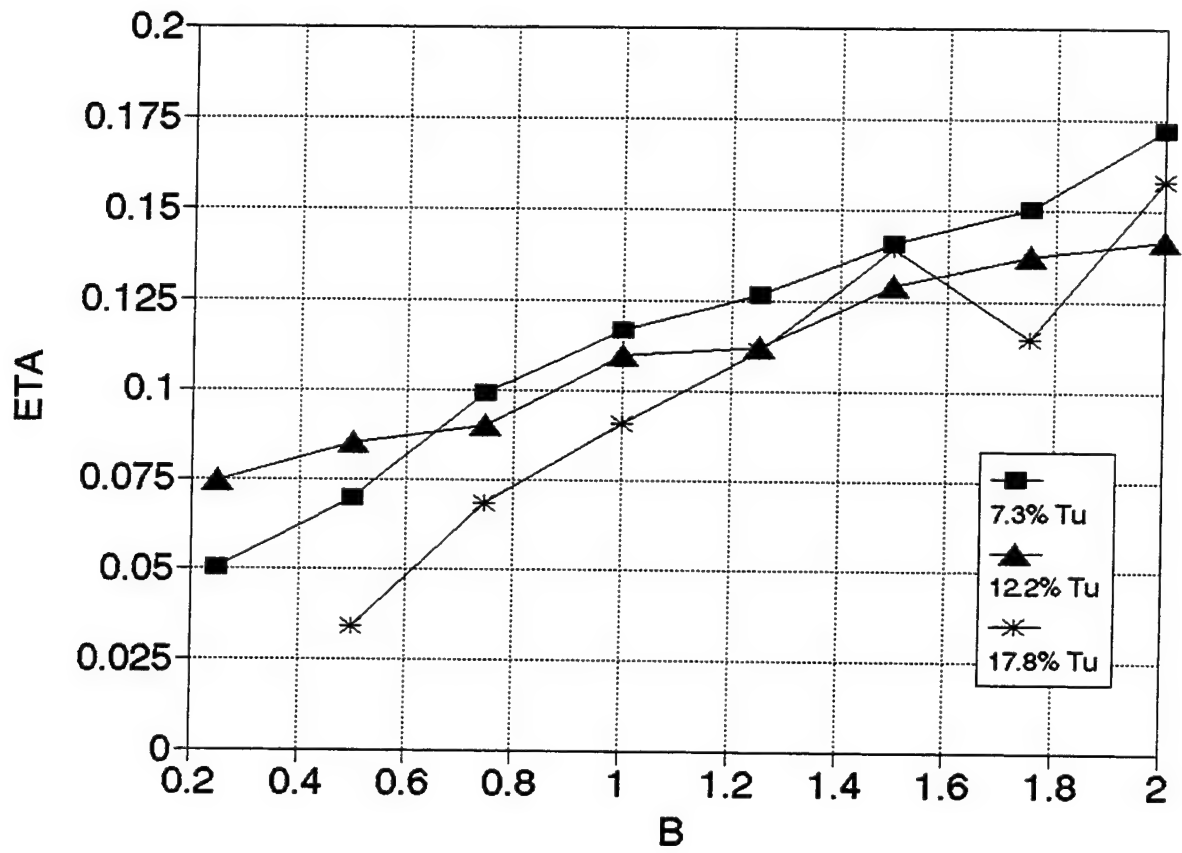


Figure E-22. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 65$ , Free stream velocity at injection = 10 m/s

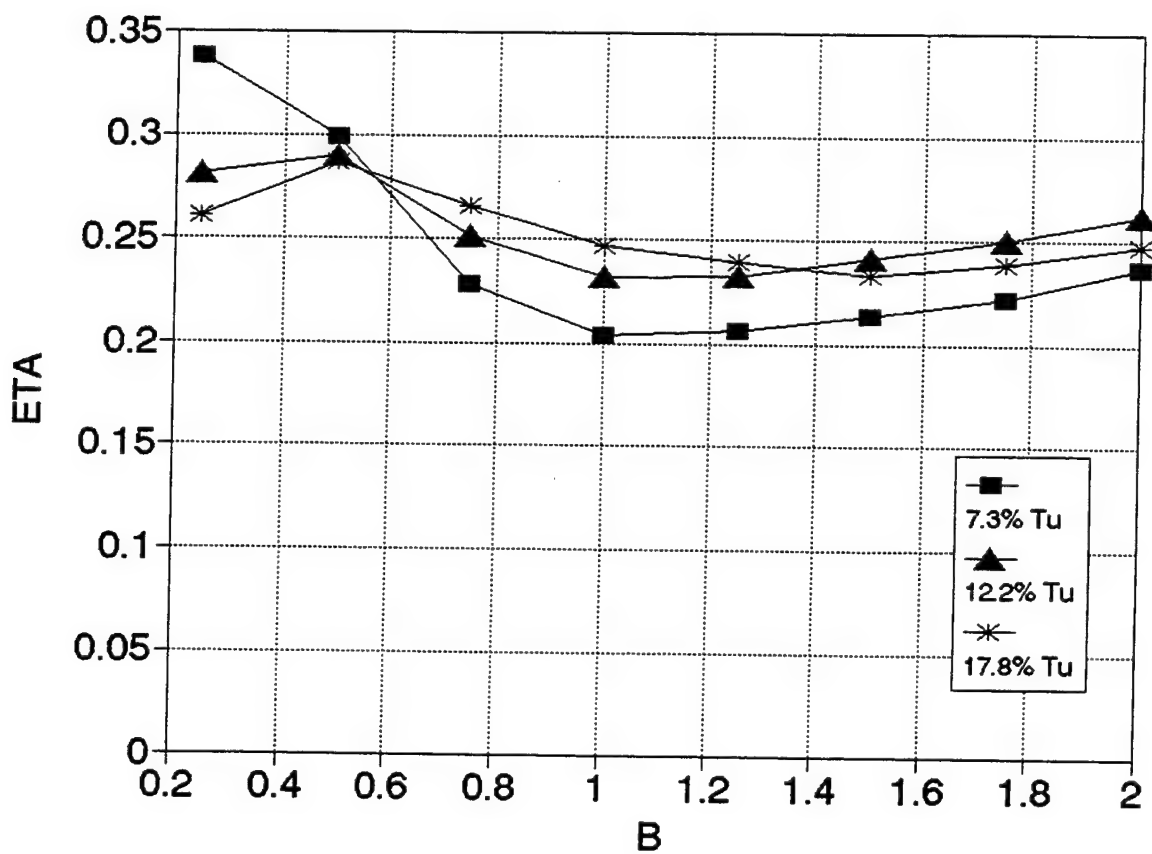


Figure E-23. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 70$ , Free stream velocity at injection = 10 m/s



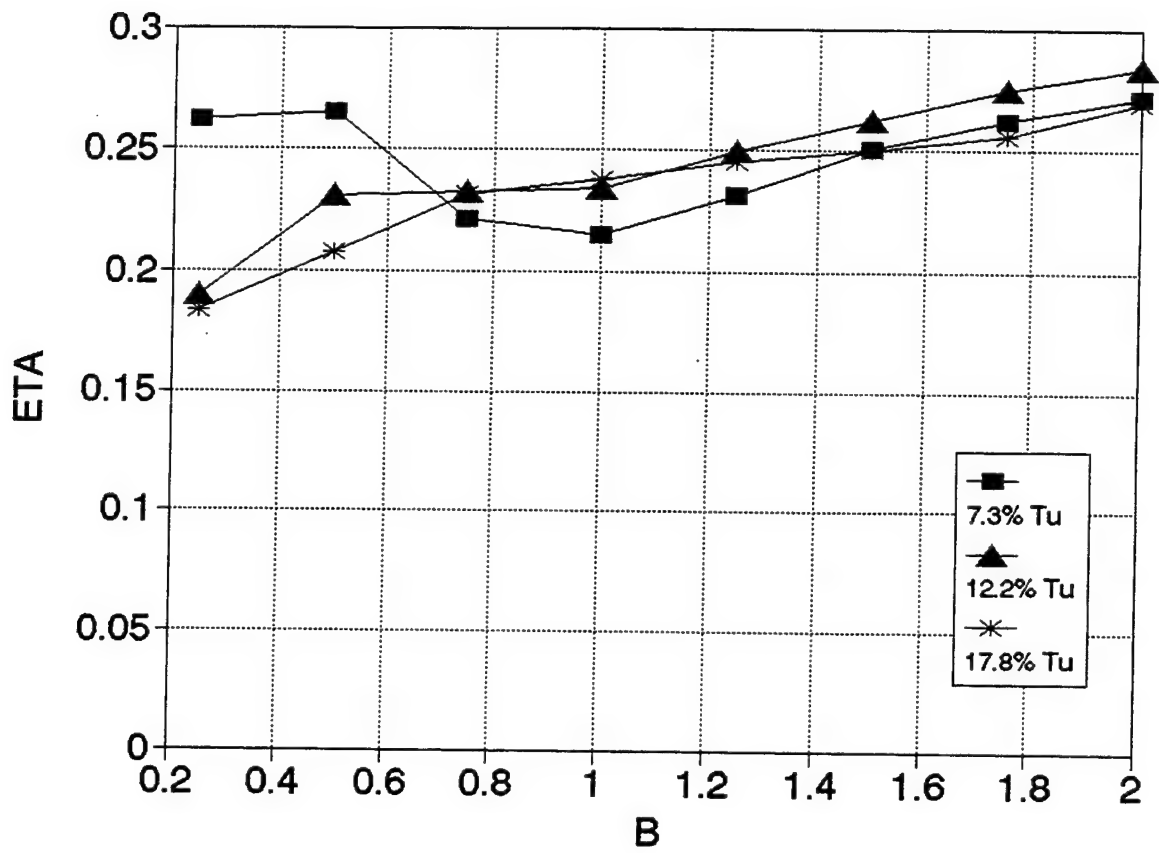


Figure E-24. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 5$ , Free stream velocity at injection = 18 m/s

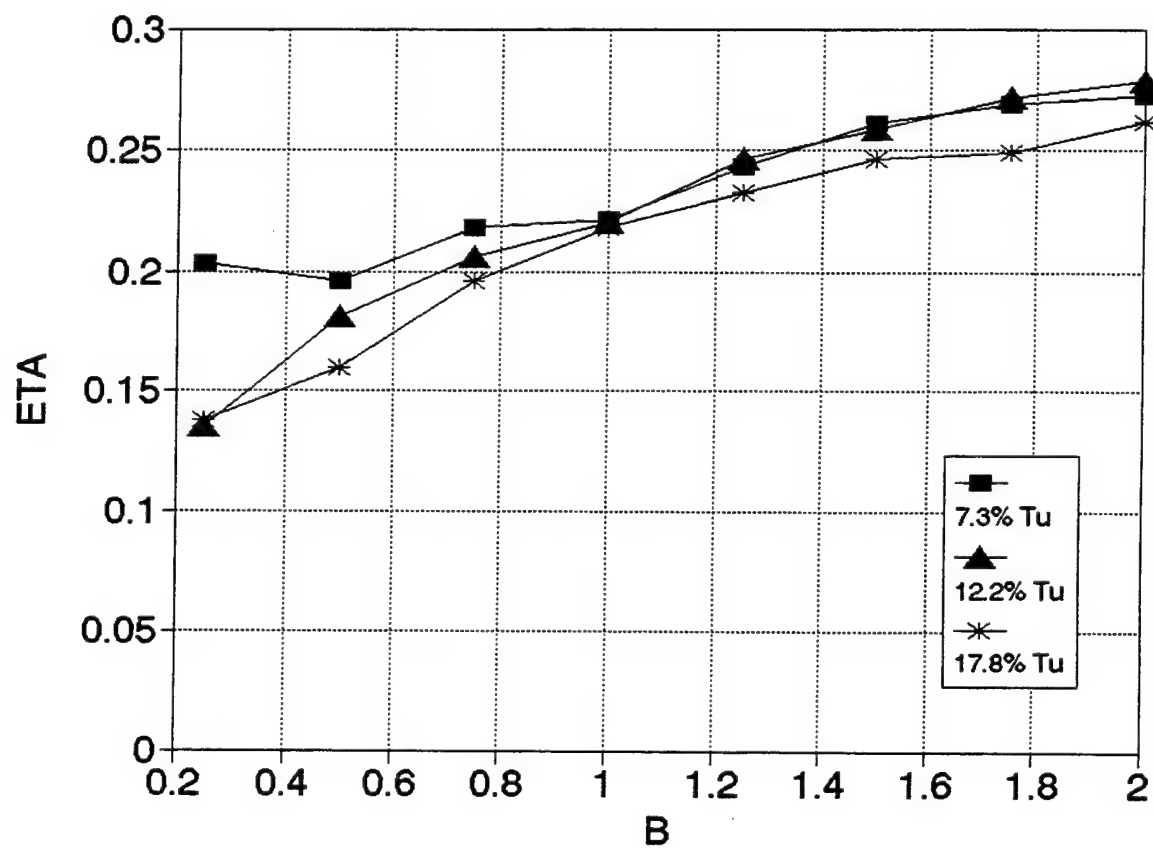


Figure E-25. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 10$ , Free stream velocity at injection = 18 m/s

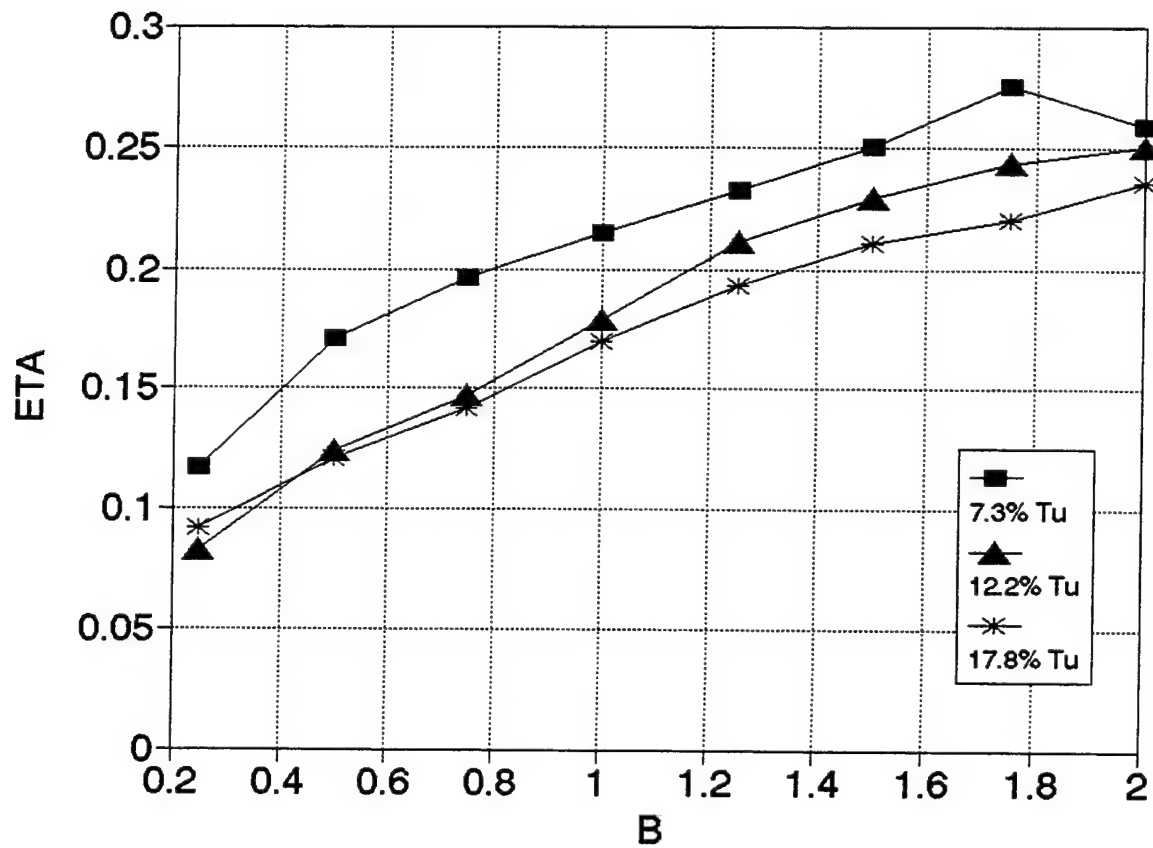


Figure E-26. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 15$ , Free stream velocity at injection = 18 m/s

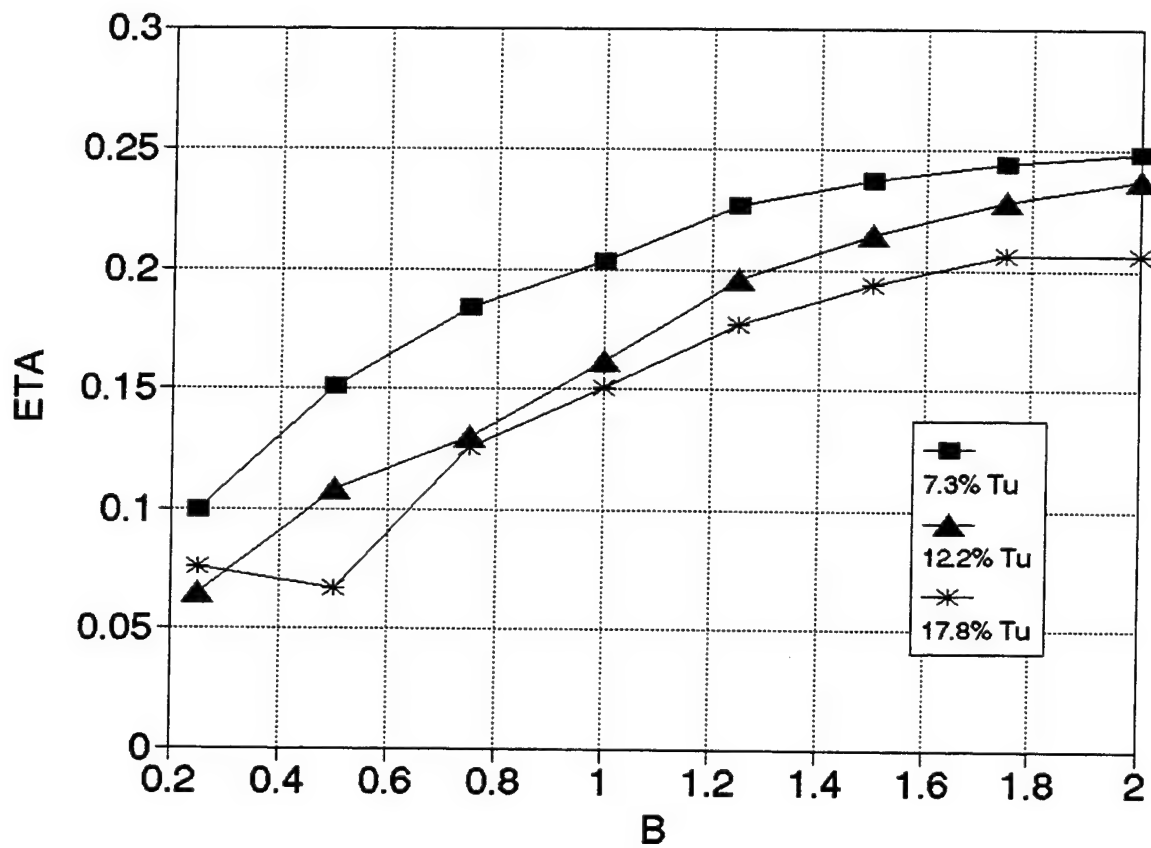


Figure E-27. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 25$ , Free stream velocity at injection = 18 m/s

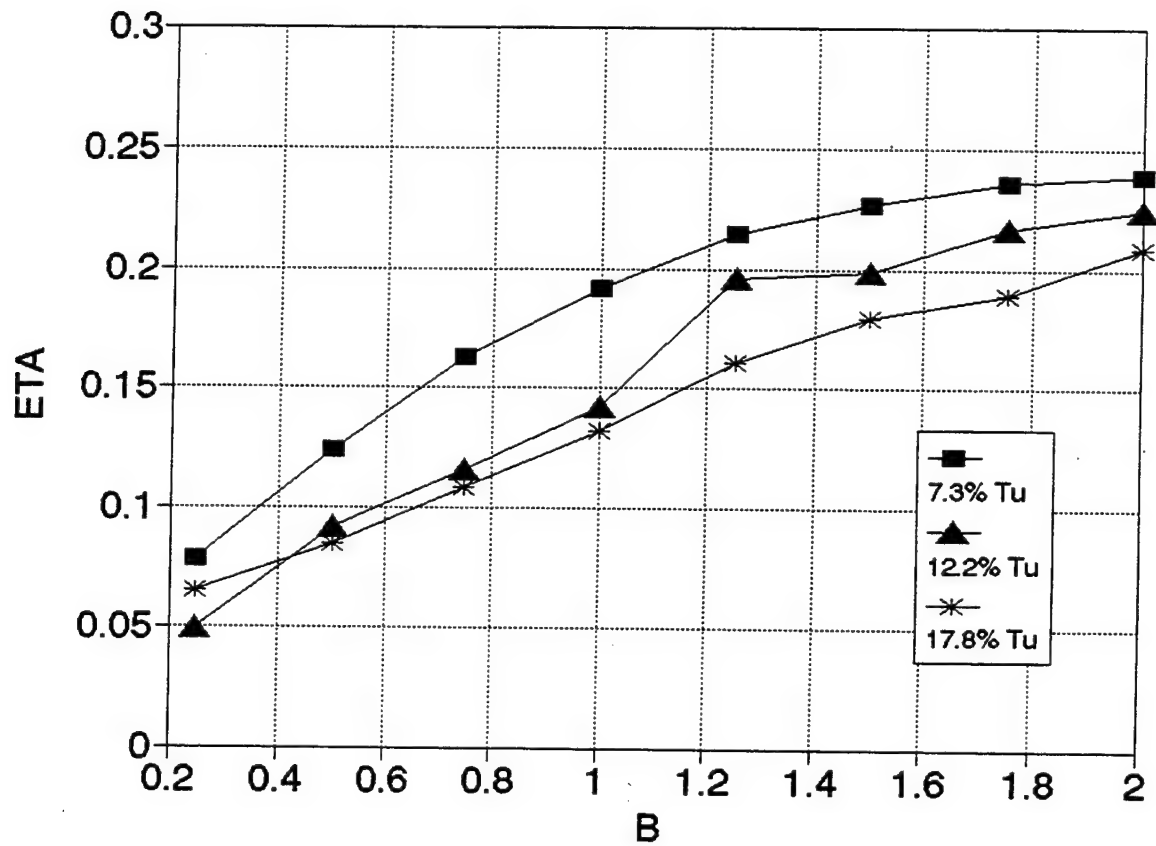


Figure E-28. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 30$ , Free stream velocity at injection = 18 m/s

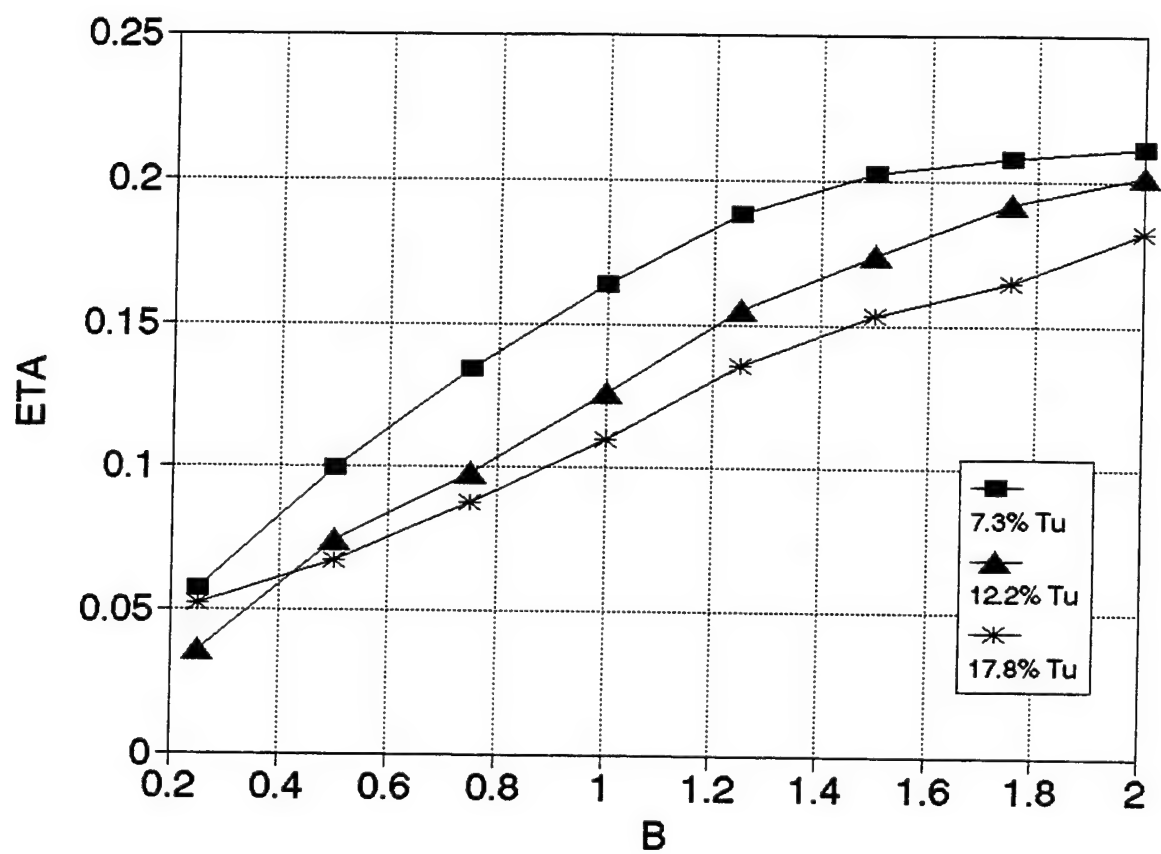


Figure E-29. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 35$ , Free stream velocity at injection = 18 m/s

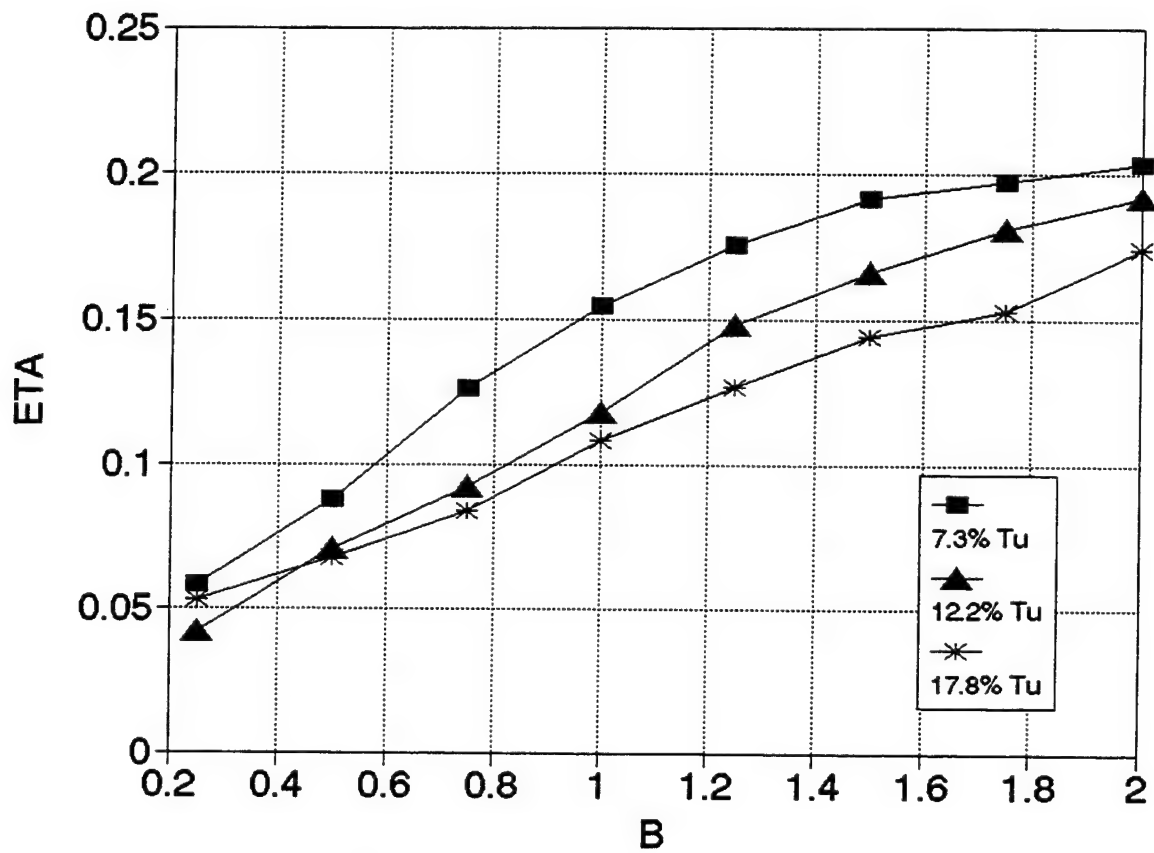


Figure E-30. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 45$ , Free stream velocity at injection = 18 m/s

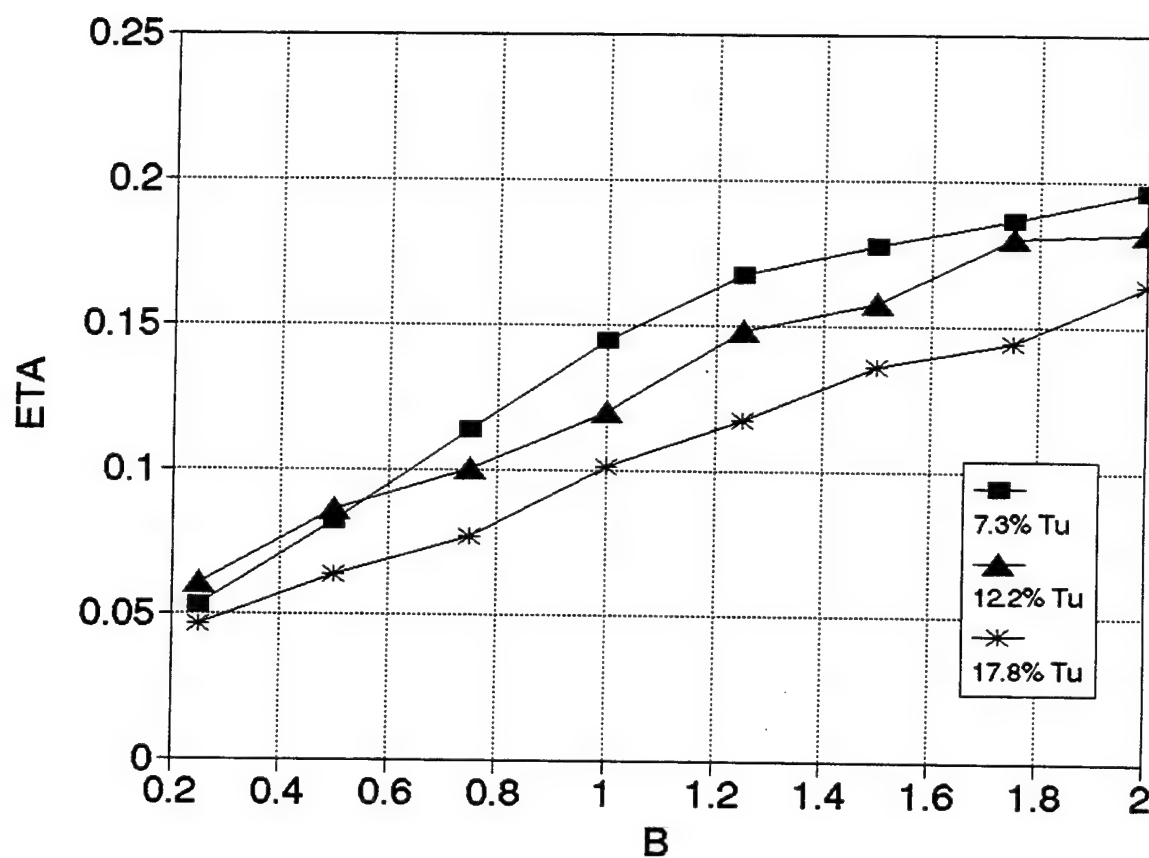


Figure E-31. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 50$ , Free stream velocity at injection = 18 m/s



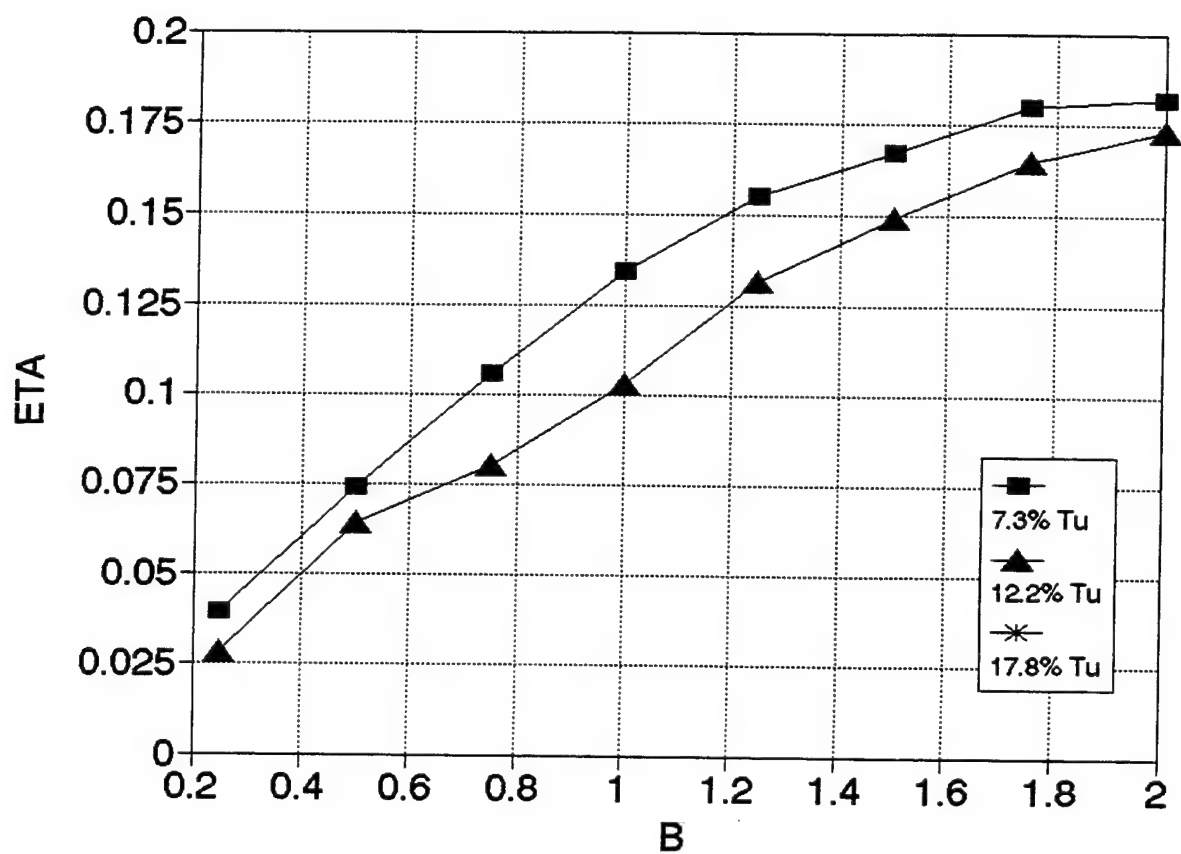


Figure E-32. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 55$ , Free stream velocity at injection = 18 m/s

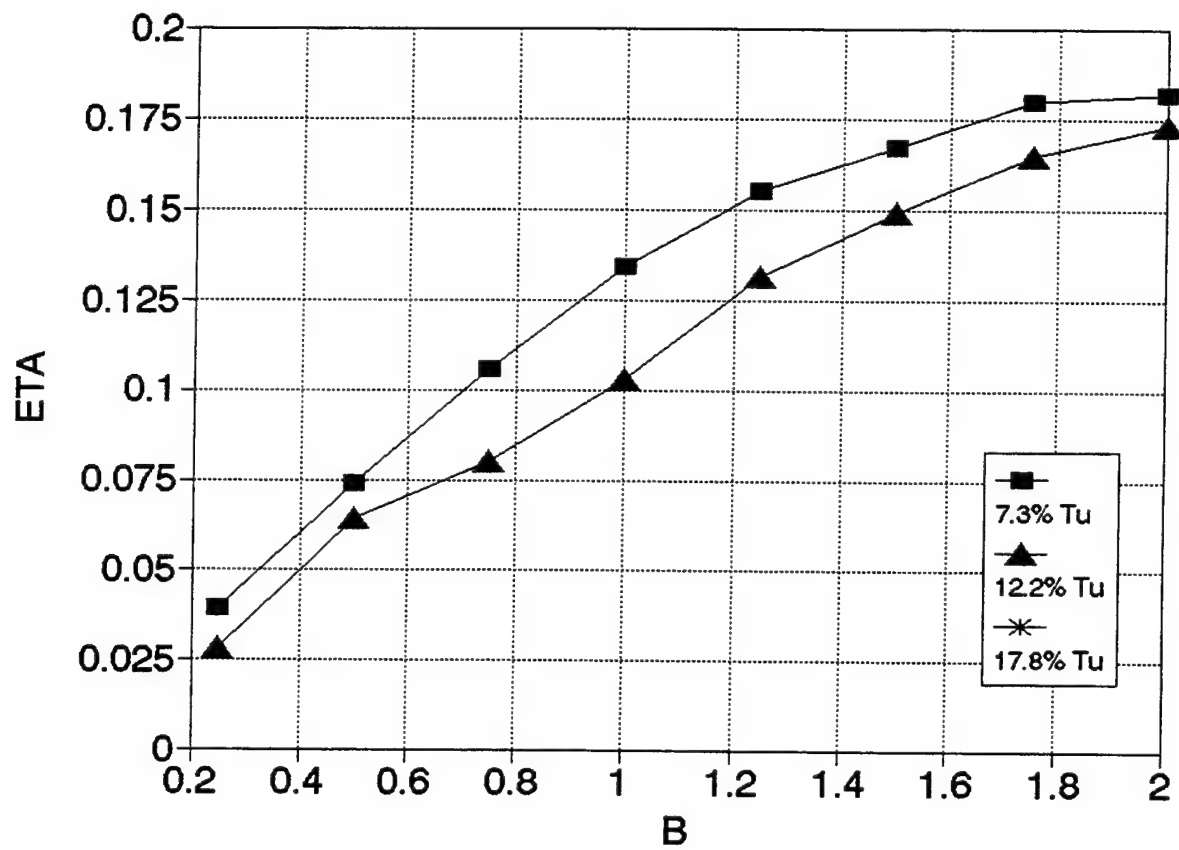


Figure E-33. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 60$ , Free stream velocity at injection = 18 m/s

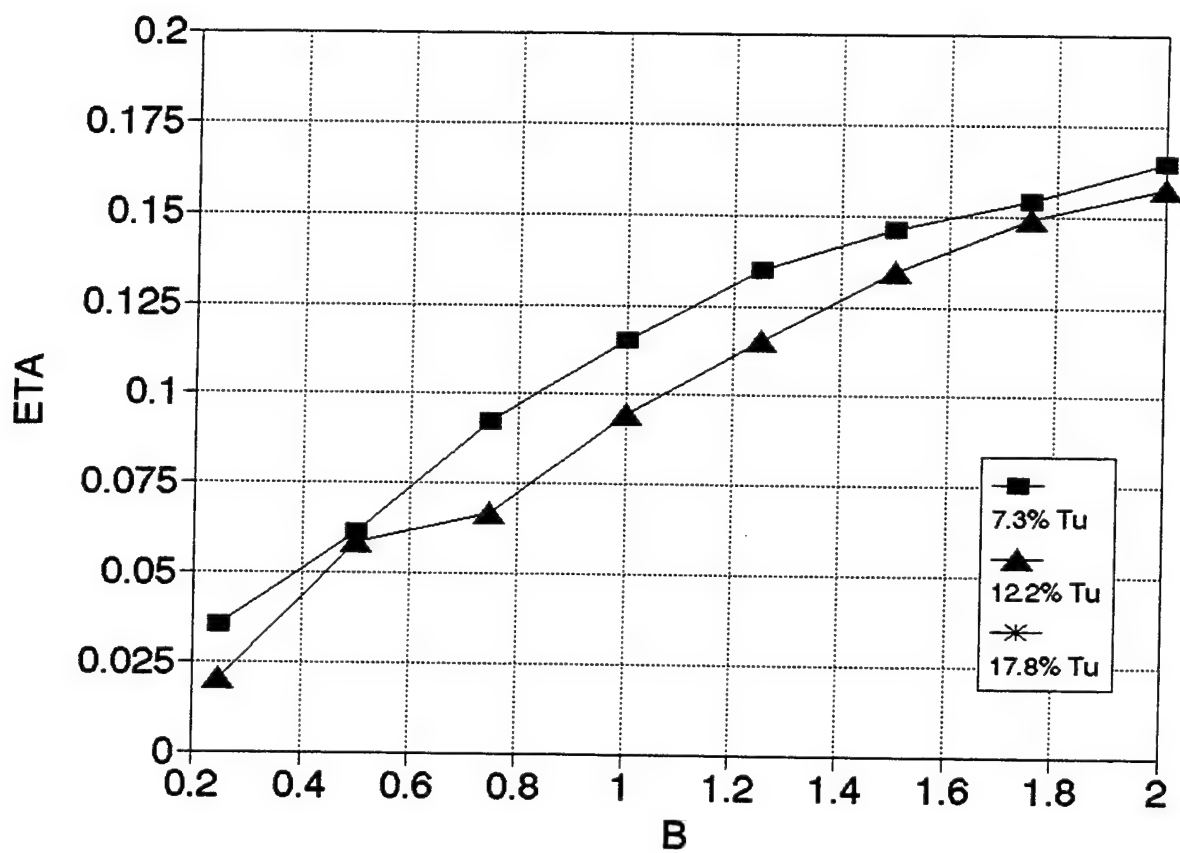


Figure E-34. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 65$ , Free stream velocity at injection = 18 m/s

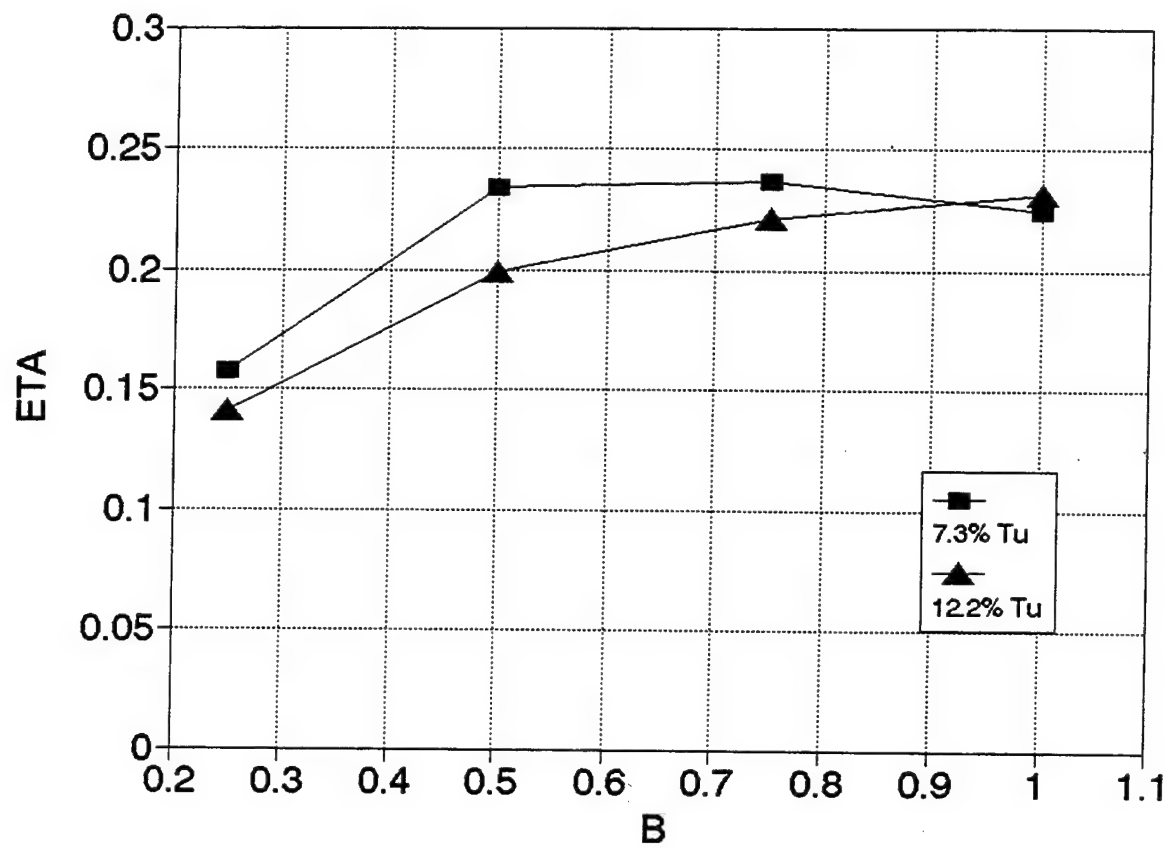


Figure E-35. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 70$ , Free stream velocity at injection = 18 m/s

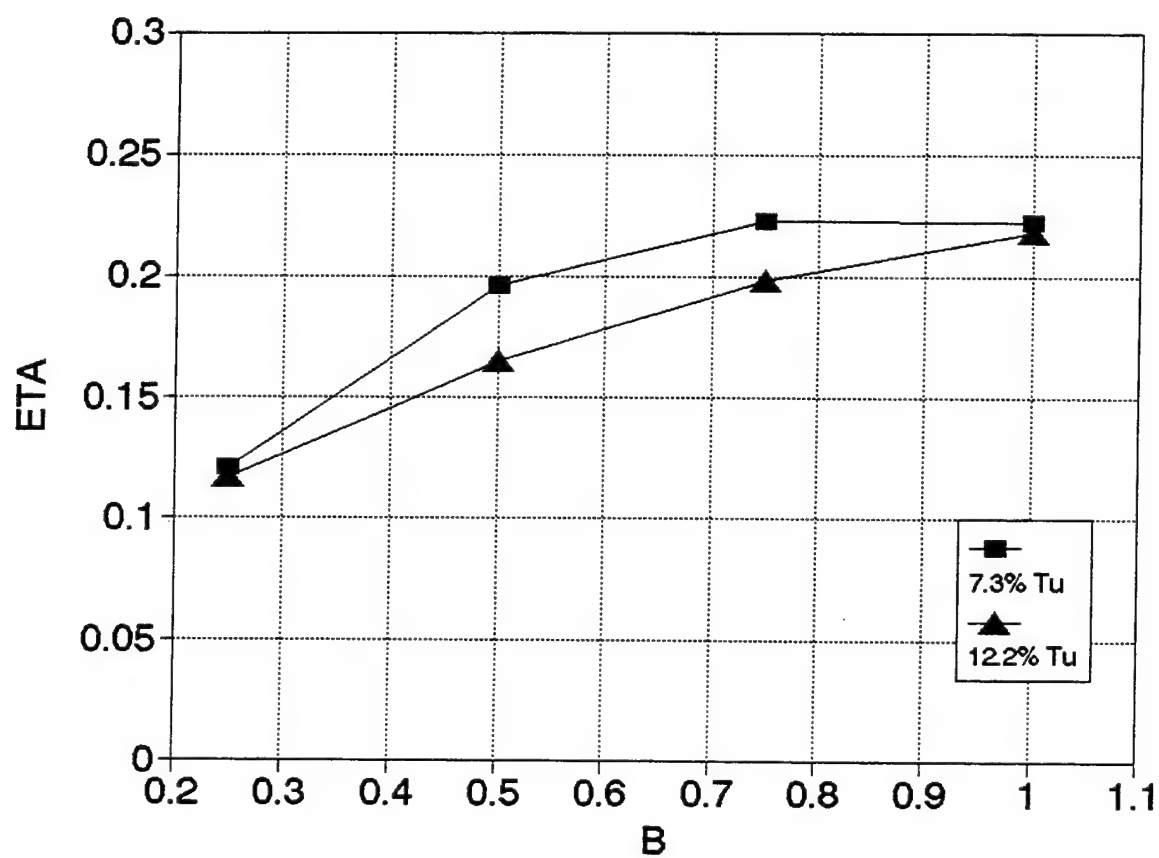


Figure E-36. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 15$ , Free stream velocity at injection = 37 m/s

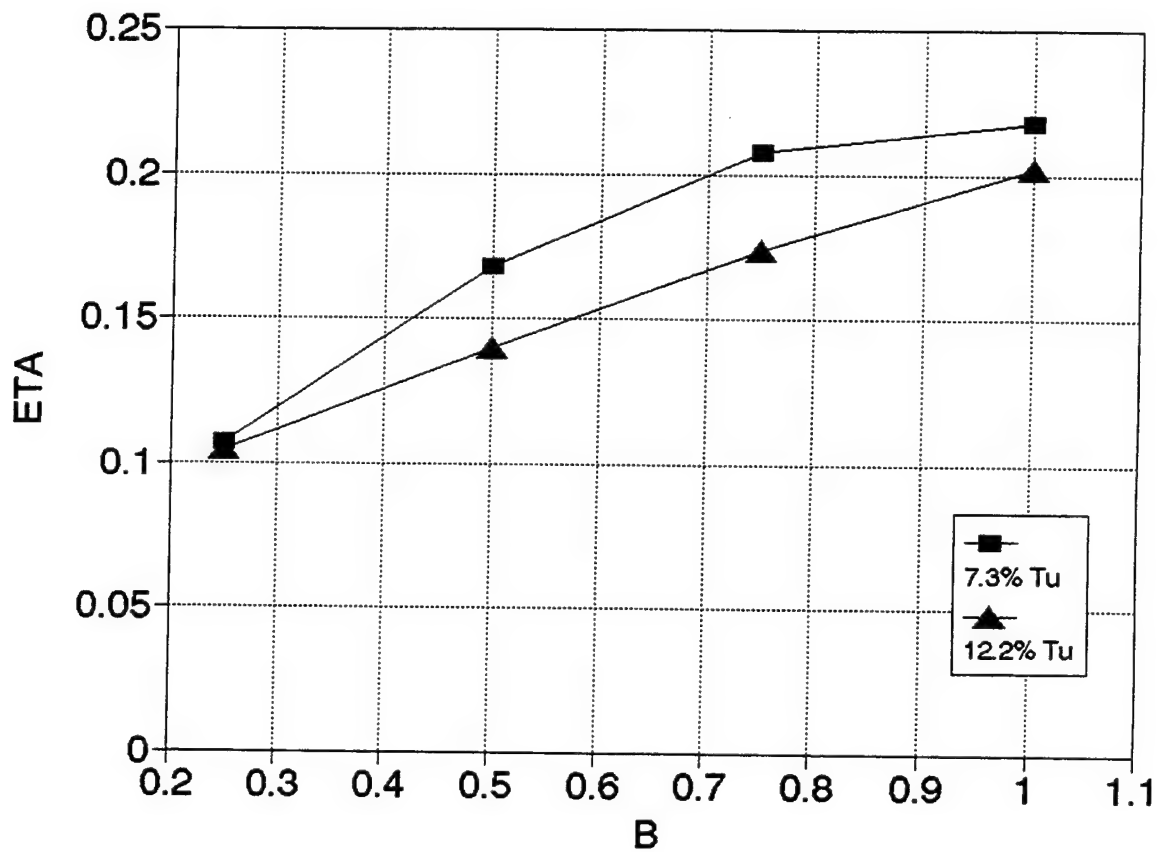


Figure E-37. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 20$ , Free stream velocity at injection = 37 m/s

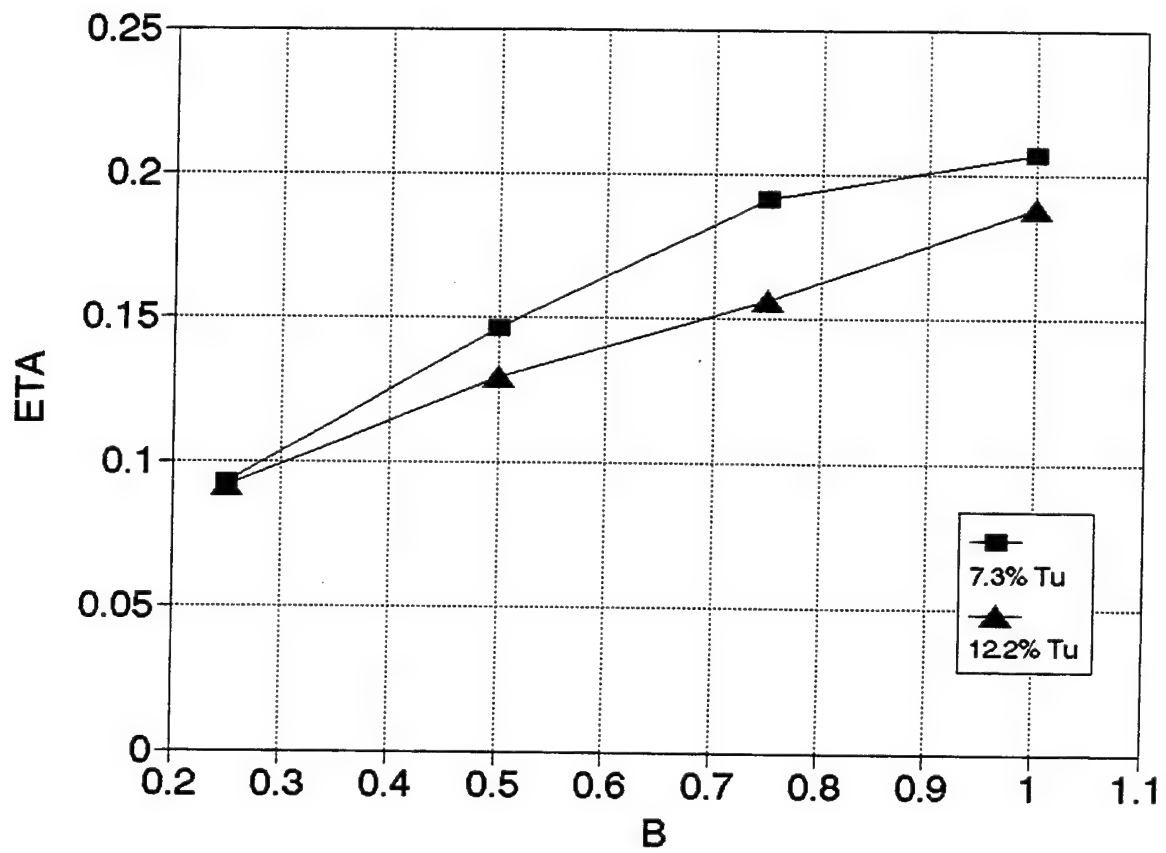


Figure E-38. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 25$ , Free stream velocity at injection = 37 m/s

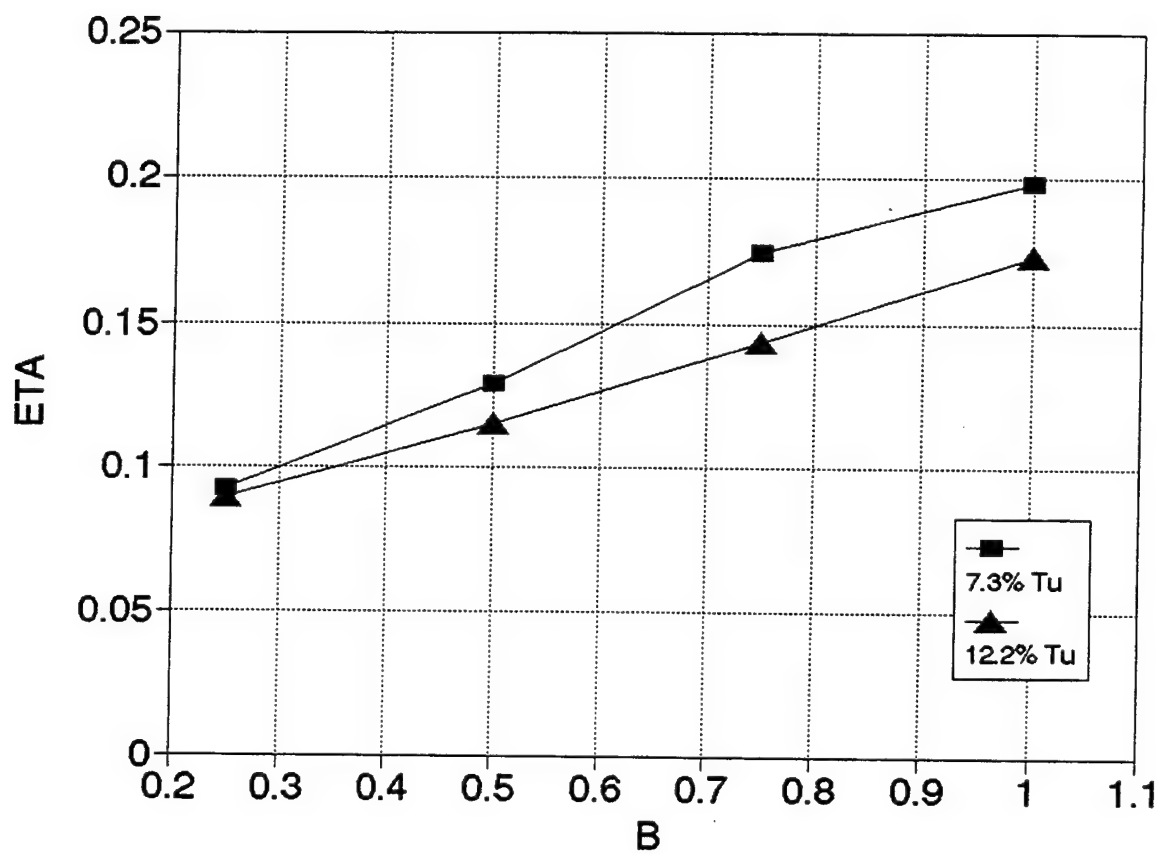


Figure E-39. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 30$ , Free stream velocity at injection = 37 m/s



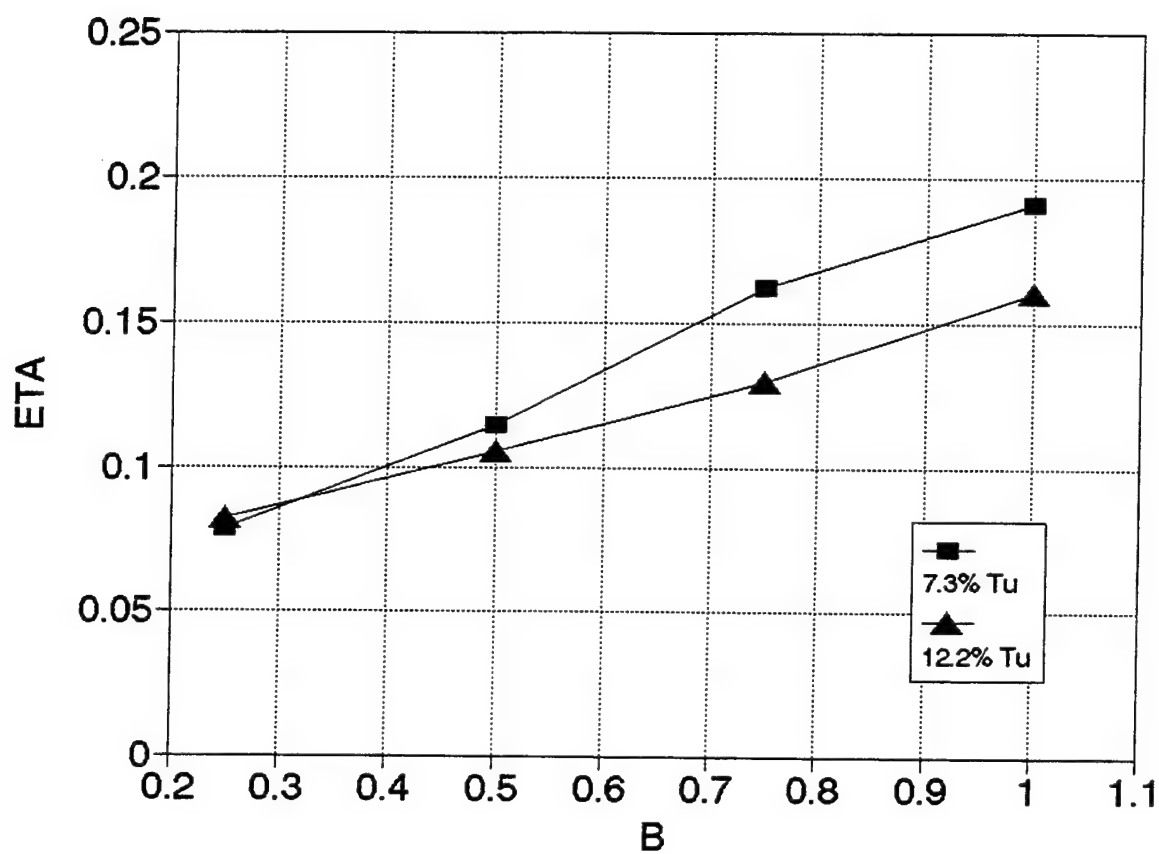


Figure E-40. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 35$ , Free stream velocity at injection = 37 m/s

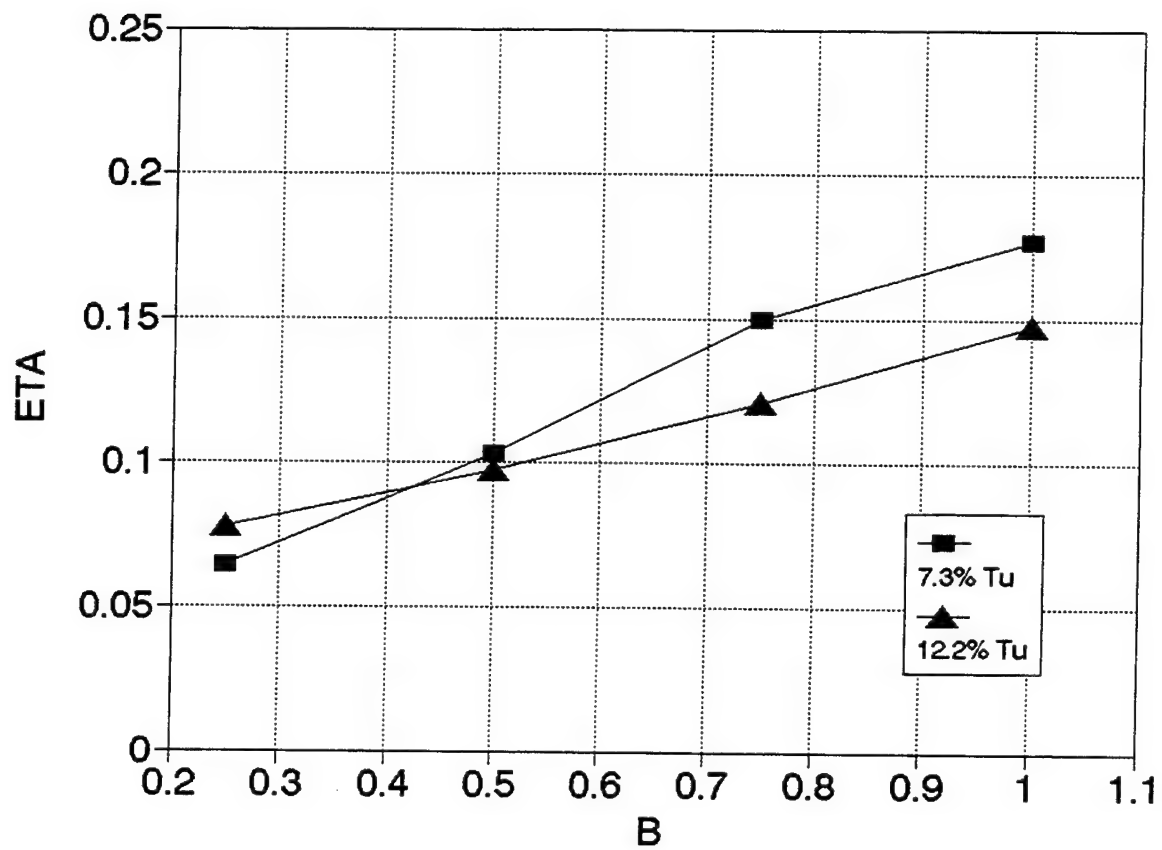


Figure E-41. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 40$ , Free stream velocity at injection = 37 m/s

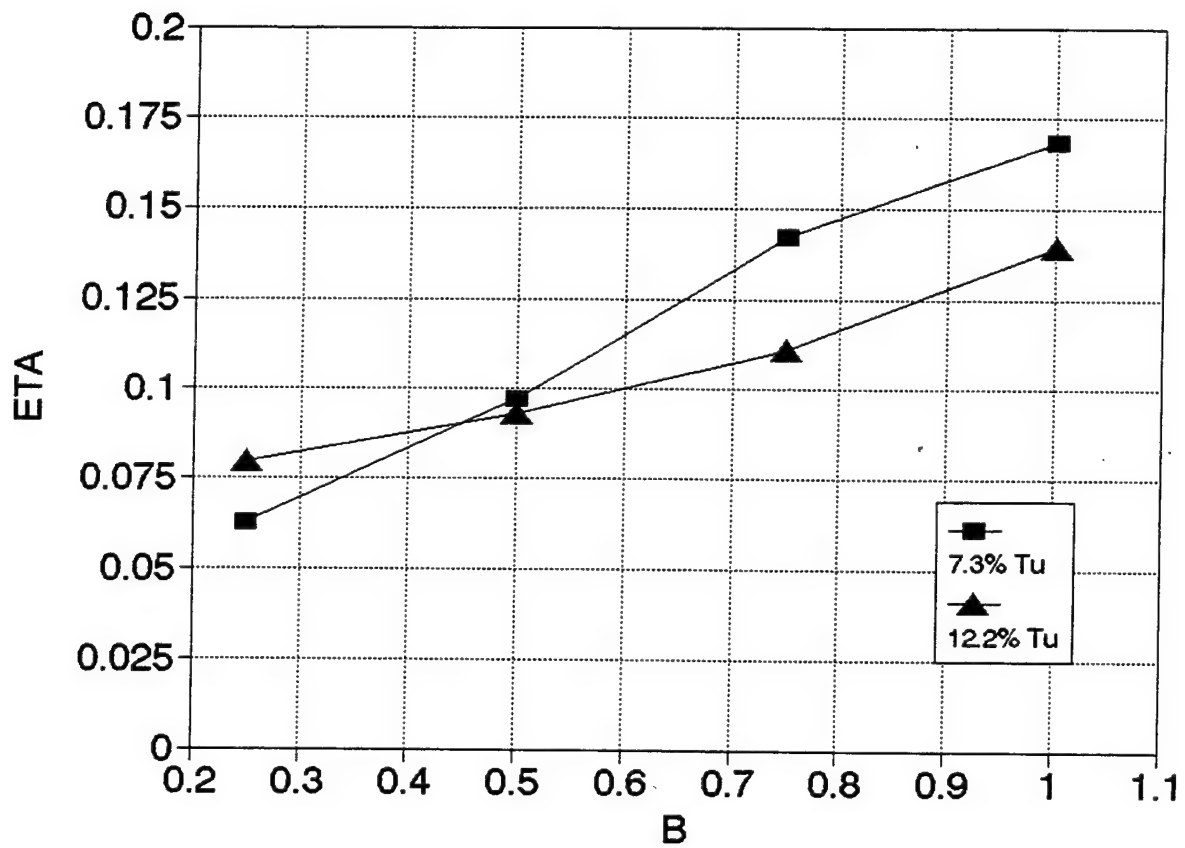


Figure E-42. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 45$ , Free stream velocity at injection = 37 m/s

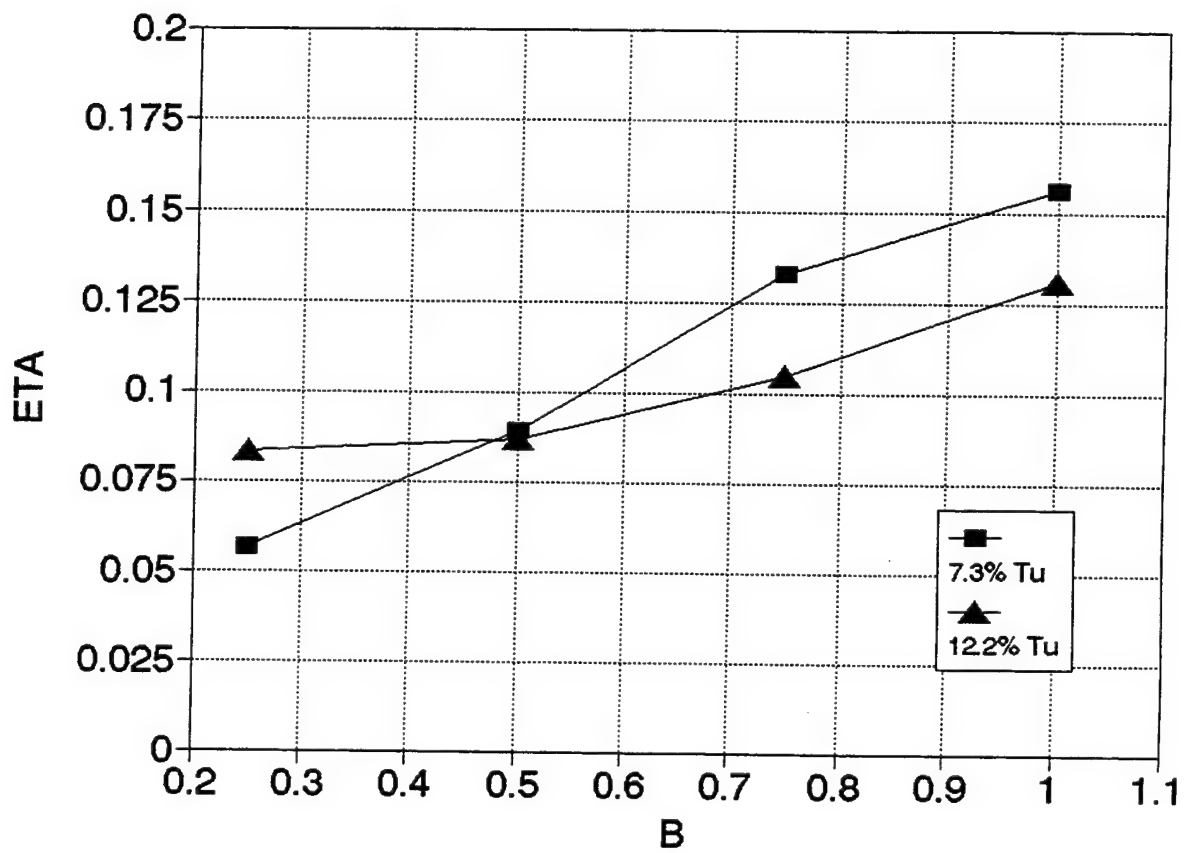


Figure E-43. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 50$ , Free stream velocity at injection = 37 m/s

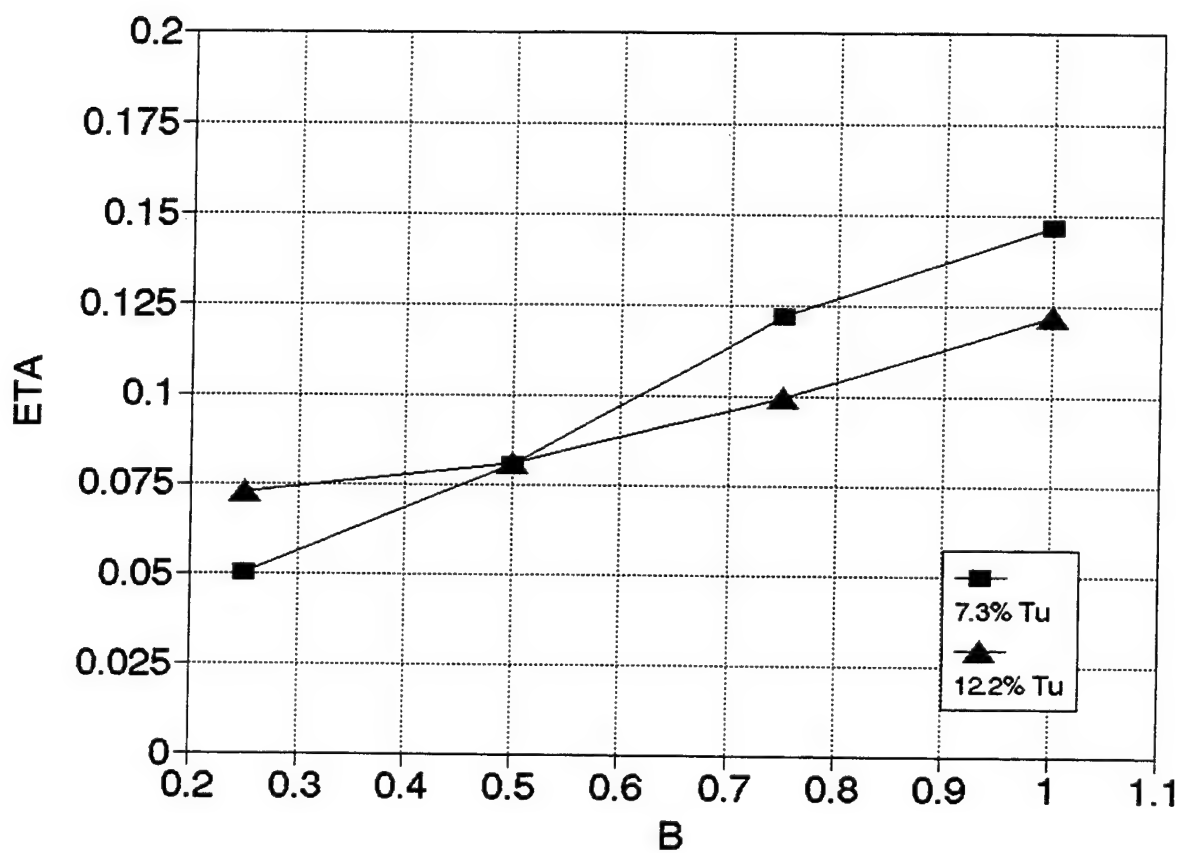


Figure E-44. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 55$ , Free stream velocity at injection = 37 m/s

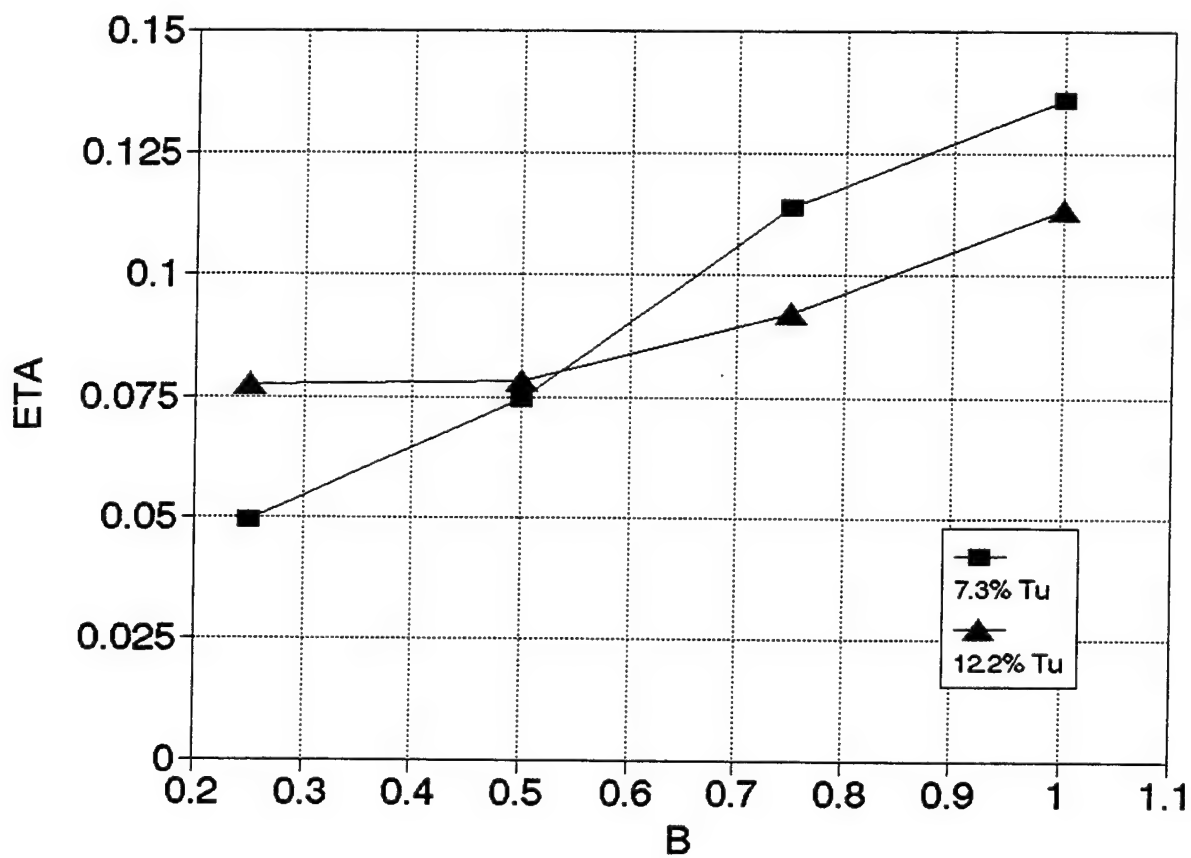


Figure E-45. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 60$ , Free stream velocity at injection = 37 m/s

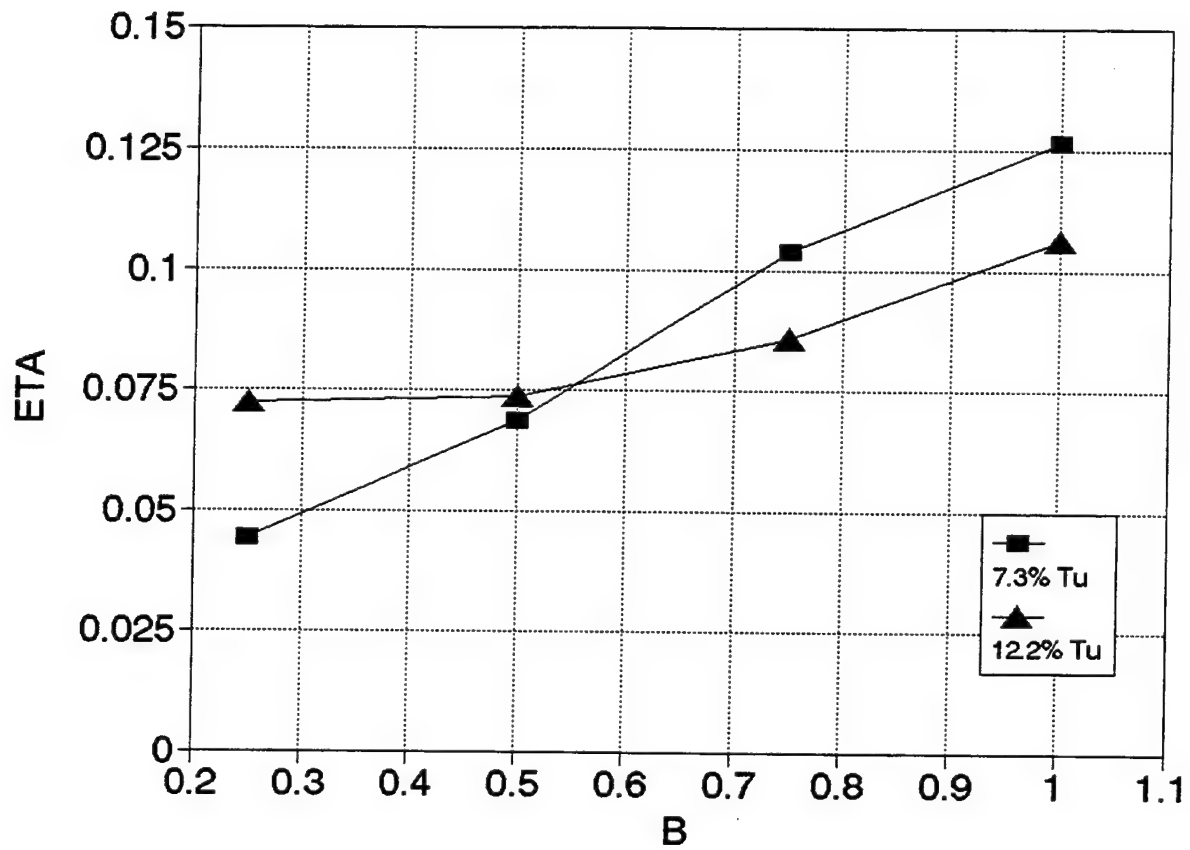


Figure E-46. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 65$ , Free stream velocity at injection = 37 m/s

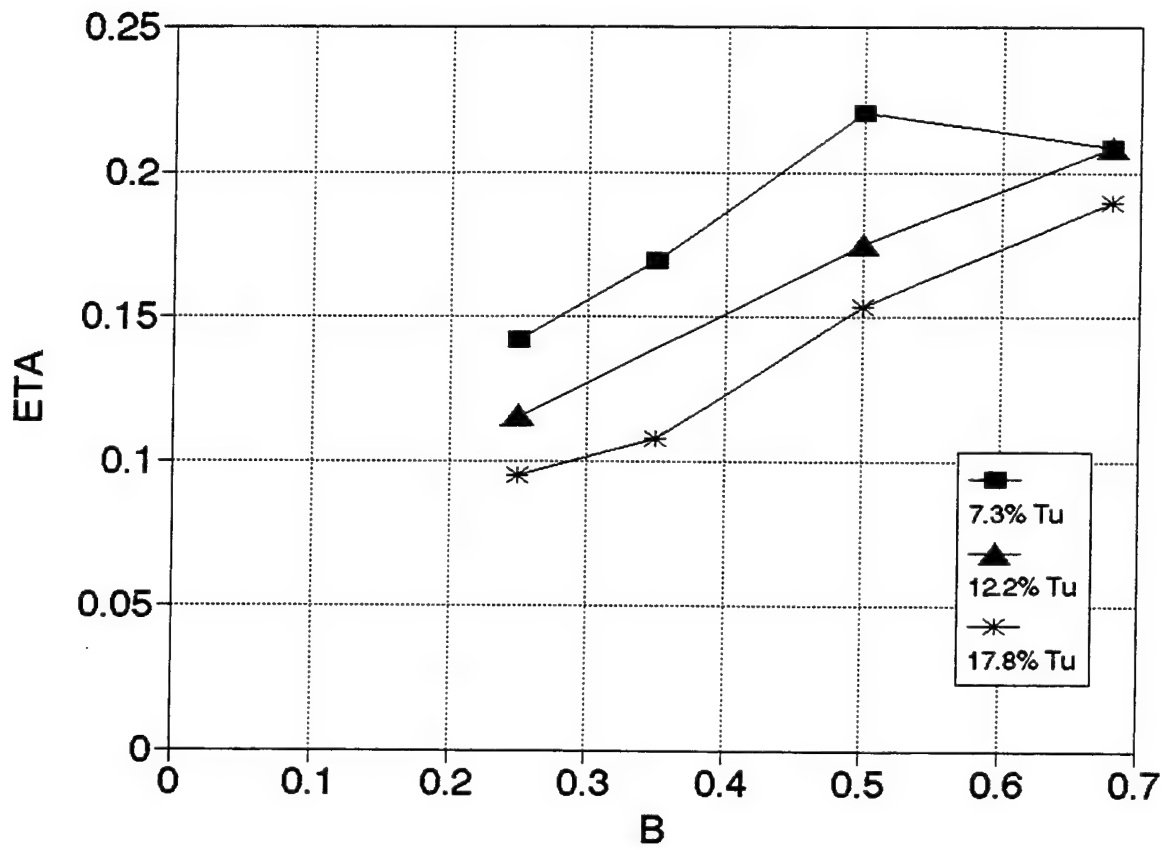


Figure E-47. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 70$ , Free stream velocity at injection = 37 m/s



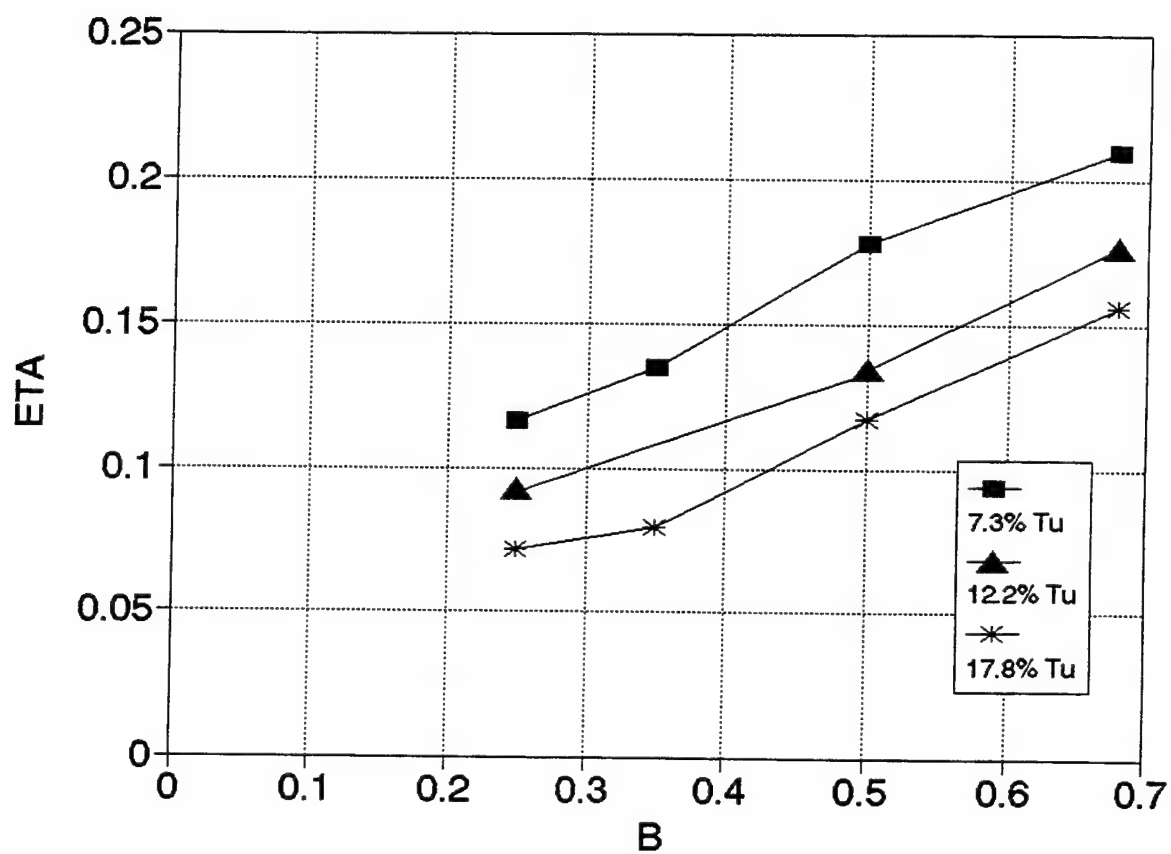


Figure E-48. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 15$ , Free stream velocity at injection = 60 m/s

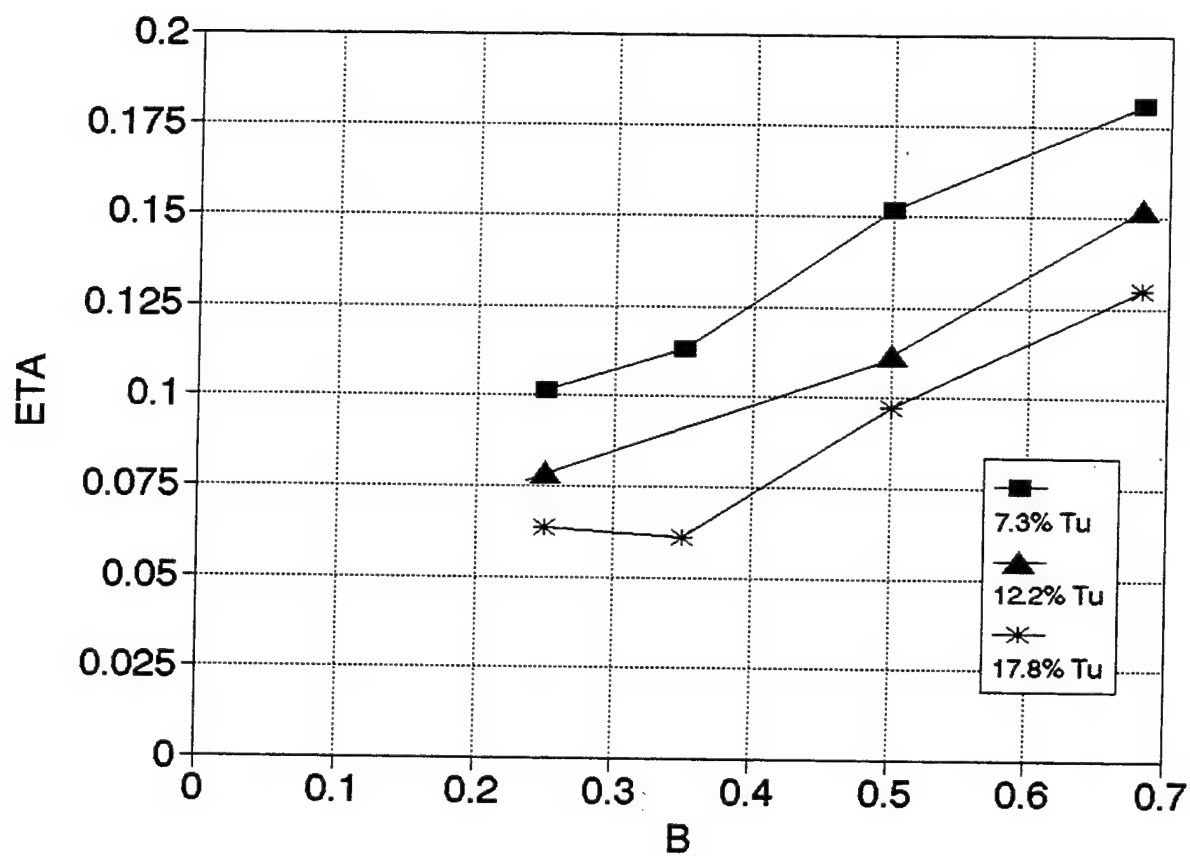


Figure E-49. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 20$  Free stream velocity at injection = 60 m/s

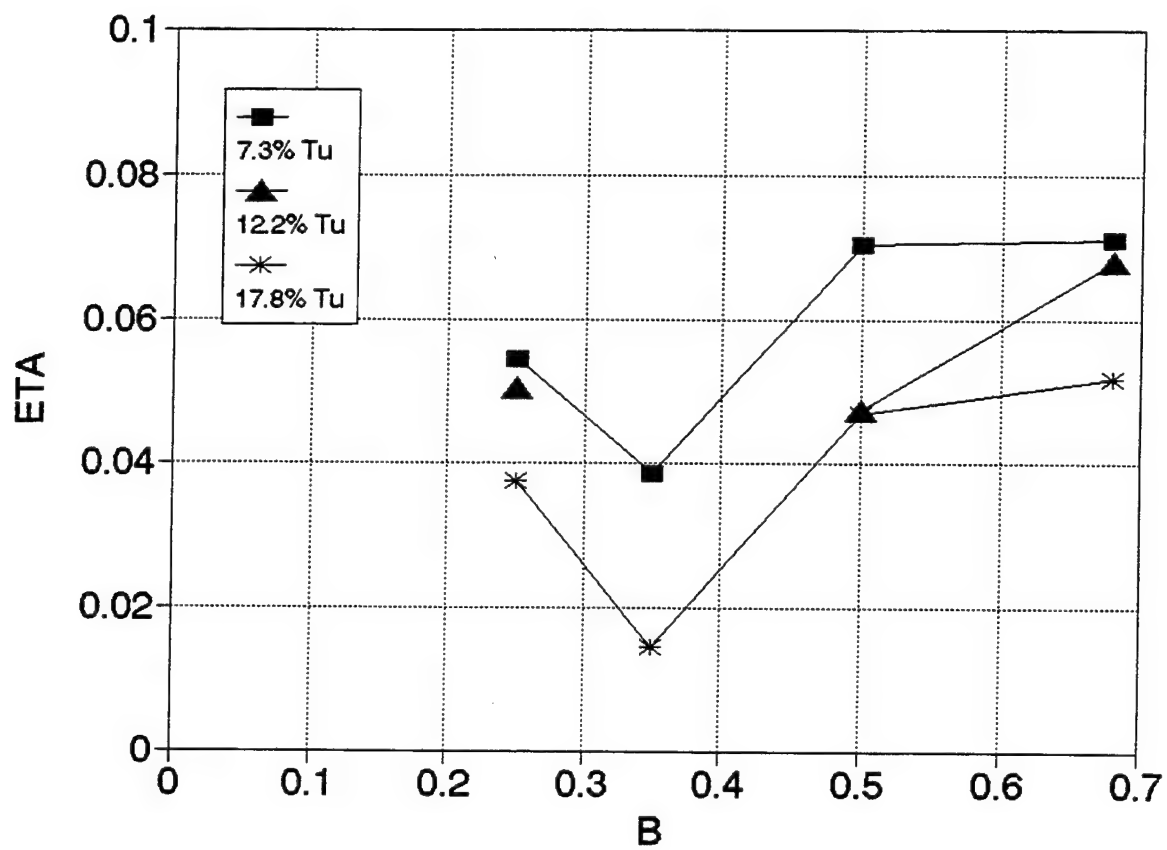


Figure E-50. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 25$ , Free stream velocity at injection = 60 m/s

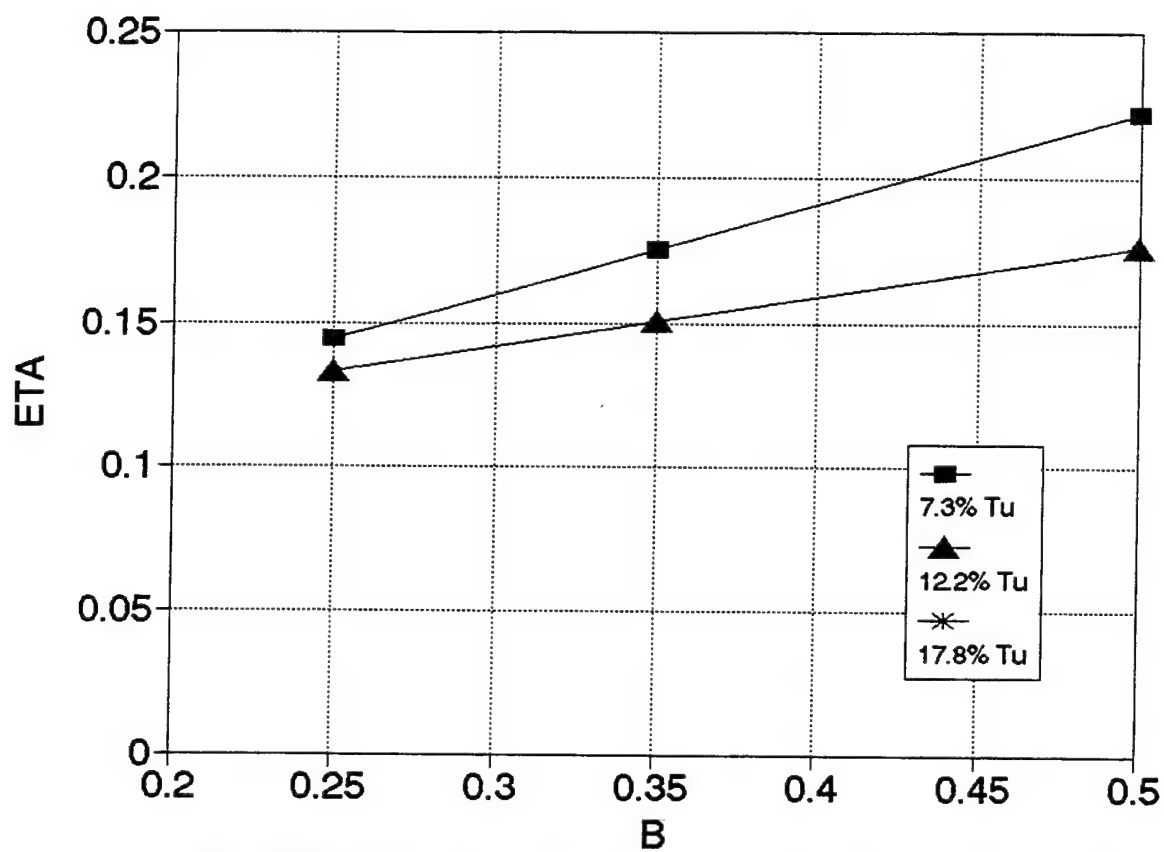


Figure E-51. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 65$ , Free stream velocity at injection = 60 m/s

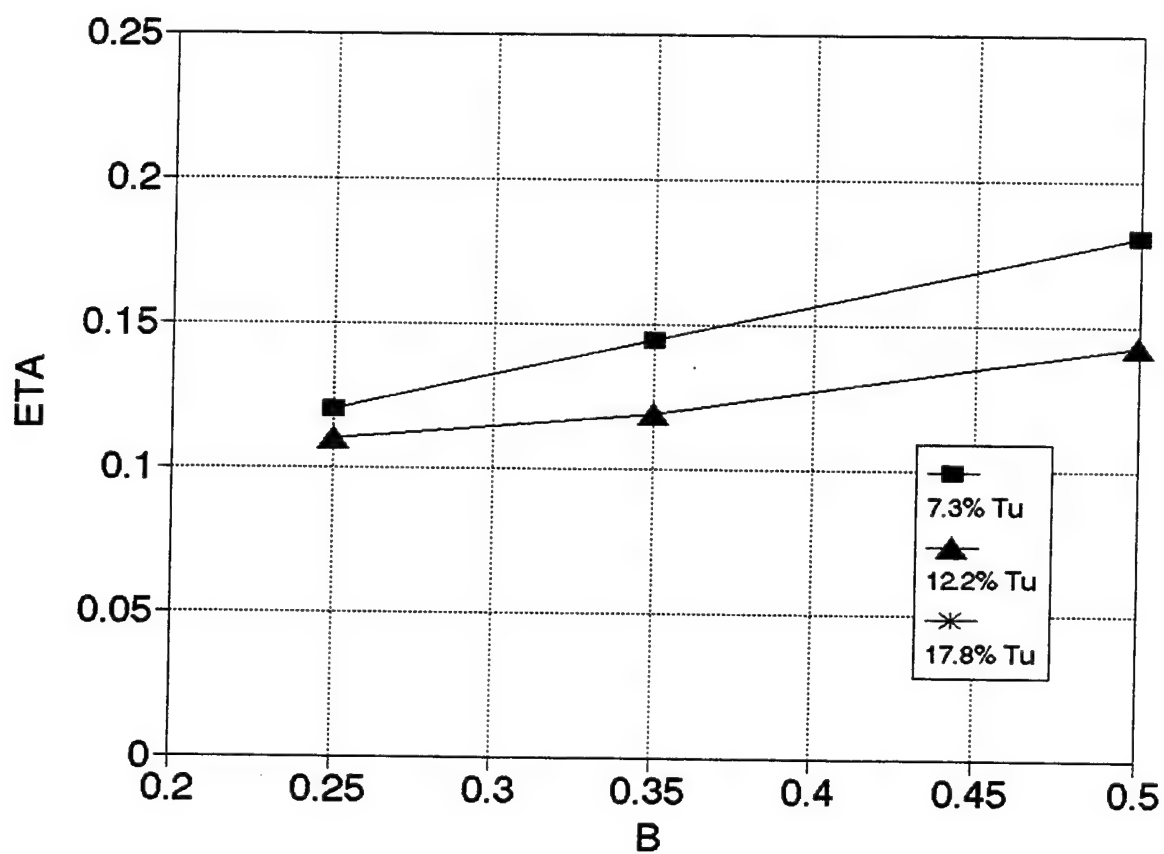


Figure E-52. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 15$ , Free stream velocity at injection = 85 m/s

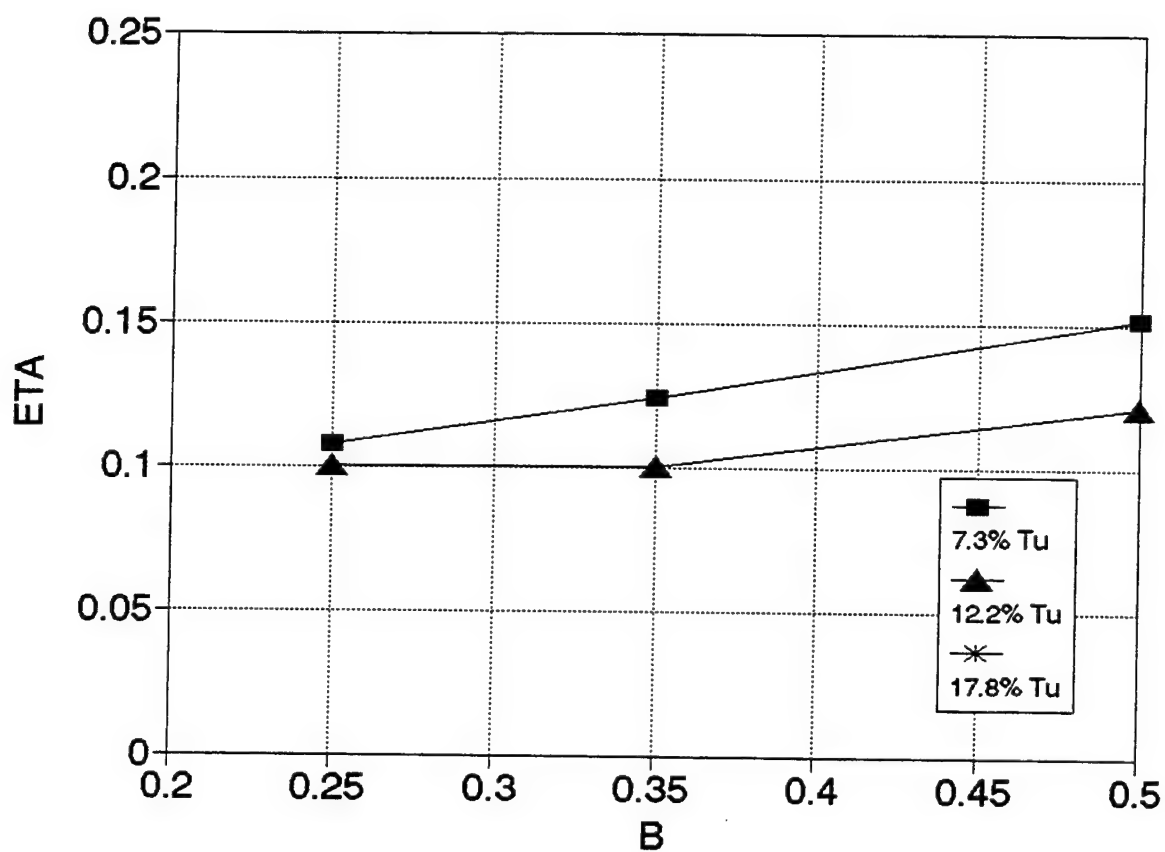


Figure E-53. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 20$ , Free stream velocity at injection = 85 m/s

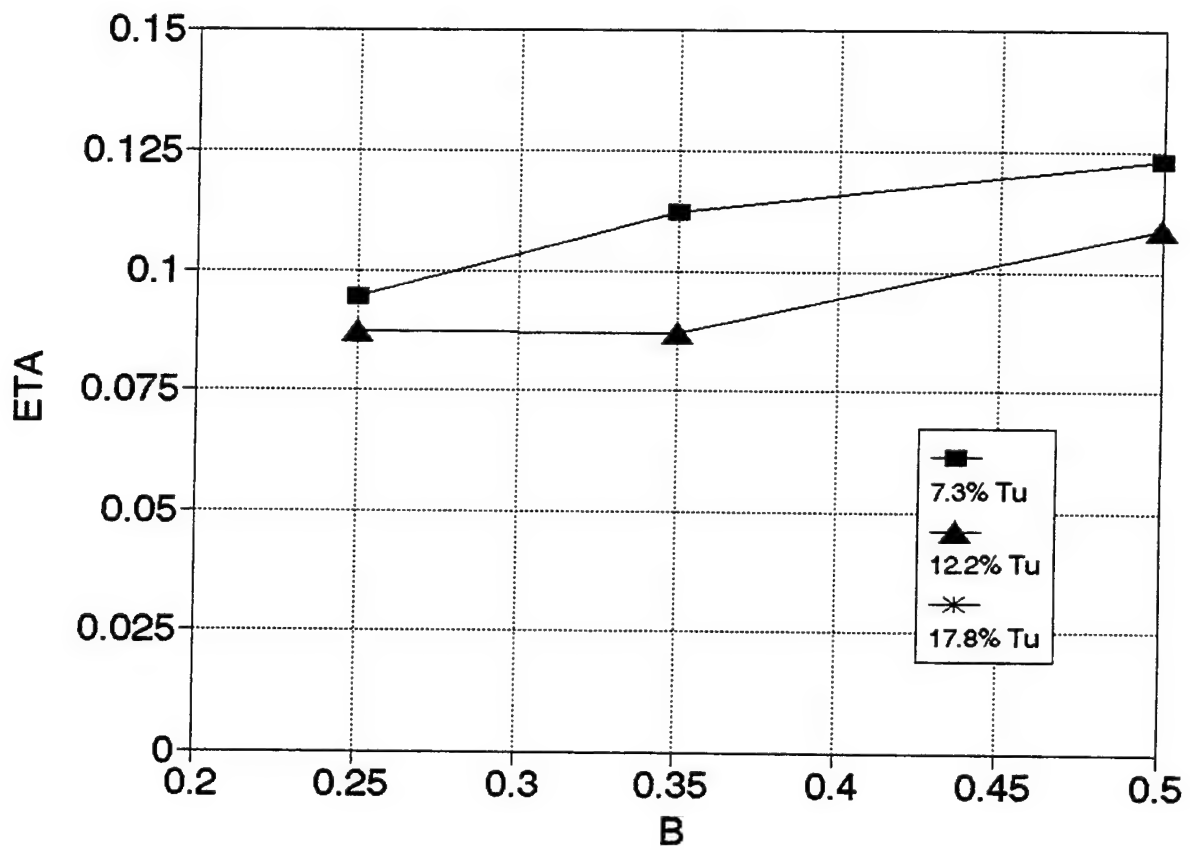


Figure E-54. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 25$ , Free stream velocity at injection = 85 m/s

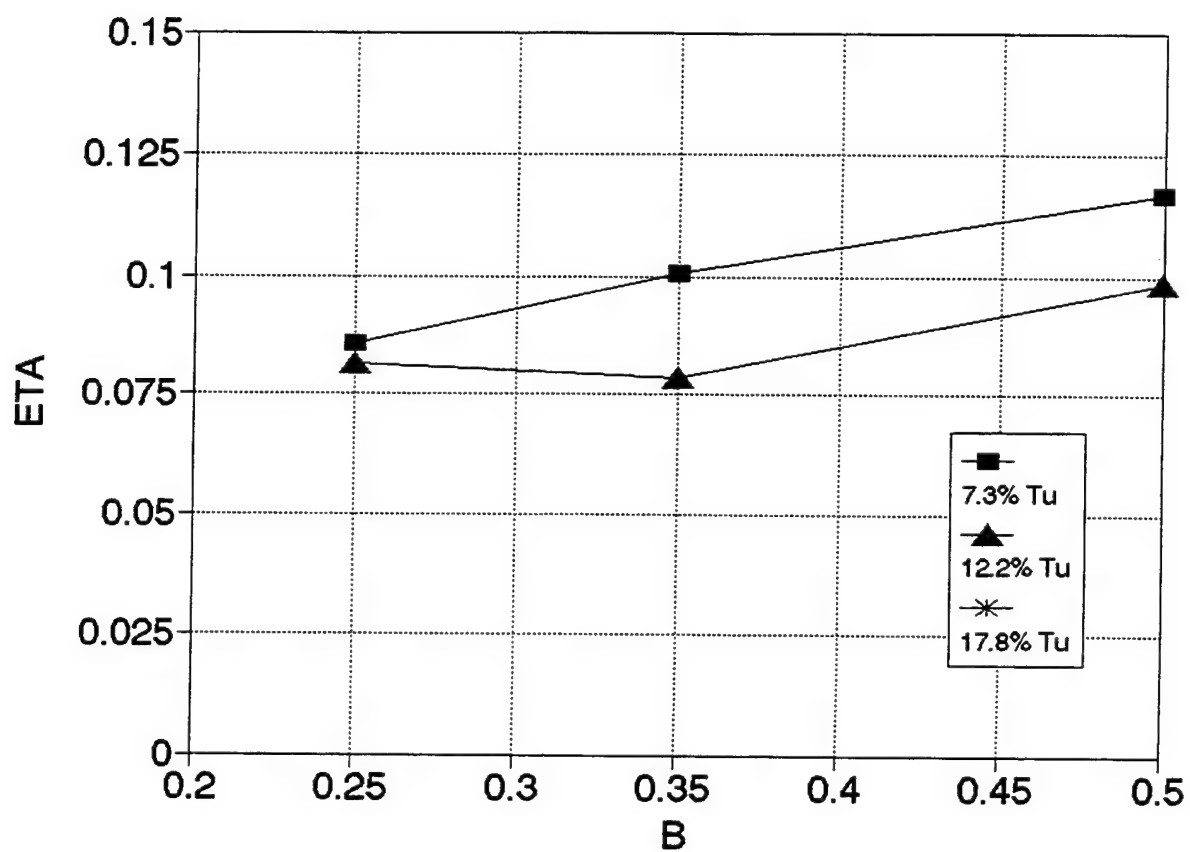


Figure E-55. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 30$  Free stream velocity at injection = 85 m/s



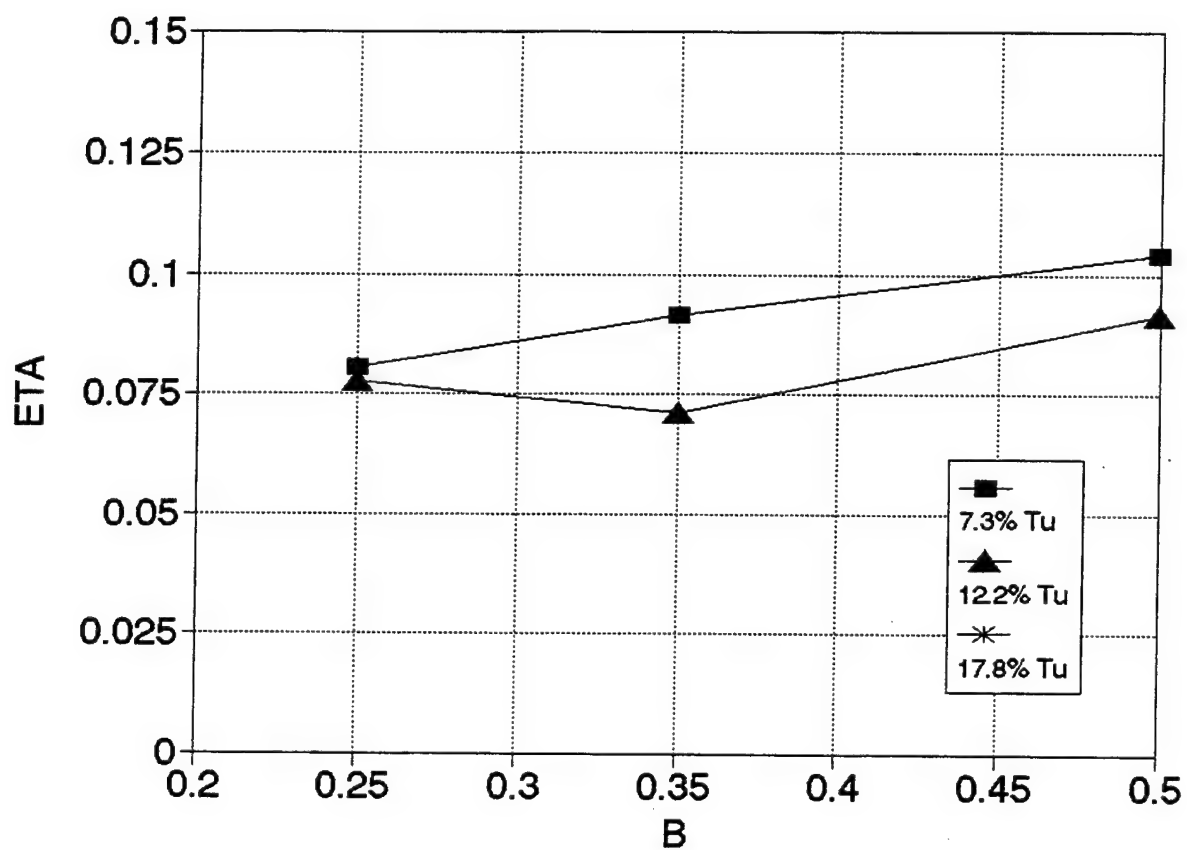


Figure E-56. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 35$ , Free stream velocity at injection = 85 m/s

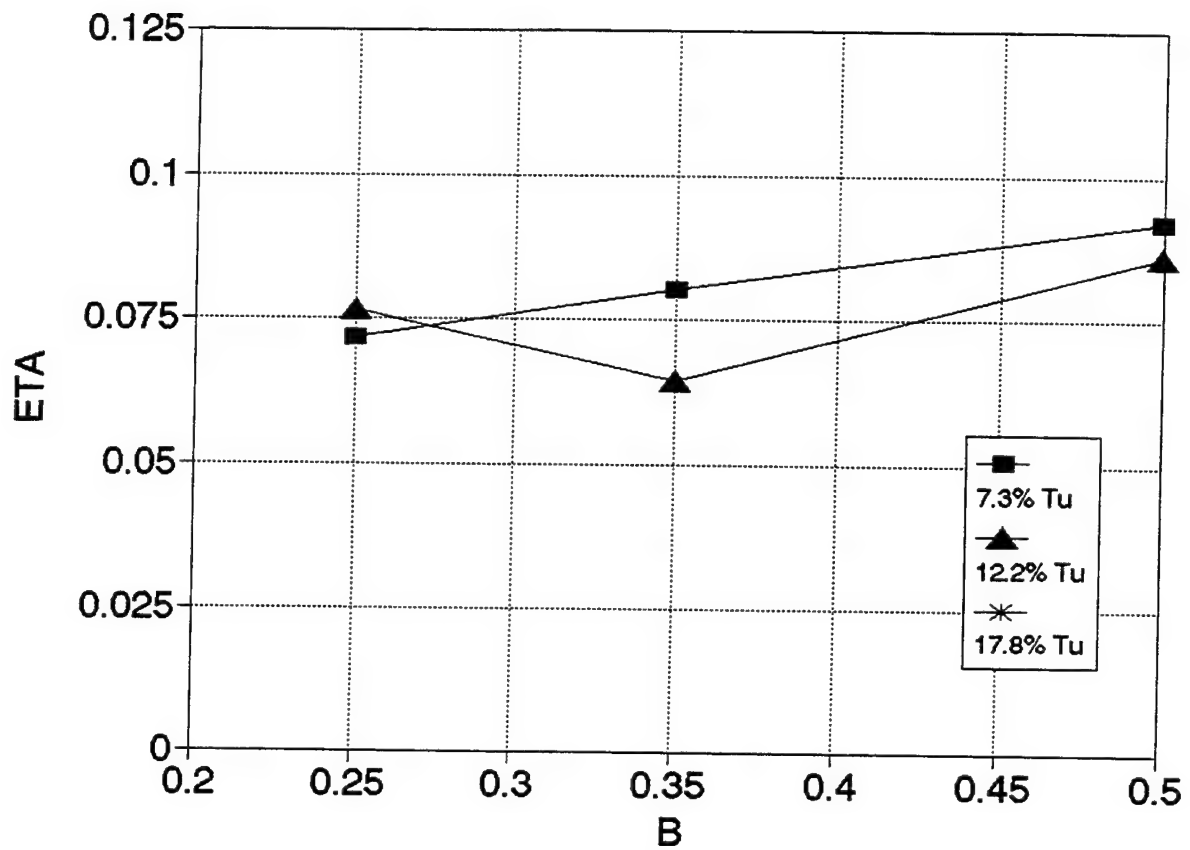


Figure E-57. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 40$ , Free stream velocity at injection = 85 m/s

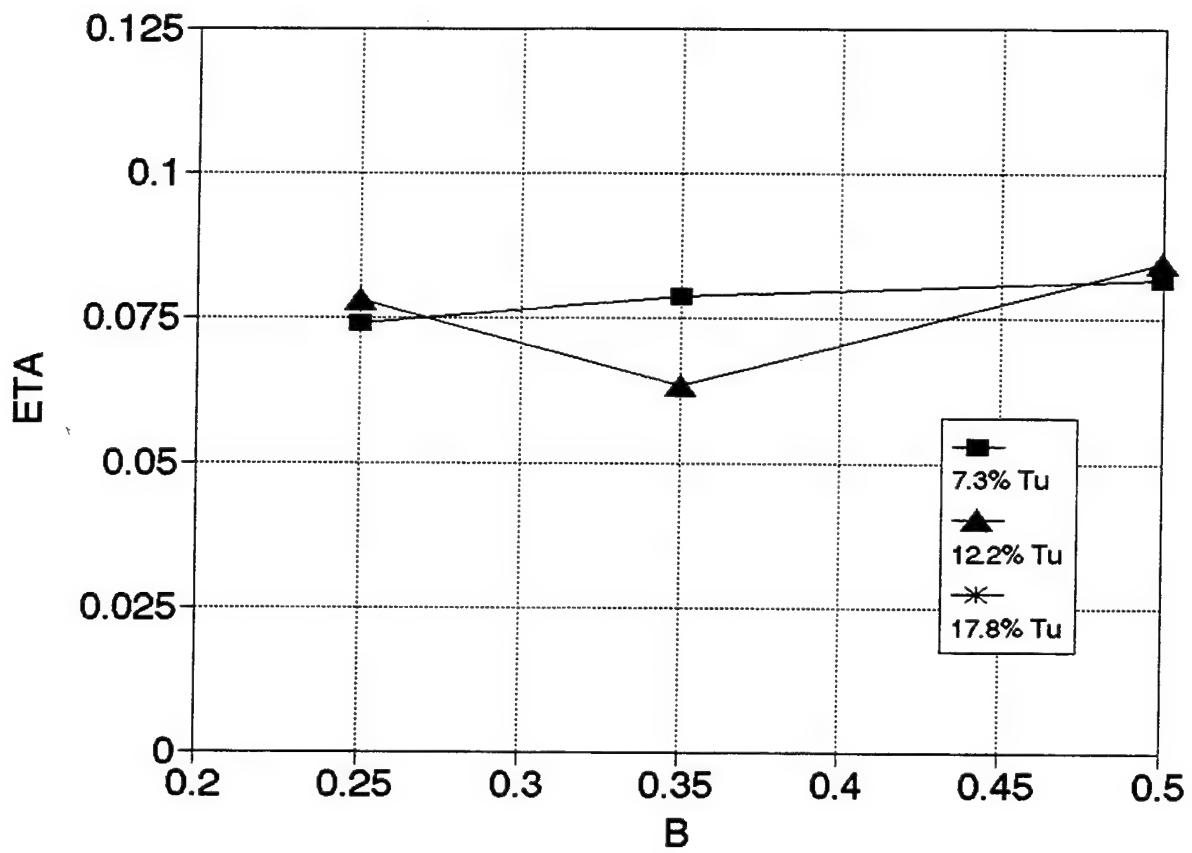


Figure E-58. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 45$ , Free stream velocity at injection = 85 m/s

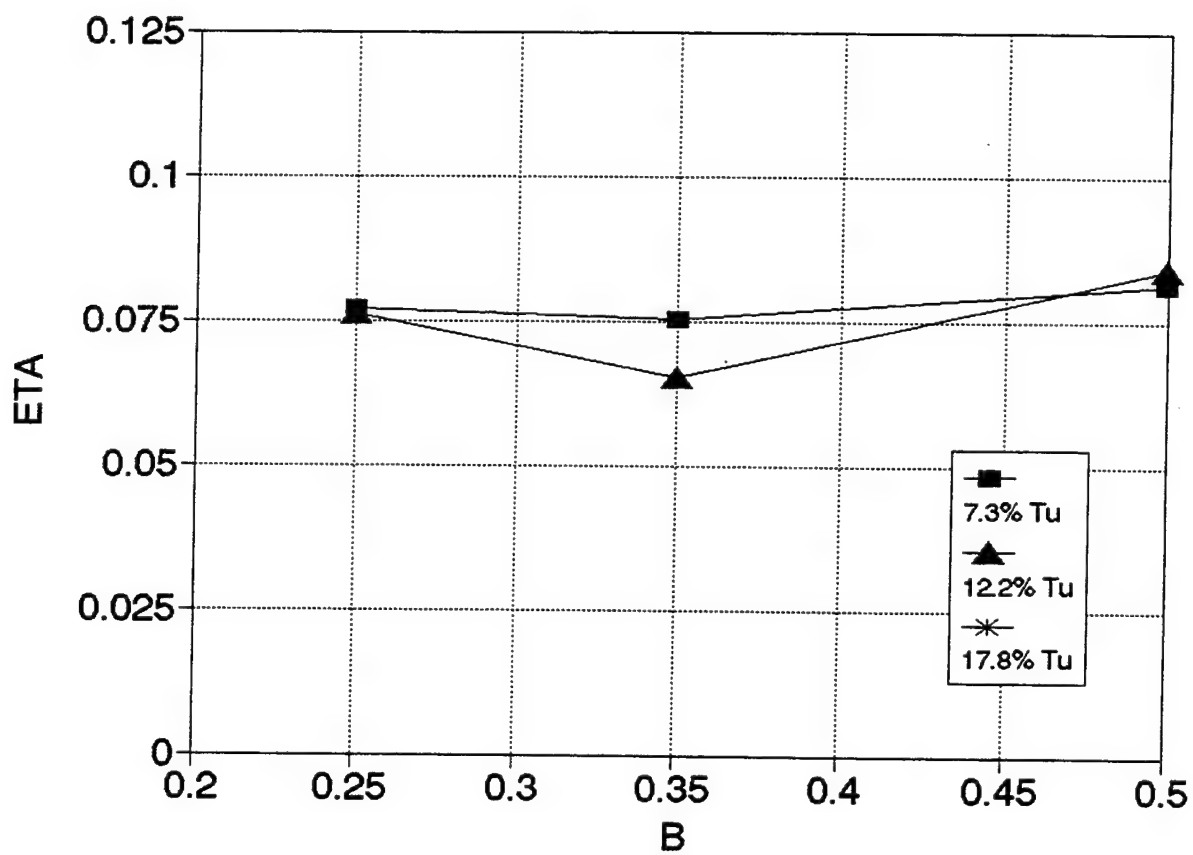


Figure E-59. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 50$ , Free stream velocity at injection = 85 m/s

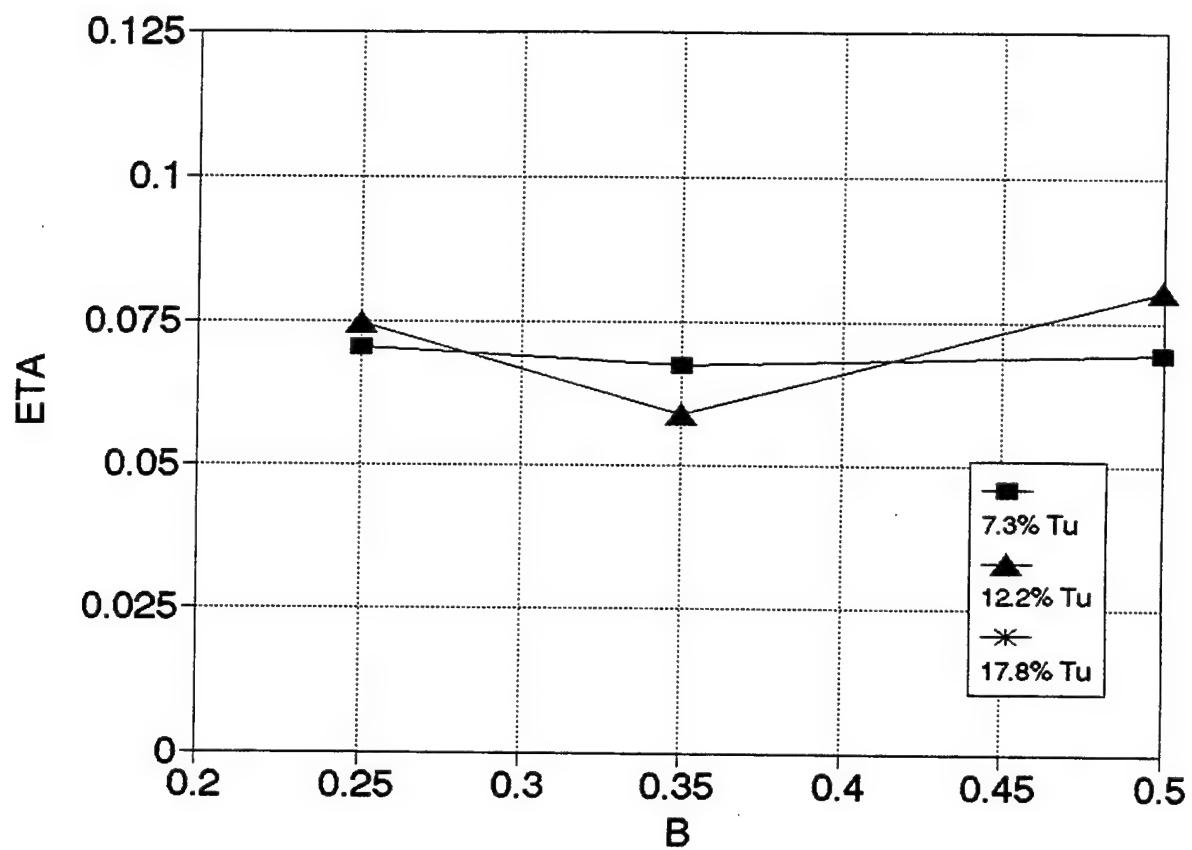


Figure E-60. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 55$ , Free stream velocity at injection = 85 m/s

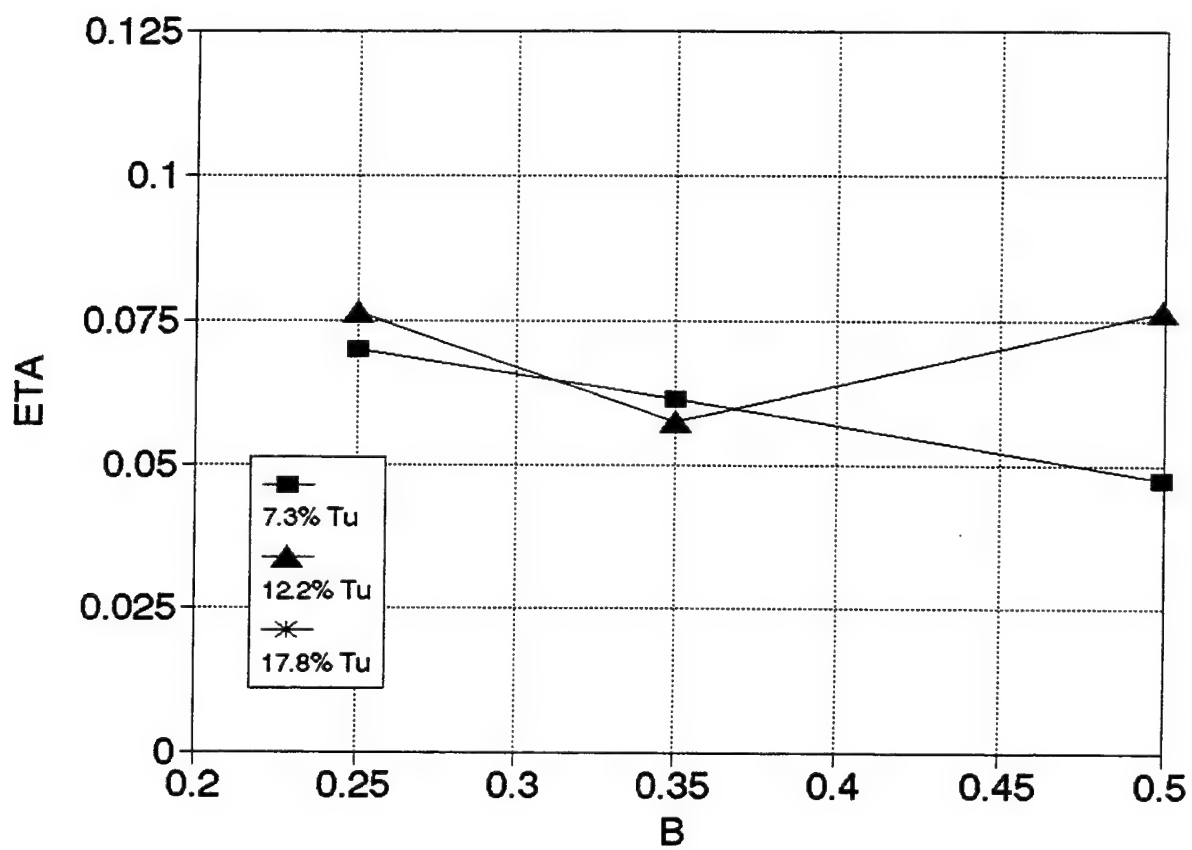


Figure E-61. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 60$ , Free stream velocity at injection = 85 m/s

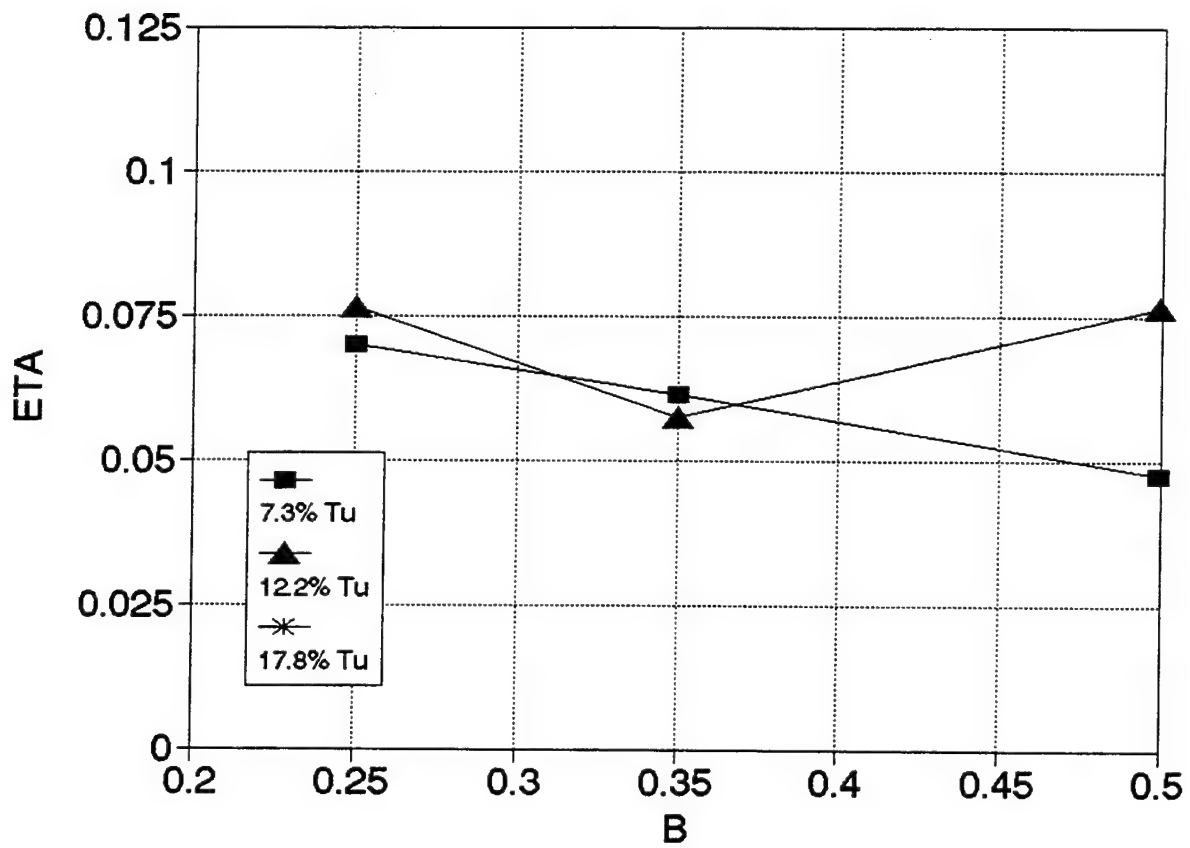


Figure E-62. Effect of Free Stream Turbulence on Film Cooling Effectiveness  
 $X/D = 70$ , Free stream velocity at injection = 85 m/s

## Appendix F : Effectiveness Correlation with B, X, I and S

A correlation of effectiveness with various non-dimensional combinations of parameters such as: blowing ratio (B), distance from injection to a station (X), momentum flux (I), and distance between injection hole centers (S), was attempted unsuccessfully. The specific combinations were,  $\eta/B \sim XI/S$ ,  $\eta \sim X/BS$ ,  $\eta \sim XB/S$ ,  $\eta/I \sim XI/S$  and  $\eta/IB \sim XI/S$ . These combinations were plotted for each blowing rate and all three Tu levels.

In research done by Cirrello (1991), with one and two rows of holes and no significant free stream turbulence, the first grouping,  $\eta/B \sim XI/S$ , came the closest to a collapse for blowing rates from 0.50 to 1.74. In the present research none of the groupings produced a collapse of effectiveness amongst different blowing rates or different Tu levels.



## Appendix G :Sample calculation of $\tau_0$ and $z_0$

To obtain the kinematic shear stress,  $\tau_0$ , the definition for the friction parameter

$$\gamma = \frac{\tau_0^{1/2}}{kU_0} \quad (G-1)$$

was used, where  $U_0$  is the local maximum value of the velocity at a station. Solving equation (G-1)

for  $\tau_0$  yields

$$\tau_0 = (\gamma k U_0)^2 \quad (G-2)$$

The value for the friction parameter in equation (G-2) is obtained from a plot of  $\gamma$  with  $Re_x$  (Figure 7.3 in Townsend, 1976). This plot displays several curves, each corresponding to a different velocity ratio defined as  $V = U_1/U_0$ , in which  $U_1$  is the velocity of a free stream superimposed on the wall jet (for this research  $U_1 = 0$ , therefore  $V = 0$ ). The value of  $U_0$  is obtained from the velocity profile at the  $x$  location where the self-preserving distribution is sought.

For example, at  $x = 64.6$  cm:  $U_0 = 51.95$  m/s,  $\gamma = 0.131$ ,  $k = 0.41$ . Substituting these values into equation (G-2) yields

$$\tau_0 = [(0.131)(0.41)(51.95 \text{ m/s})]^2 = 7.79 \text{ m}^2 / \text{s}^2$$

To find  $z_0$  at the  $x$  location in question, the self-preserving distribution equation

$$U = \frac{\tau_0^{1/2}}{k} \left[ \log \left( \frac{z}{z_0} \right) - \frac{z}{\eta_1 l_0} \right] \quad (5-2)$$

is solved for  $z_0$  by dividing both sides of the equation by the kinematic shear term and isolating the log term in the right hand side

$$U \left( \frac{k}{\tau_0^{1/2}} \right) + \frac{z}{\eta_1 l_0} = \log \left( \frac{z}{z_0} \right),$$

taking the log base 10 of both sides of the equation

$$10^{\left[ U \left( \frac{k}{\tau_0^{1/2}} \right) + \frac{z}{\eta_1 l_0} \right]} = \frac{z}{z_0},$$

and finally multiplying through by  $z_0$

$$z_0 = \frac{z}{10^{\left[ U \left( \frac{k}{\tau_0^{1/2}} \right) + \frac{z}{\eta_1 l_0} \right]}} \quad (\text{G-3})$$

Next, an arbitrary point (U, z) in the profile at  $x = 64.6$  cm is substituted into equation (G-3), along with the value of  $\eta_1 l_0 = Y_m$  at that location. These quantities are:  $U = 51.52$  m/s,  $z = 0.01524$  m, and  $Y_m = 0.02035$  m. The substitution yields

$$z_0 = \frac{0.01524m}{10^{\left[ 51.52m/s \left( \frac{0.41}{(7.79m^2/s^2)^{1/2}} \right) + \frac{0.01524m}{0.02032m} \right]}} = 7.373 \times 10^{-11}$$

## **Vita**

Lilith I. Sorensen was born on June 9<sup>th</sup> 1965 in Manhattan, New York. She graduated from Ana Roque high school in Humacao, Puerto Rico in 1982 and attended the Interamerican University, Hato Rey, Puerto Rico then transferred to Embry Riddle Aeronautical University, Daytona Beach, Florida from which she graduated with a Bachelor of Science in Aeronautical Engineering in April 1989. Upon graduation she was employed as a Mechanical Engineer for ECC International, Orlando, Florida where she designed flight control components for the Air Force C-17 maintenance trainer. She later worked as an Electronics Engineer for the Defense Electronics Supply Center (DESC), Dayton, Ohio, performing audits and providing management oversight of the test and manufacture of electronics component to be used in military weapons systems. While at DESC she entered the School of Engineering, Air Force Institute of Technology, as a part-time student in July 1992. She is now employed as a Heat Transfer Engineer for TK Engineering, Springdale, Ohio where she performs heat transfer analysis of gas turbine engine components.

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REPORT DOCUMENTATION PAGE			Form Approved OMB No. 0704-0188	
Public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Washington Headquarters Services, Directorate for Information Operations and Reports, 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302, and to the Office of Management and Budget, Paperwork Reduction Project (0704-0188), Washington, DC 20503.				
1. AGENCY USE ONLY (Leave blank)		2. REPORT DATE March 1998		3. REPORT TYPE AND DATES COVERED Thesis
4. TITLE AND SUBTITLE Effect of High Free Stream Turbulence on Film Cooling Using Double Row of 30° Slant-Hole Injectors			5. FUNDING NUMBERS	
6. AUTHOR(S) Lilith I. Sorensen				
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) Air Force Institute of Technology, 2950 P Street WPAFB OH 45433-7765			8. PERFORMING ORGANIZATION REPORT NUMBER AFIT/GAE/ENT/98M-03	
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES) Dr. Richard Rivir Phone: (937)255-7621 AFRL/POTT 1950 Fifth Street WPAFB OH 45433-7251			10. SPONSORING/MONITORING AGENCY REPORT NUMBER	
11. SUPPLEMENTARY NOTES Advisor Dr. Paul King				
12a. DISTRIBUTION/AVAILABILITY STATEMENT Approved for public release; distribution unlimited			12b. DISTRIBUTION CODE	
13. ABSTRACT (Maximum 200 words) This study free stream turbulence levels from 7.3 % to 17.8 % are applied to film cooling over a flat plate with two rows of 30 ° slant-holes using a wall jet as the main stream air supply and source of turbulence. Blowing ratios are varied from 0.25 to 2.0 and free stream velocities at injection range from 10 m/s to 85 m/s. A constant density ratio of 1.07 is kept throughout the experiment. Results show that for different magnitudes of blowing ratio free stream turbulence has a different influence on effectiveness. At the forward stations, where blow-off was present at high blowing ratios, high free stream turbulence increased effectiveness. While at mid to aft stations for all blowing ratios high free stream turbulence decreased effectiveness. A correlation for effectiveness with a non-dimensional group of film cooling parameters is offered. The effectiveness correlated best 10 to 15 diameters after injection where blow-off effects are no longer present, and before the very far downstream stations where uncertainties in the measurements were greater.				
14. SUBJECT TERMS Film Cooling, Effectiveness, Blowing Ratio, Coolant Injection, Free Stream Turbulence, Discrete hole Coolant Injection, Flat Plate, Wall Jet, Slant-Holes, Mass Flux Ratio			15. NUMBER OF PAGES 200	
			16. PRICE CODE	
17. SECURITY CLASSIFICATION OF REPORT Unclassified	18. SECURITY CLASSIFICATION OF THIS PAGE Unclassified	19. SECURITY CLASSIFICATION OF ABSTRACT Unclassified	20. LIMITATION OF ABSTRACT UL	